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**Development of a Multiple-Parameter
Nonlinear Perturbation Procedure for
Transonic Turbomachinery Flows:
Preliminary Application to
Design/Optimization Problems**

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Development of a Multiple-Parameter Nonlinear Perturbation Procedure for Transonic Turbomachinery Flows: Preliminary Application to Design/Optimization Problems

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DEVELOPMENT OF A MULTIPLE-PARAMETER NONLINEAR
PERTURBATION PROCEDURE FOR TRANSONIC
TURBOMACHINERY FLOWS: PRELIMINARY
APPLICATION TO DESIGN/OPTIMIZATION
PROBLEMS

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SUMMARY

An investigation was conducted to continue the development of perturbation procedures and associated computational codes for rapidly determining approximations to nonlinear flow solutions, with the purpose of establishing a method for minimizing computational requirements associated with parametric design studies of transonic flows in turbomachines. The results reported here concern the extension of the previously-developed successful method for single-parameter perturbations to simultaneous multiple-parameter perturbations, and the preliminary application of that multiple-parameter procedure in combination with an optimization method to blade design/optimization problems.

In order to provide as severe a test as possible of the method, attention is focused in particular on transonic flows which are highly supercritical. Flows past both isolated blades and compressor cascades, involving simultaneous changes in both flow and geometric parameters, are considered. Comparisons with the corresponding 'exact' nonlinear solutions display remarkable accuracy and range of validity, in direct correspondence with previous results for single-parameter perturbations. Initial applications of the perturbation method combined with an optimization procedure demonstrate the ability of the multiple-parameter method to work accurately in a design environment and establish its potential for reducing the computational work required in such applications by an order of magnitude.

1. INTRODUCTION

With the continuing success of advanced computational methods to determine solutions to increasingly complex fluid dynamic phenomena, it has become clearly apparent that in order to employ these methods in applications requiring routine high-frequency use, a means must be found to reduce the computational demands necessary in their straightforward application. While this need exists across a spectrum of aerodynamic users, it is particularly high in turbomachinery applications. There both the basic aerodynamic computation is time-consuming and the number of variable flow and geometry parameters are large, making any turbomachinery parametric or design study computationally expensive under the best of circumstances, and in many instances using more advanced codes, prohibitively so.

The ultimate objective underlying this study is to develop the means of reducing substantially the overall computational requirements necessary for turbomachinery design or parametric studies by minimizing the actual number of "expensive" numerical flow solutions required. That such procedures are achievable has been successfully demonstrated in the previous phase (ref. 1) of this study. In that work, a perturbation method was developed and tested on a large number of nonlinear flow problems involving single-parameter changes of a variety of flow and geometric parameters. Subcritical and supercritical flows past isolated blades and compressor cascades were considered, with particular emphasis placed on supercritical transonic flows which exhibited large surface shock movements over the parametric range studied. Comparisons of the perturbation predictions with the corresponding 'exact' nonlinear solutions indicated a remarkable accuracy and range of validity of the perturbation method.

The work reported here describes the continued extension and refinement of that perturbation technique, and has focused on the development of its capability for actual application to practical turbomachinery design problems requiring the highly repetitive use of computational codes to determine a large number of related flow solutions. Two primary tasks were involved. The first consisted of the extension of the method to treat simultaneous multiple-parameter perturbations. The second involved the combination of the perturbation method with an optimization procedure, and the preliminary application of that combination to blade design/optimization problems. The nature of the present work is both exploratory and developmental in that aspects of the procedure--such as its validity, range of accuracy, and computational economy for multiple-parameter perturbation problems and its workability in an optimization design environment--will be investigated, and a computational code for multiple-parameter perturbations will be developed.

2. ANALYSIS

2.1 Perturbation Concept and Previous Applications

The classical approach of performing a perturbation analysis--consisting of establishing and solving a series of linear perturbation equations--appears an an obvious choice for the current applications. However, results from the initial phase of this study (ref. 2) demonstrate that for applications to sensitive flows such as occur in transonic situations, the basic linear variation assumption fundamental to the technique is sufficiently restrictive that the permissible range of parameter variation is so small to be of little practical use. An interesting alternative to the linear perturbation equation approach has recently been successfully examined in which a correction technique is used that employs two or more nonlinear base solutions. For that method, the basic perturbation solution is determined simply by differencing two nonlinear base flow solutions removed from one another by some nominal change of a particular flow or geometrical quantity. A unit perturbation solution is then obtained by dividing that result by the change in the perturbed quantity. Related solutions are determined by multiplying the unit perturbation by the desired parameter change and adding that result to the base flow solution. This simple procedure, however, only works directly for continuous flows for which the perturbation change does not alter the solution domain. For those perturbations which change the flow domain, coordinate stretching is necessary to ensure proper definition of the unit perturbation solution. Similarly, for discontinuous flows, coordinate straining is necessary to account for movement of discontinuities due to the perturbation.

In a number of recent applications of the method (refs. 1-7), results have been obtained which demonstrate the accuracy, range of validity, and versatility of perturbation methods based on such ideas. The most extensive and systematic of these are provided in references 1 and 7, where results are reported for a variety of flow and geometry parameter perturbation case studies of nonlinear subsonic and transonic flows past both isolated blades and compressor cascades. In those results, particular emphasis was placed on strongly supercritical transonic flows which exhibit large surface shock movements over the parameter range studied. Comparisons of the perturbation results with the corresponding 'exact' nonlinear solution display remarkable accuracy across the spectrum of examples studied.

The basis of this accuracy lies in the use of coordinate straining. This provides the means in determining the unit perturbation to account properly for the displacement of discontinuities and maxima of high-gradient regions due to a

parameter change. This in turn enables the perturbation method to maintain uncommon accuracy in regions of high gradients over large parametric ranges. In reference 1, a detailed examination was made of the effect of employing different classes of straining functions. Those results have illustrated deficiencies in certain classes of straining functions and have lead to the identification of a superior class of straining functions. These results are discussed in more detail in section 2.3.

2.2 Simultaneous Multiple-Parameter Perturbation Formulation

To provide the theoretical basis of the perturbation method as applied to simultaneous multiple-parameter perturbations of flows containing multiple shocks or high-gradient regions, consider the formulation of the procedure at the full potential equation level, since all of the results presented here are based on that level. Denote the operator L acting on the full velocity potential Φ as that which results in the two-dimensional full-potential equation for Φ , i.e.,

$$L[\Phi] = 0 \quad (1)$$

If we now expand the potential in terms of zero- and higher-order components in order to account for the variation of M arbitrary geometrical or flow parameters q_j

$$\Phi = \Phi_0 + \sum_{j=1}^M \epsilon_j \Phi_{1,j} + \dots \quad (2)$$

$$q_j = q_{0,j} + \Delta q_j$$

and then insert this into the governing equation (1), expand the result, order the equations into zero- and first-order components, and make the obvious choice of expansion parameters $\epsilon_j = \Delta q_j$ we obtain the following governing equations for the zero- and M first-order components

$$L[\Phi_0] = 0 \quad (3)$$

$$L_1[\Phi_{1,j}] + \frac{\partial}{\partial q_j} L[\Phi_0] = 0$$

Here L_1 is a linear operator whose coefficients depend on zero-order quantities and $\partial L[\Phi_0]/\partial q_j$ represents a 'forcing' term due to the q_j^{th} perturbation. Actual forms of L_1 and the 'forcing' term are provided in reference 2 for a variety of flow and geometry parameter perturbations of a two-dimensional turbo-machine, and in reference 4 for profile shape perturbations of an isolated airfoil. An important point regarding Equation (3) for the first-order perturbations Φ_{1j} is that these equations

represent a unit perturbation independent of the actual value of the perturbation quantity ε_j .

Appropriate account of the movement of a multiple number of discontinuities and maxima of high-gradient regions due to the perturbation is now accomplished by the introduction of strained coordinates (s, t) in the form

$$x = s + \sum_{j=1}^M \varepsilon_j x_{1j}(s, t) \quad (4)$$

$$y = t + \sum_{j=1}^M \varepsilon_j y_{1j}(s, t)$$

where

$$x_{1j}(s, t) = \sum_{i=1}^N \delta x_i x_{1ji}(s, t) \quad (5)$$

$$y_{1j}(s, t) = \sum_{i=1}^N \delta y_i y_{1ji}(s, t)$$

and $\varepsilon_j \delta x_i$, $\varepsilon_j \delta y_i$ represent individual displacements due to perturbation of the q_j^{th} parameter of the N strained points, and $x_{1ji}(s, t)$, $y_{1ji}(s, t)$ are straining functions associated with each of the N strained points. Introducing the strained coordinate Equations (4) and (5) into the expansion formulation leaves the zero-order result in Equation (3) unchanged, but results in a change of the following form for the j^{th} perturbation

$$L_1[\Phi_{1j}] + L_2[\Phi_0] + \frac{\partial}{\partial q_j} L[\Phi_0] = 0 \quad (6)$$

Here the operators are understood to be expressed in terms of the strained (s, t) coordinates, and the additional operator L_{2j} arises specifically from displacement of the strained points. In references 3 and 4, specific expressions for L_{2j} are provided for selected perturbations involving transonic small-disturbance and full-potential equation formulations. The essential point, however, with regard to perturbation Equation (6) expressed in strained coordinates is that it remains valid as before for a unit perturbation and independent of ϵ_j .

In employing the correction method, Equation (6) for the j th unit perturbation is solved by taking the difference between two solutions obtained by the full nonlinear procedure after appropriately straining the coordinates. If we designate the solutions for some arbitrary dependent flow quantity Q as base Q_0 and calibration Q_{cj} , respectively, of the varied independent parameter q_j , we have for the predicted flow at some new parameter value q_j

$$Q(x, y) = Q_0(s, t) + \sum_{j=1}^M \epsilon_j Q_{1j}(s, t) + \dots \quad (7)$$

where

$$Q_{1j} = \frac{Q_{cj}(\bar{x}_j, \bar{y}_j) - Q_0(s, t)}{\bar{\epsilon}_j}$$

$$\bar{x}_j = s + \sum_{i=1}^N \bar{\epsilon}_j \delta x_i x_{1i}(s, t)$$

$$\bar{y}_j = t + \sum_{i=1}^N \bar{\epsilon}_j \delta y_i y_{1i}(s, t)$$

$$x = s + \sum_{j=1}^M \frac{\epsilon_j}{\bar{\epsilon}_j} (\bar{x}_j - s)$$

$$y = t + \sum_{j=1}^M \frac{\epsilon_j}{\bar{\epsilon}_j} (\bar{y}_j - t)$$

(Continued)

[Eq. (8) Continued]

$$\left. \begin{aligned} \bar{\varepsilon}_j &= q_{c_j} - q_{o_j} \\ \varepsilon_j &= q_j - q_{o_j} \end{aligned} \right\} \quad (8)$$

In the following section, applications of the correction procedure are made to predict surface properties. Also provided is the particular form of the straining functions Equation (5) found to be most effective in those applications.

2.3 Application to Surface Properties

For the current applications, we have employed coordinate straining with the correction method to predict distributions of surface properties for simultaneous multiple-parameter perturbations of aerodynamic flows. In that instance where flow properties are required along some contour, the strained-coordinate solutions can be represented by

$$Q(x; \varepsilon) \sim Q_0(s) + \sum_{j=1}^M \varepsilon_j Q_{1j}(s) + \dots \quad (9)$$

$$x \sim s + \sum_{j=1}^M \varepsilon_j x_{1j}(s) + \dots \quad (10)$$

where x is the independent variable measuring distance along the contour or a convenient projection of that distance, s is the strained coordinate, and ε_j a small parameter representing the change in one of M flow or geometrical variables which we wish to vary simultaneously.

In order to determine the first-order corrections $Q_{1j}(s)$, we require one base and M calibration solutions in which the calibration solutions are determined by varying each of the M arbitrary independent parameters q_j by some nominal amount from the base flow value while keeping the others fixed at their base values.

In this way, the first-order corrections $Q_{1j}(s)$ can be determined as

$$Q_{1j}(s) = \frac{Q_{cj}(\bar{x}_j) - Q_o(s)}{\bar{\varepsilon}_j} \quad (11)$$

where Q_{cj} is the calibration solution corresponding to changing the j^{th} parameter to a new value q_{cj} , \bar{x}_j is the strained coordinate pertaining to the Q_{cj} calibration solution, and $\bar{\varepsilon}_j = q_{cj} - q_{oj}$ represents the change in the q_j parameter from its base flow value. If we now desire to keep invariant during the perturbation process a total of N points corresponding to discontinuities or high-gradient maxima, we can represent the solution by

$$Q(x, \varepsilon_j) = Q_o(s) + \sum_{j=1}^M \varepsilon_j Q_{1j}(s) \quad (12)$$

where $Q_{1j}(s)$ is given above and

$$\bar{x}_j = s + \sum_{i=1}^N \bar{\varepsilon}_j \delta x_i x_{1i}(s) \quad (13)$$

$$x = s + \sum_{i=1}^N \varepsilon_j \delta x_i x_{1i}(s) \quad (14)$$

$$\bar{\varepsilon}_j = q_{cj} - q_{oj} \quad (15)$$

$$\varepsilon_j = q_j - q_{oj} \quad (16)$$

$$\bar{\varepsilon}_j \delta x_i = (x_i^c - x_i^o)_j \quad (17)$$

$$\varepsilon_j \delta x_i = \frac{\varepsilon_j}{\bar{\varepsilon}_j} (x_i^c - x_i^o)_j \quad (18)$$

Here $\bar{\varepsilon}_j \delta x_i$ given in Equation (17) represents the displacement of the i^{th} invariant point in the j^{th} calibration solution from its base flow location due to the selected change $\bar{\varepsilon}_j$ in the q_j

parameter given by Equation (15), $\epsilon_j \delta x_i$ given in Equation (18) represents the predicted displacement of the i^{th} invariant point from its base flow location due to the desired change ϵ_j in the q_j parameter given by Equation (16), and $x_{l_i}(s)$ is a unit-order straining function having the property that

$$x_{l_i}(x_k^0) = \begin{cases} 1 & k = i \\ 0 & k \neq i \end{cases} \quad (19)$$

which assures alignment of the i^{th} invariant point between the base and calibration solutions.

In addition to the single condition Equation (19) on the straining function, it may be convenient or necessary to impose additional conditions at other locations along the contour. For example, it is usually necessary to hold invariant the end points along the contour, as well as to require that the straining vanish in a particular fashion in those locations. All of these conditions, however, do not serve to determine the straining uniquely. The nonuniqueness of the straining, nevertheless, can often be turned to advantage, either by selecting particularly simple classes of straining functions or by requiring the straining to satisfy further constraints convenient for a particular application.

The fact of nonuniqueness of straining function, however, raises a further question of the dependence of the final perturbation-predicted result on choice of straining function. An initial example of the effect of employing two different straining functions for a strongly supercritical flow was provided in reference 3, and in reference 1 a detailed examination was made of the dependence of perturbation results on several classes of different straining functions. Although it can be demonstrated (ref. 8) that the final perturbation-predicted result obtained when employing strained coordinates is formally independent of the particular straining function used--provided that the straining function moves the invariant points to the proper locations--the results of reference 1 demonstrate that, under certain conditions, particular classes of straining functions can induce spurious perturbation results. The underlying reason is that, while the perturbation-predicted results at and in the vicinity of invariant points are independent of the choice of straining function (provided invariant point locations are preserved), some classes of straining functions have the undesirable property of producing unwanted straining in certain regions removed from the invariant points. The correction for this deficiency, which was found in reference 1 and has proven effective in all case studies undertaken, is to employ linear piecewise-continuous straining functions. This

both preserves the accuracy of the perturbation results in the vicinity of the invariant points, and introduces no excessive straining in regions removed from those locations.

For linear piecewise-continuous straining functions, the functional forms of the straining can be compactly written.

$$\bar{x}_j = s + \left\{ \frac{x_{i+1}^o - s}{x_{i+1}^o - x_i^o} \cdot (x_i^c - x_i^o) \right. \\ \left. + \frac{s - x_i^o}{x_{i+1}^o - x_i^o} \cdot (x_{i+1}^c - x_{i+1}^o) \right\} H(x_{i+1}^o - s) \cdot H(s - x_i^o) \quad (20)$$

where H denotes the Heaviside step function. As discussed above, it is usually necessary to hold invariant both of the end points along the contour in addition to the points corresponding to discontinuities or high-gradient maxima. Consequently, for the results reported here, the array of invariant points in the base and calibration solutions have been taken as

$$x_i^o = \{0, x_1^o, x_2^o, \dots, x_n^o, 1\} \\ x_{i,j}^c = \{0, x_{1,j}^c, x_{2,j}^c, \dots, x_{n,j}^c, 1\} \quad (21)$$

where the contour length has been normalized to unity and where n is the number of invariant points along the blade contour exclusive of the end points.

3. RESULTS

Because the ultimate utility of the perturbation methods being developed under this investigation is in engineering design or parametric analysis, the two primary objectives of the current study were to develop the simultaneous multiple-parameter capability of the method and then to examine the accuracy and range of validity of the multiple-parameter method in situations characteristic of that environment. Toward that end, we have tested the method in a series of problems involving simultaneous multiple-parameter variations of both flow and geometric quantities. As with the testing of the single-parameter method (ref. 1), emphasis was placed on transonic flows past both isolated blades and compressor cascades that are strongly supercritical and exhibit large surface shock movement over the parametric range studied. Additionally, we have coupled the multiple-parameter method with an optimization procedure to test the method's ability to perform in an actual design environment. These preliminary case studies of the combined perturbation/optimization method actually resulted in the most demanding tests of the perturbation method undertaken to date for observing its ability to work accurately under extreme interpolation/extrapolation conditions.

3.1 Simultaneous Multiple-Parameter Perturbations

In Figure 1, we present a comparison of results for the simultaneous perturbation of thickness ratio and oncoming Mach number of highly-supercritical flows past a series of isolated NACA four-digit (00XX) blade profiles. The base flow chosen for these results is at $M_\infty = 0.820$ and $\tau = 0.120$, and is indicated on both plots shown in Figure 1 as the dashed line. Those results were obtained by solving the full-potential equation based on the finite-difference relaxation approximate-factorization method of reference 9. The body-fitted mesh employed had 75 points on both upper and lower surfaces. The calibration solution selected to account for Mach number changes is at $M_\infty = 0.800$ and $\tau = 0.120$, and is displayed as the dotted line in the plot on the left; while the calibration flow selected to account for thickness-ratio changes is at $M_\infty = 0.820$ and $\tau = 0.110$ and is displayed as the dotted line in the plot on the right. The open circles represent the perturbation-predicted solution for $M_\infty = 0.790$ and $\tau = 0.115$, which is a parameter extrapolation in M_∞ and interpolation in τ . Those results are meant to be compared with the 'exact' nonlinear results which is indicated as the solid line. We note that the indicated results for base, perturbation, and exact nonlinear solution in both plots in Figure 1 are the same; the reason

for presenting two plots is to indicate clearly the separation between the base, the two calibration solutions, and the perturbation-predicted result. The straining employed is linear piecewise-continuous [see Eq. (20)], with leading and trailing edge and shock point held invariant. The shock point locations for the base and calibration flows for this example, as well as for all the multiple-parameter perturbation results presented here, were determined as the point where the pressure coefficient passed through critical with comprehensive gradient.

With regard to the results, we note that the comparison between the perturbation-predicted and the exact nonlinear result is, as in the case of single-parameter perturbation of these flows (see Figs. 4 to 6, ref. 1), extremely good, in particular in the region of the shock. The multiple-parameter perturbation result is able to accurately predict both shock location and the critical post-shock expansion behavior. Results for the region from the stagnation point to points just ahead of the shock are essentially identical to the exact nonlinear solution, as are results aft of the post-shock region. We note that the particular parameter values of $(M_\infty, \tau) = (0.790, 0.115)$ selected for the prediction solution represent reasonably substantial extrapolations and interpolations from the base and calibration values. Nevertheless, the perturbation method is able to treat simultaneous parameter variations over this range accurately.

Figure 2 presents analogous three-parameter perturbation results when angle-of-attack variations are included for the flows shown in Figure 1. Here, the base flow selected is at $\alpha = 0.2^\circ$, $M_\infty = 0.800$, $\tau = 0.110$, and is indicated in all of the three plots provided as the dashed line. The calibration flow to account for angle-of-attack change is at $\alpha = 0.25^\circ$ at the same (M_∞, τ) as the base flow, and is displayed as the dotted curve in the plot on the upper left. The corresponding calibration flow to account for Mach number change is at $M_\infty = 0.810$ at the same (τ, α) as the base flow, and is displayed in the upper right plot; while the calibration flow for thickness-ratio change is at $\tau = 0.115$ at the same (M_∞, α) as the base flow, and shown in the lower plot. The predicted result is for parameter values of $\alpha = 0.3^\circ$, $M_\infty = 0.820$, $\tau = 0.100$ and again represents substantial extrapolations of all three parameters, as can be observed in Figure 2 from the relative differences between the base and calibration flows. The reason for selecting such small angles-of-attack for these flows was to preserve the shock wave on the lower surface, and thereby create a set of multiple-shock flows which were highly sensitive to parameter changes. The comparisons between perturbation and exact nonlinear results for this case is again extremely good, with the prediction of both the locations of the shocks on the upper and lower surface given very well, as well as the pressure distributions in the regions immediately ahead and behind those shocks. For these results, linear

piecewise-continuous straining was employed with the invariant points being the lower surface trailing edge, lower surface shock, stagnation point, upper surface shock, and upper surface trailing edge.

The final multiple-parameter perturbation result is provided in Figure 3 for a four-parameter perturbation of strongly-supercritical full-potential flows past a cascade of blades having NACA four-digit profiles. The base flow is for an oncoming Mach number of $M_\infty = 0.780$, thickness-ratio $\tau = 0.110$, gap-to-chord ratio $t = 3.2$, and oncoming inflow angle $\alpha = 0.3^\circ$, and is indicated on the four plots as the dashed line. Those results were obtained using the full-potential equation finite-area relaxation procedure of reference 10. The four calibration solutions to account for changes in the four varied parameters are provided in the four plots shown where the individual values of the calibration parameter varied are also indicated. Thus, the calibration solution for Mach number change is at $M_\infty = 0.790$ with (τ, t, α) at the base flow values, and is indicated as the dotted result in the plot at the upper left. Corresponding results for the other three parameters are shown in the remaining plots. The comparison of the predicted and exact nonlinear results are for parameter values of $M_\infty = 0.785$, $\tau = 0.115$, $t = 3.1$, $\alpha = 0.4^\circ$. This particular set of flows was again selected because of the presence of multiple-shocks and high sensitivity to parameter change. As with the previous results for two- and three simultaneous parameter variations, we note that the perturbation predictions are once again remarkably accurate. The perturbation method is able both to track the location of the upper and lower surface shocks, as well as to predict the pressure characteristics in the pre-shock and post-shock regions.

3.2 Preliminary Application of Combined Multiple-Parameter Perturbation Method With Optimization Procedures For Blade Design Applications

The ultimate utility of the perturbation methods developed and evaluated here lies in their application to problems involving the high-frequency use of computational codes to determine a large number of related nonlinear flow solutions. In order to test the capability of the approximation method to work effectively in such practical applications, we have combined the method with the CONMIN optimization procedure (ref. 11) and have then made several preliminary case studies of the combination on isolated blade and compressor blade design/optimization problems. The objectives of these initial applications were to examine the feasibility and potential computational savings of the combined approximation/optimization procedure for some typical design problems, and to determine the accuracy of the perturbation-predicted optimization results.

The particular isolated blade design optimization problems selected for study involved the alteration of a baseline profile shape by adding to the baseline profile a set of shape functions according to the relation

$$Z(x) = Z_0(x) + \sum_{i=1}^M (DV_i - 1) F_i(x) \quad (22)$$

where Z_0 are the ordinates of the baseline profiles, F_i are the shape functions, and the coefficients DV_i are the design variables whose values are determined by the optimization program as a result of a search through design-variable solution space to achieve a desired design improvement. Here for convenience we have chosen the coefficients of F_i to be $(DV_i - 1)$ rather than DV_i . The general class of geometric shape functions employed here, and which have been found to be successful in previous applications involving optimization of supercritical airfoil sections (ref. 12), consists of exponential decay functions and sine functions. These are of the general form $(1 - x) \cdot x^p / e^{q_i x}$ and $\sin(\pi x^{r_i})^n$, where the exponents p , q , r , and n are selected to provide a desired maximum at a particular chordwise location. The exponential functions are generally employed to provide adjustments near the leading edge, while the sine functions are used to provide maximum ordinate changes at particular chordwise stations. Illustrations of the chordwise variation of typical members of these shape functions are provided in Figure 4, and it can be seen that these functions smoothly concentrate ordinate thickness at selected locations.

For the initial application of the combined perturbation/optimization method, we have examined subcritical flow at $M_\infty = 0.10$ and $\alpha = 5^\circ$ past a modified NACA 64A007 profile involving the nine profile shape functions

$$\begin{aligned} F_i &= 6(1 - x)x^{p_i}/e^{q_i x} \quad i = 1, 2 \\ F_i &= \sin(\pi x^{r_i})^2 \quad i = 3, 9 \end{aligned} \quad (23)$$

where $(p_1, q_1) = (0.5, 15)$, $(p_2, q_2) = (0.25, 10)$, and $r_i = (0.37, 0.50, 0.66, 0.87, 1.16, 1.61, 2.41)$. The exponential functions achieve their maxima within 5% of chord, while those for the sine functions are at (15%, 25%, 35%, 45%, 55%, 65%, 75%) of chord.

A strategy which has proved convenient for performing optimization studies involving aerodynamic performance parameters (ref. 12) has been to recontour the profile shape so as to tailor the surface pressure distribution to conform to

a desired distribution. This type of objective provides local control over the basic aerodynamic surface flow property of importance, and provides a means of attempting to achieve aft pressure gradients sufficiently weak to avoid separation. An important corollary advantage of using such an objective is that viscous separation can be minimized. This allows use of an inviscid aerodynamic flow solver in the optimization process rather than a much more computationally-expensive viscous solver, and assures that the optimization result thus obtained at the inviscid level is representative of the actual flow.

Consequently, for this initial case study the overall performance objective was, through modification of the surface contour, to tailor the pressure distribution along a portion of the upper surface so as to conform to a desired distribution. In particular, it was desired to minimize both the peaky behavior near the leading edge and the compressive gradient on the aft portion of the upper surface which existed at $M_\infty = 0.10$ and $\alpha = 5^\circ$ on the NACA 64A007 baseline profile. This is illustrated schematically in Figure 5. The objective function was taken as the minimization of the mean squared error between the predicted and desired surface pressure distribution, i.e.,

$$OBJ = \sum_{k=1}^K [C_{P_{\text{predicted}}}(x_k) - C_{P_{\text{desired}}}(x_k)]^2 \quad (24)$$

where K represents the number of chordwise locations x_k where desired and calculated surface pressures are compared.

Recall that in order to initiate the perturbation procedure in situations involving the simultaneous variation of M individual parameters from a baseline point, a matrix of M calibration solutions are required, each representing the solution change for a separate variation of each of the M parameters from its baseline value. Because optimum, or sometimes even typical, stepsizes for a particular optimization problem would not generally be known a priori, one of the primary goals of these initial studies was the demonstration that the perturbation method was capable of working effectively even under severe conditions imposed by a poorly-selected initial calibration solution matrix. This was accomplished by examining the sensitivity and accuracy of the perturbation predicted optimization results as a function of the initial design variable stepsizes of the calibration solution matrix.

The overall strategy, then, consisted of: (1) employing the perturbation method, based on some initial matrix of nonlinear aerodynamic solutions determined by an independent variation of each design variable, to provide all of the subsequent nonlinear aerodynamic solutions required by the optimization method for

searches through design space, and (2) comparing the final perturbation-predicted optimization results for final design variable and objective function values with those determined by using the full nonlinear aerodynamic solver throughout. Figure 6 shows the results of such a sensitivity study, and indicates that even under extreme test case conditions caused by deliberately-selected poor choices of design variable stepsizes, the perturbation method performs exceptionally well, never breaking down or yielding spurious results. Indicated on the plots are the final optimized design variable values after five search cycles as predicted by the perturbation method (●) for four different choices of the initial stepsize for the design variables, i.e., $\delta DV_i = (0.05, 0.02, 0.01, 0.001)$. Also shown are the corresponding final design variable values predicted when employing the nonlinear aerodynamic solver (ref. 13) throughout (○). As can be seen, for the extreme interpolation case $\delta DV_i = 0.05$, except for design variables 3 and 5, the perturbation prediction compares very favorably with the full nonlinear result. For $\delta DV_i = 0.02$, a more reasonable stepsize choice, the comparison of the perturbation result is quite good for all the design variables, while for $\delta DV_i = 0.01$, the perturbation prediction is essentially identical to the full nonlinear aerodynamic result. As a final illustration of the behavior of the perturbation method under extreme extrapolation conditions, the lower right-hand plot displays the perturbation predictions for $\delta DV_i = 0.001$. We note that for several of the design variables, the extrapolation range is of the order of 25 times the initial stepsize; yet, the perturbation predictions are quite reasonable and not spurious, which is remarkable and indicative of the robustness of the procedure.

Finally, we point out that all four of the perturbation-predicted results illustrated in Figure 6 are satisfactory in terms of the final objective function value obtained. These values are illustrated on the right of each of the plots. Provided for comparison are the initial (○ Initial) and final (○ Final) values obtained when using the nonlinear aerodynamic solver throughout. The value of the objective function evaluated at the final design variable point when using the perturbation method is indicated by the solid circle (●). However, the objective function value of real interest is the result indicated by the solid square (■) which represents that obtained by running the nonlinear aerodynamic solver at the perturbation-predicted final design point, and then using that solution to evaluate the objective function. This provides the overall ultimate check of the perturbation-predicted result. As can be seen from Figure 6, those results lie essentially on top of the final objective function result (○ Final) obtained when using the nonlinear aerodynamic solver throughout.

The computational savings attained for this application are shown in Figure 7. Here, a comparison of the computational work versus reduction in objective function per optimization

cycle is provided when using the perturbation procedure (●) and when not using it but employing the nonlinear aerodynamic code (○) for each flow solution required by the optimizer. As can be seen, the computational time required for both the perturbation method and when using the full nonlinear aerodynamic solver throughout are the same for the first cycle, since both require a matrix of $M + 1$ ($M=9$ for this example) nonlinear aerodynamic solutions. After that, the perturbation-predicted results required essentially no computational time for cycles 2 through 5, and then a slight amount for the one additional call to the aerodynamic solver for the final check calculation (■). The reduction in the ratio of final to initial objective function is $\text{OBJ}/(\text{OBJ})_i = 0.22$ and required approximately 20 CPU seconds on the CDC 7600. In comparison, the result when not employing the perturbation method required approximately 80 CPU seconds for the same reduction in objective function, indicating the perturbation method is able to save 75% of the computational work in this example.

Similar testing of the performance of the perturbation method for supercritical situations has also been carried out and has demonstrated a corresponding capability and potential computational savings. Because of the greater sensitivity of these shocked flows, two separate case studies were carried out in depth. Both of these involved recontouring of the upper surface of a NACA 0015 profile operating at the supercritical conditions of $M_\infty = 0.55$ and $\alpha = 6.7^\circ$, and employed four design variables related to the shape functions.

$$F_i = \sin(\pi x^{q_i})^3 \quad i = 1, 4 \quad (25)$$

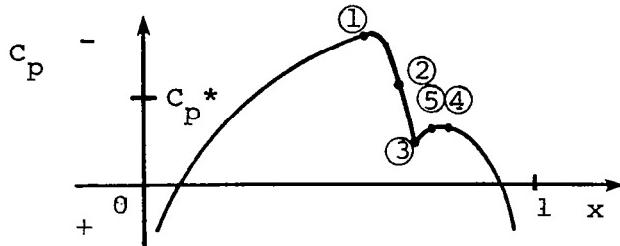
with $q_i = (0.301, 0.431, 0.576, 0.756)$. These functions have maxima at (10%, 20%, 30%, 40%) of chord. For the first of these supercritical studies, the objective function was chosen to be the drag coefficient squared, i.e.

$$\text{OBJ} = CD^2$$

For this problem, an evaluation of the accuracy of the perturbation-predicted optimization results as a function of initial calibration solution matrix was also made. The results of this study are provided in Figure 8, which displays the results of the perturbation-predicted final design variables (●) after eight search cycles for three different choices of the initial stepsize for the design variables, i.e., $\delta DV_i = (0.001, 0.002, 0.004)$. Also shown are the corresponding final design variable values predicted when not using the perturbation method but employing the nonlinear aerodynamic solver (ref. 13) throughout (○).

Because drag minimization, particularly at supercritical conditions, is an extremely sensitive optimization problem, this study provides one of the most severe tests of the perturbation procedure in a design optimization environment. This is so because the accurate prediction of drag at supercritical speeds depends almost entirely on the resolution of the flow behavior at and in the vicinity of the shock waves present on the surface of the profile. Hence, what is ultimately under evaluation in this example case study is the ability of the perturbation method to predict, under extreme extrapolation conditions and with simultaneous multiple-parameter perturbations, the location and strength of all surface shock waves and the flow behavior in the pre- and post-shock regions.

The most important results to emerge from the calculations involved in this case study were the discovery of a particular deficiency of the perturbation method in this regard, and the subsequent development of the means to improve the accuracy of the perturbation predictions in shock regions and other high gradient regions under extreme extrapolation conditions. The improvement in the basic procedure developed to meet these requirements consists of employing additional invariant points in those high gradient locations. For example, it was found that by characterizing a shock which has a post-shock expansion region, as sketched below,



with five invariant points--which correspond to: ① pre-shock minimum pressure, ② maximum gradient point, ③ post-shock maximum pressure, ④ post-shock minimum expansion pressure location, and ⑤ point of inflection between points ③ and ④ --rather than just the one point corresponding to the critical pressure location, which was standardly done in the past; that significantly improved perturbation results are obtained in the shock region for extreme solution extrapolations. This five invariant point characterization of the shock has been employed in determining the results of the two supercritical case studies reported here.

With regard to the drag minimization results indicated in Figure 8, we note that when selecting an initial design variable stepsize of $\delta DV_i = 0.001$, the agreement between the final perturbation-predicted design variable values and the exact nonlinear result is reasonable. The solution extrapolation indicated for design variables 2 and 3 are of the order of 5 times the initial design variable stepsize, but the perturbation method does not break down or provide spurious results for these strongly supercritical flows. We note that the optimization procedure working with the perturbation method is able to drive the perturbation-predicted drag ($\bullet P$) to essentially zero. Although the final check of the perturbation-predicted design using the nonlinear solver ($\blacksquare A$) indicates a drag coefficient of 0.005, that represents nevertheless almost a factor of 3 in drag reduction from the baseline configuration.

When a somewhat more reasonable initial design variable stepsize of $\delta DV_i = 0.002$ is used, which reduces the solution extrapolations over that for $\delta DV_i = 0.001$, we note that the perturbation results closely approach those obtained when using the nonlinear aerodynamic solver throughout. The final drag value based on the perturbation design ($\blacksquare A$) is also improved, displaying over a factor of 5 in drag reduction. The final perturbation result obtained for an initial stepsize $\delta DV_i = 0.004$ and illustrated in the bottom plot displays almost exact agreement with the full nonlinear aerodynamic result. This particular choice of stepsize is the most reasonable of the three shown, since the design variable solution range to be searched by the optimizer requires moderate interpolation/extrapolations from the calibration solution matrix. The final drag value ($\blacksquare A$) indicates over a factor of 5 drag reduction, and is very close to the final full nonlinear aerodynamic (\circ Final) result.

The computational savings obtained for this application are indicated in Figure 9. Here a comparison of the computational work versus reduction in objective function per optimization search cycle is given for eight search cycles when using the approximation procedure (\bullet) and for four cycles when using instead the full aerodynamic solver (\circ). We note for this example that when using the perturbation method, the optimization procedure requires approximately 40 CPU seconds for search cycles 2 through 8. This is in contrast to the essentially zero time needed for search cycles in the pressure distribution tailoring case study presented in Figure 7. The reason for the additional time in this minimization case study is that in order to evaluate the drag at the points in the design solution space required by the optimizer, additional geometry calculations are needed to determine the new surface slopes. Consequently, even though the perturbation procedure provides the means to determine the new pressure distributions at essentially no computational cost, because surface pressure integrations are required to evaluate the drag, calls to the geometry portion of the aerodynamic code are necessary to

determine the new profile shapes. These geometry calculations cause the CPU time increase indicated. Finally, another 20 CPU seconds are needed for the final aerodynamic check solution (■), totaling 120 CPU seconds. That total represents just under a 60% computational savings of the 4 search cycle CPU time required when not employing the perturbation method.

The final supercritical case study presented in Figure 10 involved tailoring of the surface pressure distribution as done in the previous subcritical case, but this time specifically concentrating on alleviating the sharp gradient at the shock. It was specified in the following fashion. Using the pressure distribution from the optimized profile of the previous drag minimization case study, we returned to the original NACA 0015 baseline profile. Then, with that previous pressure distribution as the desired objective, by using the optimization method we attempted to reproduce the same optimized profile that resulted from the drag minimization problem. What resulted from these calculations was a demonstration that the perturbation method is capable in certain cases, of not only providing an enormous savings in computational cost but also an improved optimization result. In Figure 10, we provide the results of the sensitivity study of the perturbation method as a function of the initial design variable stepsize of the calibration solution matrix. In the plot on the left, the comparison is provided of the perturbation-predicted final design variables (●) based on an initial design variable stepsize $\delta DV_i = 0.001$ together with that obtained when using the full nonlinear aerodynamic solver throughout (△). Also shown is the 'optimum' full aerodynamic result (○) obtained from the drag minimization case study which is the result that is sought to be reproduced. What the results indicate is that the prediction obtained by not using the perturbation method is inferior to that obtained when employing it. Also, we note the large extrapolations involved in this perturbation result, i.e., about 4 times or 400% of the initial stepsize for design variables 1 and 2. These results are emphasized even more in the plot on the right which shows the corresponding perturbation results when using a more reasonable initial design variable stepsize of $\delta DV_i = 0.002$ to define the calibration solution matrix. In this instance, the perturbation prediction is essentially identical to the optimum result. We note that comparisons of the objective function reduction, shown on the scales to the right of the plots, display a reduction in objective function to almost zero for the perturbation result, while only an essentially 50% reduction for the full aerodynamic result.

This reduction in objective function is emphasized in Figure 11, which displays the comparison of computational work and objective function reduction per optimization cycle for the perturbation procedure and the full nonlinear aerodynamic result for this case study. Here, we see clearly that the perturbation method is able to drive the objective function to

essentially zero, while the full aerodynamic procedure becomes fixed in a local minimum and is only able to reduce the objective function to 50% of its initial value. If we had carried the full aerodynamic result to eight optimization cycles, as was done with the perturbation result, the time savings would have been over an order of magnitude.

This result demonstrates that it is possible in certain instances for the perturbation method to provide not only an enormous savings in computational cost but also an improved optimization result. The latter is undoubtedly accomplished by the selection of a reasonable initial calibration solution matrix which permits the optimization method an enhanced rather than local view of the design solution space, thereby avoiding shallow local minimas in favor of a more global minima.

The final application, which has been carried out as far as possible in this phase, was directed toward laying the foundations of a practical turbomachinery blade design/optimization procedure coupled with the perturbation method. This preliminary version is based on TSONIC blade-to-blade solutions (ref. 14) with generalized circular-arc blade geometry routines BLADE (ref. 15) describing the blade profile, and employs the COPES/CONMIN optimization procedure (ref. 16).

The initial case study for the combined procedure involved, as a design objective, the minimization of the velocity diffusion on the blade suction surface, $q_{\max, \text{suction}}$, i.e.

$$\text{OBJ} = \frac{q_{\max, \text{suction}}}{q_{\text{avg}, \text{exit}}} \quad (27)$$

where $q_{\text{avg}, \text{exit}}$ is the average exit velocity in the freestream. Six design variables were employed and correspond to the following blade geometry parameters used to characterize NASA circular arc blade section profiles (ref. 15): blade outlet camber angle, transition location between fore and aft circular arc sections, maximum thickness location, inlet to outlet turning rate ratio, maximum thickness, and radius of the leading edge circle. During the optimization process, each of the design variables was constrained to remain within certain prescribed bounds in order to prevent a physically-unrealistic blade design from occurring. Furthermore, several active side constraints were additionally imposed both to insure design of a physically realistic blade and also to achieve certain desirable flow characteristics on the blade. The active side constraints employed were: (1) maintenance of nonzero local blade thickness, (2) maintenance of low velocity diffusion on the blade pressure surface, and (3) trailing edge closure. These constraints were enforced by employing bounding criteria on the following quantities according to:

$$0. < \frac{\text{thick}(x) - d_{tc}}{d_{tc}} < 10. \quad (28)$$

$$0. < \frac{q_{\max, \text{press}}}{q_{\min, \text{press}}} < 1.3 \quad (29)$$

$$-1.0 < \frac{q_{ITE-2, \text{suction}} - q_{ITE-2, \text{press}}}{25} < 1.0 \quad (30)$$

Here, $\text{thick}(x)$ is the local blade thickness, d_{tc} is the diameter of the trailing edge circle, $q_{\max, \text{press}}$ is the maximum blade pressure surface velocity over the front half of the blade, $q_{\min, \text{press}}$ is the minimum blade pressure surface velocity over the last two-thirds of the blade, and $(q_{ITE-2, \text{suction}}, q_{ITE-2, \text{press}})$ are the third last surface velocities on the grid near the trailing edge on the suction and pressure surface, respectively.

We have successfully completed a preliminary series of calculations of the new combined PERTURB/TSONIC/BLADE/COPES/CONMIN procedure in which the accuracy and sensitivity of the perturbation method was tested as a function of choice of the initial calibration solution matrix. The initial or base values of the design variables for the baseline blade profile, and the upper and lower bounds of the design variables that were specified for this test problem were:

<u>Design Variable Number</u>	<u>Description</u>	<u>Lower Bound</u>	<u>Upper Bound</u>	<u>Initial Value</u>
1	Outlet blade camber angle-KOCR	-15.0°	0.0°	-10.0°
2	Transition loc./chord-T	0.20	0.60	0.25
3	Max. thick. loc./chord-ZM	0.20	0.55	0.45
4	Inlet/outlet turning/chord-P	0.50	4.00	1.50
5	Max, thick./chord-TMX	0.03	0.10	0.05
6	L.E. rad./chord-THLE	0.003	0.012	0.005

The results of these calculations are summarized in Table 1. There we have provided comparisons of the final design variables and objective function predicted when employing full nonlinear TSONIC solutions throughout the optimization process with

corresponding results when using the perturbation method. For the perturbation results, six different choices of the calibration solution matrix were made and are noted in the table. All the results represent converged solutions, with each calculation employing 10 optimization search cycles or less if no change in objective function should occur in three successive iterations.

Two separate results are presented for the case when the perturbation method is not employed and TSONIC solutions are used throughout the optimization search process. These were obtained on the Ames Research Center CDC-7600 and Lewis Research Center IBM 3033 employing the same TSONIC/BLADE/COPES/CONMIN code. The differences in the two final design results provides an indication of the sensitivity of this particular optimization problem with regard to choice of objective function, since the difference between these two results is due solely to the number of significant figures maintained in the respective calculations, i.e., 8 for the IBM 3033 and 14 for the CDC 7600. As is evident, there clearly is a sensitivity to search direction and final design result in this problem. We note that of the six design variables, the two which agree between the two results (ZM, TMX) have reached a limit boundary. The others have all trended in the same direction from the baseline value, but have reached somewhat different values at the final design point. We note that the objective functions have reached essentially the same level, indicating, as is often the case in such optimization problems, the existence of multiple local minimums.

The analogous results obtained employing the perturbation method with various choices of calibration solution matrix are exhibited as cases 1 to 6. We note that, with the exception of the design variable ZM which consistently moves to its upper bound regardless of the choice of calibration solution matrix, the final design variable predictions via the perturbation method basically exhibit a behavior quite similar to that obtained from the full nonlinear result when TSONIC solutions are used throughout. That is, the optimizer drives the design variables in generally the same direction from the baseline values as the full nonlinear result, but to somewhat different values. The exception to this is noted in cases 4 and 5, where the calibration solution matrices were selected such that TSONIC solution extrapolations rather than interpolations were involved for all or most of the design variables during the search procedure. In those two cases, we note further that the final objective function values obtained are somewhat inferior to those obtained in Cases 1, 2, 3, and 6, where solution interpolations were primarily involved. For those four cases, the final objective function result is almost identical to that obtained by the full nonlinear result.

The computational time needed to obtain the perturbation results in Cases 1 to 6 were 76-78 secs. of CDC 7600 CPU time per case. The benchmark full nonlinear CDC 7600 result shown in Table 1 required 644 secs. Thus, the perturbation method provides a savings of $(644-78)/644 = 88\%$ of the computational time for this example.

The primary conclusions to be drawn for this preliminary study are that the perturbation method can work effectively even for sensitively-defined optimization problems such as this and provide both meaningful final design results and large computational savings over not using the method. The choice of objective function such as was made for this case study, i.e., a point quantity located in a high-gradient region, requires that a reasonably good choice be made of the calibration solution matrix. This is so since if large solution extrapolations are required by the optimizer for minimization searches through design space, the final design result will most likely be less improved than would otherwise result if only modest solution interpolations/extrapolations were involved.

4. CONCLUSIONS AND RECOMMENDATIONS

An investigation was conducted to continue the development of perturbation procedures and associated computational codes for rapidly determining approximations to nonlinear flow solutions. The ultimate purpose is to establish a method for minimizing computational requirements associated with parametric design studies of transonic flows in turbomachines. The procedures being developed employ unit perturbations, determined from two or more nonlinear 'base' solutions which differ from one another by a nominal change in some geometry or flow parameter, to predict a family of related nonlinear solutions which can be either continuous or discontinuous. The results reported here relate to the extension of the previously-developed successful method for single-parameter perturbations: (1) to simultaneous multiple-parameter perturbations, and (2) to the preliminary application of the multiple-parameter procedure in combination with an optimization procedure and applied to blade design problems.

Calculations based on full-potential nonlinear solutions have been carried out to establish the accuracy and range of validity of the multiple-parameter capability. These involved flows past both isolated blades and compressor cascades involving simultaneous changes in both flow and geometric parameters, with attention focused on strongly supercritical situations involving large surface shock movements over the parameter ranges studied. Preliminary applications of the multiple-parameter perturbation method coupled with an optimization procedure were made for blade design problems in order to examine the capability of the method to produce accurate results in a typical design environment, and also to evaluate its potential for computational savings. Both subcritical and supercritical case studies were carried out involving multiple-design variables. Sensitivity studies were also performed to examine the accuracy dependence of the perturbation method on the choice of the initial calibration solution matrix.

Comparisons of the multiple-parameter perturbation results with the corresponding 'exact' nonlinear solutions display remarkable accuracy and indicate a large range of validity, in direct correspondence with previous results for single-parameter perturbations. The preliminary case studies of the multiple-parameter perturbation method combined with optimization procedures have clearly demonstrated the capability of the method to work accurately in a design environment where large solution extrapolations are often necessary, and have also established the methods' potential for reducing the computational work required in such applications by nearly an order of magnitude. The sensitivity studies indicate that for certain subcritical applications, the perturbation method is able to work quite accurately and effectively in spite of poor choices of the initial calibration solution matrix which require large extrapolations or interpolations. For supercritical flows, the

initial calibration matrix choice is more important, but when employing multiple invariant point clustering about high-gradient locations (shocks, stagnation points, etc.), the perturbation method predictions can nevertheless maintain high accuracy in certain supercritical situations for extrapolations as large as 4 to 5 times the parameter separation between base and calibration solutions.

Based on these results, we conclude that perturbation methods formulated on these ideas are both accurate and clearly workable in design environments, and can provide the means for substantially reducing the computational work required in such applications. We suggest the development of the combined multiple-parameter perturbation procedure and optimization methods into a robust design tool for subcritical and supercritical turbomachinery blade design.

APPENDIX A
USER'S MANUAL FOR COMPUTER PROGRAM PERTURB

A.1 INTRODUCTION

The purpose of this appendix is to describe the operation of the computer code which was developed in conjunction with the theoretical work presented in this report, and to provide sufficient detail to permit convenient use and change of the program. The program computes and plots an arbitrary flow variable on a contour surface by employing the strained-coordinate perturbation method discussed in the main text.

A description of the general operating procedure of the program is given, together with complete description of both input and output. The program is written in FORTRAN IV and has been developed on the Ames Research Center CDC 7600 computer facility. Consequently, the plot package included in this version refers to system routines at that facility. In general, the plotting software must be supplied by the user according to the requirements of his operating system. This can be accomplished directly by replacing or modifying the subroutine DRVPLT. Typical program run times for flows involving approximately 150 surface mesh locations are 1 to 3 CPU seconds. The storage requirements are 105K_B for small core memory and no large core memory.

A.2 PROGRAM DESCRIPTION

The program calculates both continuous and discontinuous nonlinear perturbation solutions which represent a multiple-parameter change in either geometry or flow conditions by employing a strained-coordinate procedure. The method utilizes unit perturbations, determined for each parameter from a previously-calculated common base solution and a calibration solution displaced from it by some reasonable change in the relevant parameter, to predict new nonlinear solutions over a range of parameter variation.

This current version of the procedure is configured to predict and plot an arbitrary flow variable (e.g., pressure coefficient) on the surface of a blade or airfoil, and can account for the motion of:

1. one or more critical points (shock points),

2. a stagnation point,
3. a maximum-suction-pressure point,

or simultaneously for any combination of these.

The program is also configured to compare the perturbation-predicted solutions with the corresponding 'exact' solutions obtained by employing the same 'expensive' computational procedure used to determine the base and calibration solutions.

The coordinate straining employed is piecewise linear with the end points and up to six interior points held invariant. At the option of the user, these additional interior points may be arbitrarily preselected, or chosen from among the minimum, maximum, and critical points automatically located by the program itself.

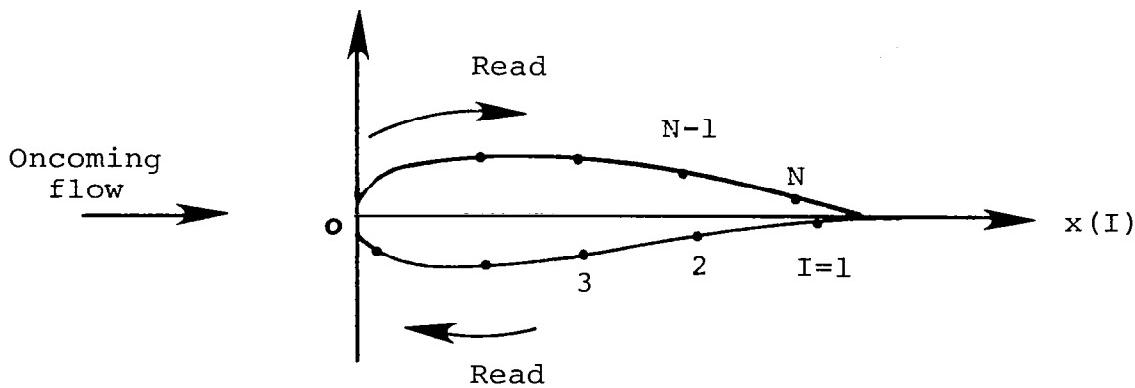
Critical or shock points are located on the basis of a user-supplied statement function defining the critical value of the dependent variable as a function of some single flow variable. The program default is with dependent variable y defined as pressure coefficient, with the independent variable being Mach number. In this case, the critical value is defined as

$$y_{\text{crit}} = C_p^* = \frac{2}{\gamma M_\infty^2} \left(\frac{2 + (\gamma - 1)M_\infty^2}{\gamma + 1} \right)^{\frac{\gamma}{\gamma-1}} - 1 \quad (\text{A-1})$$

where γ is the ratio of specific heats. If instead of surface pressure coefficient, the surface velocity distribution were used, then the value of y_{crit} would be given by

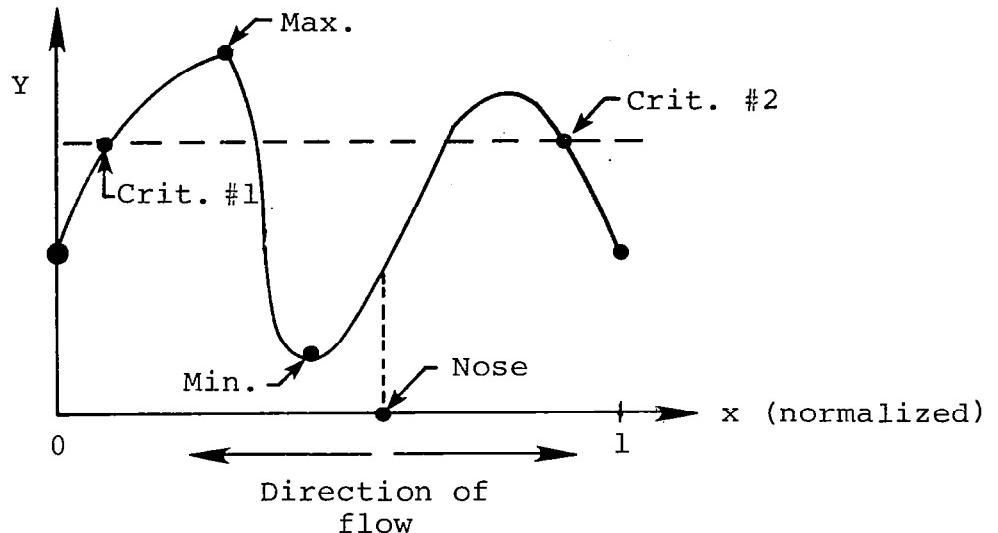
$$y_{\text{crit}} = \frac{V^*}{V_\infty} = \left(\frac{\gamma + 1}{2 + (\gamma - 1)M_\infty^2} \right)^{\frac{1}{\gamma-1}} \quad (\text{A-2})$$

Data for base, calibration, and comparison solutions (if available) are input as an array $x(I)$ of coordinates a corresponding array $y(I)$ giving the dependent variable at each coordinate location, where $1 \leq I \leq N$ and $N \leq 200$.



The leading edge is at $x = 0$; the data are read in beginning on the lower surface at the point farthest from the leading edge and proceeding clockwise around the surface as shown in the sketch. Data for the different solutions need not correspond to identical locations on the surface, except for the initial and final points, i.e., $x(1)$ and $x(N)$ must be the same for all cases. The program normalizes the x coordinates ($0 < x \leq 1$) such that $x = 0$ corresponds to $I = 1$ and $x = 1$ to $I = \bar{N}$.

The base and calibration solutions are searched for minimum, maximum, and critical points, e.g.,

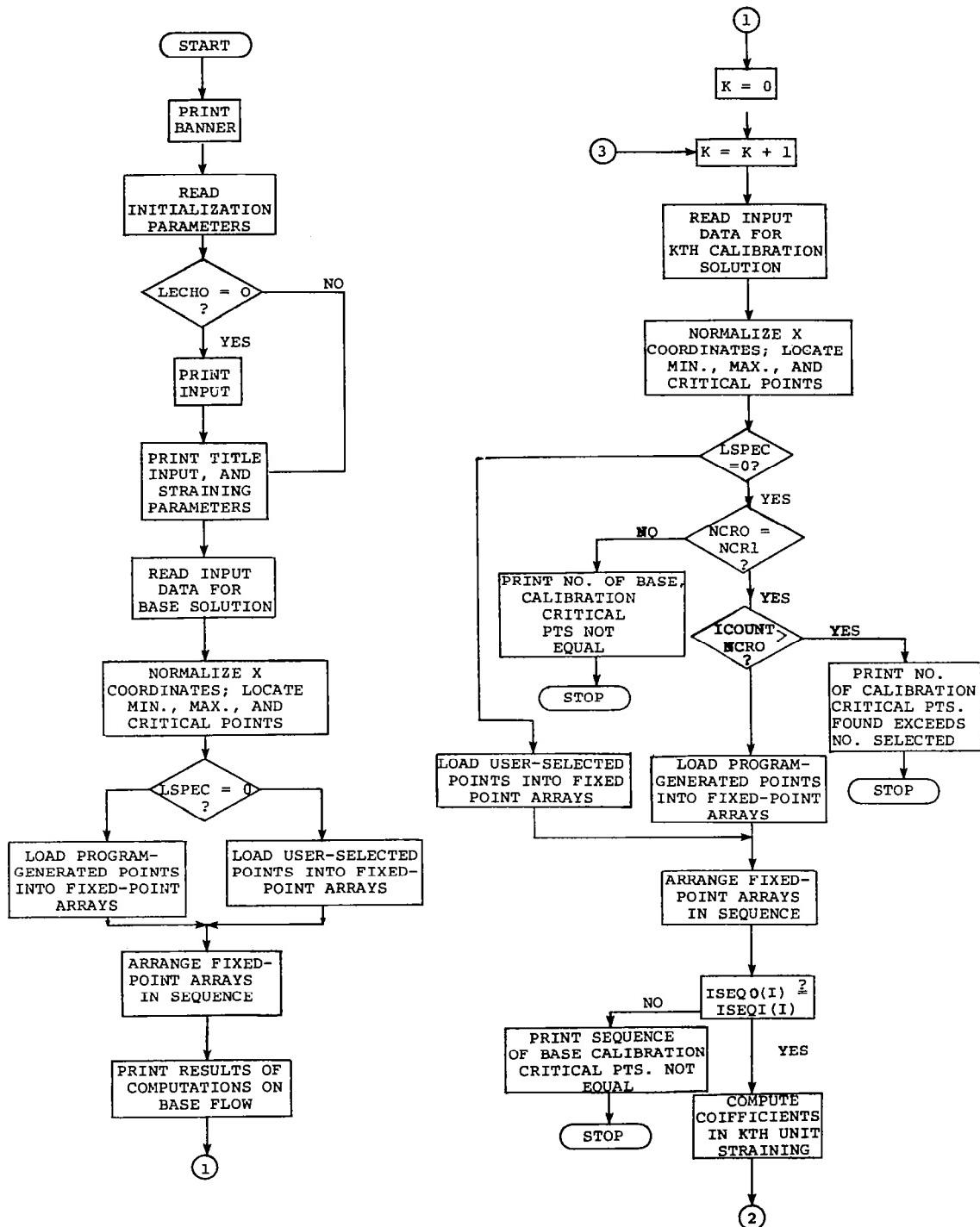


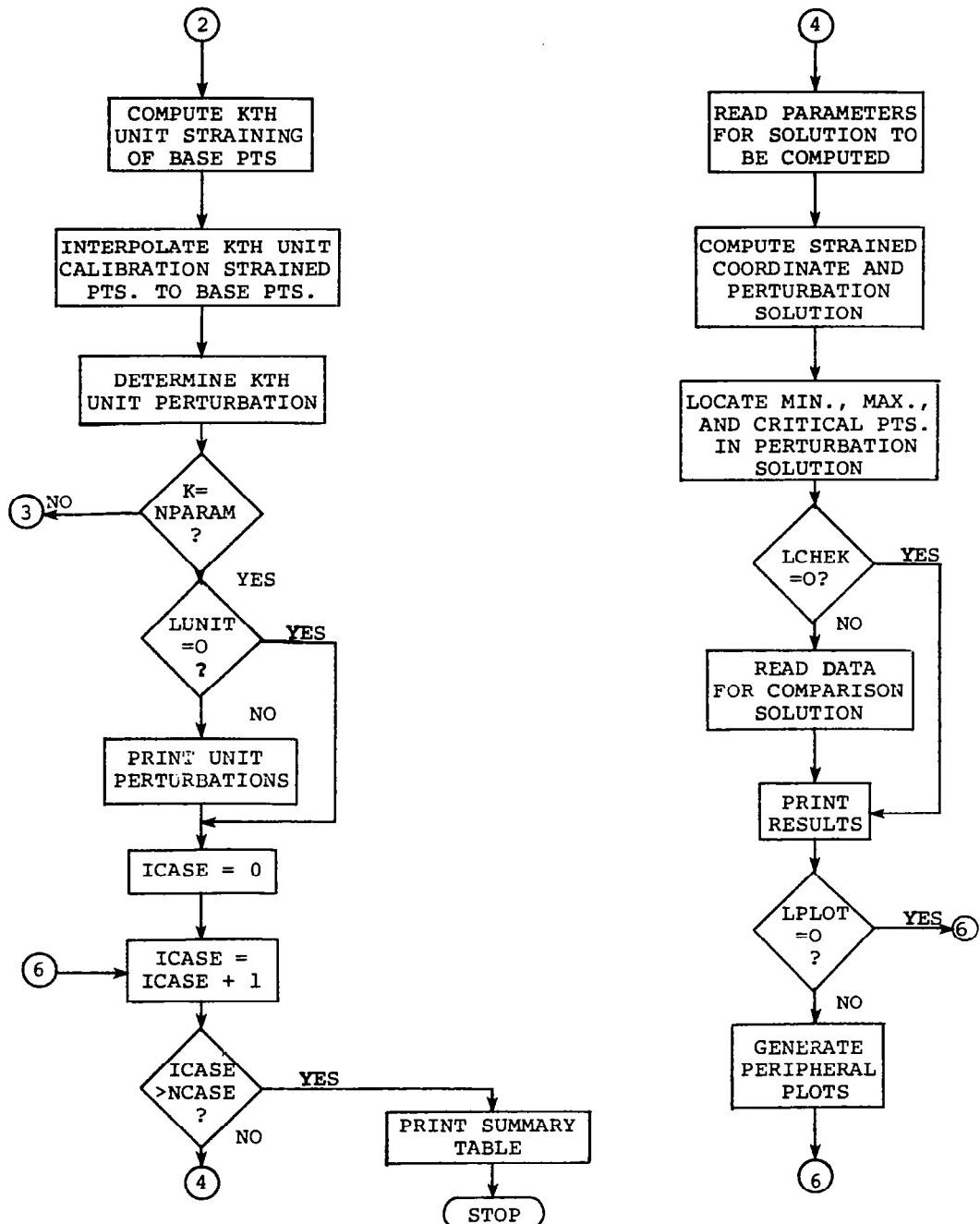
Note that the sign of dy/dx in physical coordinates is used in determining the critical points. For example, both critical points indicated on the above figure correspond to $dy/dx < 0$ in physical coordinates, since at point #1 the physical coordinate increases in the direction from right to left, whereas at point #2 it increases from left to right.

The points to be held invariant in straining are either selected from among those (1) located by the program or (2) individually specified by the user, after which the unit coordinate straining and unit perturbation are computed.

Data for the test cases is then read in and nonlinear perturbation solutions constructed from the unit perturbation.

A 3. PROGRAM FLOW CHART





A.4 DICTIONARY OF INPUT VARIABLES

A	Scaling parameter in straining procedure. A = -x(1), where x(1) is location of first data point on lower surface (see PROGRAM DESCRIPTION).
B	Scaling parameter in straining procedure. B = x(N), where x(n) is location of last data point on upper surface (see PROGRAM DESCRIPTION).
LCHEK	Specifies whether or not perturbation solution is to be checked against an exact comparison solution. A printer plot is made in either case. LCHEK = 0 ... no comparison LCHEK = 1 ... comparison
LECHO	Controls whether or not input deck is printed. LECHO = 0 ... no print LECHO = 1 ... print
LOC0(I)	Array of length 6 of which NSELCT elements are read in. Specifies subscripts of those user-specified base flow points which are to be held invariant; operational only when LSPEC = 1.
LOC1(I)	Array of length 6 of which NSELCT elements are read in. Specifies subscripts of those user-specified points in the Kth calibration solution which are to be held invariant; operational only when LSPEC = 1.
LPLOT	Specifies whether or not an additional plot by a peripheral device is to be made. Software must be supplied by user in subroutine DRVPLT. LPLOT = 0 ... No peripheral plot LPLOT = 1 ... Peripheral plot
LSELCT(I)	Array of length 6 of which NSELCT elements are read in; operational only when LSPEC = 0, and specifies nature of points to be held invariant according to the code: 1 ... minimum point held invariant 2 ... maximum point held invariant 3 ... 1st critical point held invariant 4 ... 2nd critical point held invariant 5 ... 3rd critical point held invariant 6 ... 4th critical point held invariant Note that critical point ordering is determined from order of occurrence starting at the lower surface at the point furthest from the leading edge and proceeding clock-wise around the surface (see PROGRAM DESCRIPTION).

Note that the code numbers can be assigned in any order, e.g.,

LSELCT(1) = 1	LSELCT(1) = 4
LSELCT(2) = 3	and LSELCT(2) = 1
LSELCT(3) = 4	LSELCT(3) = 3

are equivalent, both corresponding to NSELCT = 3, with the minimum, and first and second critical points held invariant.

LSPEC Controls how invariant points in straining are specified.

LSPEC = 0 ... invariant points selected from among those located by the program using the array LSELCT(I)

LSPEC = 1 ... invariant points preselected by user, using the arrays LOC0(I), LOCL(I)

LUNIT Controls whether or not unit coordinate strainings and unit perturbation(s) are printed.

LUNIT = 0 ... no output
LUNIT = 1 ... output

M0,M1,M2 Oncoming Mach numbers in base, calibration and computed perturbation solutions.

N Number of locations for which data are input for base, calibration, and comparison solutions.

NCASE Number of cases for which perturbation solutions are to be computed.

NPARAM Number of parameters perturbed.

NSELCT Number of points (in addition to end points) to be held invariant in straining; note: $1 \leq NSELCT \leq 6$.

PARNAM(K) Array of 8-character strings which identify the parameters varied. NPARAM elements of the array are read in.

Q0(K) Array of length 8 giving values of perturbation parameters in base solution. NPARAM elements of the array are read in.

Q1 Value of Kth perturbation parameter in Kth calibration solution.

Q2(K) Array of length 8 giving values of the perturbation parameters in the solution to be computed. NPARAM elements of the array are read in.

TITLE	Character string of length 80; identifies job and is printed as headline on first page of output. First nine characters are printed in upper-right- corner of banner page, and in upper-left corner of summary page.
VNAM	Character string of length 2 which symbolizes dependent variable, e.g., "CP" for pressure coeffi- cient.
XBASE(I), XCALB(I), XCHEK(I)...	Arrays of surface coordinates in base, calibration, and comparison solutions.
YBASE(I), YCALB(I), YCHEK(I)...	Arrays of dependent variables in base, calibration, and comparison solutions.

A.5 PREPARATION OF INPUT DATA

A.5.1 Description of Input

- Item 1 One card, identifies job-printed as headline on first page of output. First nine characters are printed in upper-right corner of banner page, and in upper-left corner of summary page.
- Item 2 One card, containing the parameter, LECHO
- Item 3 One card, containing the parameters N, NCASE, NPARAM, NSELCT, LSPEC, LUNIT, LCHEK, LPLOT.
- Item 4 One card, containing the parameter array (LSELCT(I), I = 1, NSELCT). This item omitted if LSPEC = 1
- Item 5 One card, containing the character string VNAM.
- Item 6 One card, containing the character strings, PARNAM(K), K = 1, NPARAM
- Item 7 One card, containing the scaling parameters A and B.
- Item 8 One card, containing the parameter MO
- Item 9 One card, containing the parameter array QO(K), K = 1, NPARAM

- Item 10 One set of J cards, where $J = 1 + \text{INT}(N/8)$, containing data for the base flow coordinates XBASE(I), $I = 1, N$
- Item 11 One set of J cards, J as above, containing data for the dependent variable in the base solution, YBASE(I), $I = 1, N$
- Item 12 One card, containing the parameter array LOC0(I), $I = 1, NSELCT$. This item omitted if LSPEC = 0
- Item 13 One card, containing the parameters M1, Q1
- Item 14 One set of J cards, J as above, containing data for the coordinates in the Kth calibration solution, XCAL(I), $I = 1, N$
- Item 15 One set of J cards, J as above, containing data for the dependent variable in the Kth calibration solution, YCALB(I), $I + 1, N$
- Item 16 One card, containing the parameter array LOC(I), $I = 1, NSELCT$. This item omitted if LSPEC = 0
- Item 17 One card, containing the parameter MC
- Item 18 One card, containing the parameter array Q2(k), $K = 1, NPARAM$
- Item 19 One set of J cards, J as above, containing data for the coordinates in the comparison solution, XCHEK(I), $I = 1, N$. This item is required only when LCHEK = 1.
- Item 20 One set of J cards, J as above, containing data for the dependent variables in the comparison solution, YCHEK(I), $I = 1, N$. This item is required only when LCHEK = 1.

NOTE: One set of items 13 through 16 is required for each of the NPARAM calibration solutions.

One set of items 17 through 20 is required for each of the NCASE solutions to be computed.

A.5.2 Format of Input Data

Item no. 1: 1 card (8A10)

Variable	Title							
Card column	10	20	30	40	50	60	70	80
Format Type								A

Item no. 2: 1 card (I5)

Variable	LECHO
Card column	5
Format type	I

Item no. 3: 1 card (16I5)

Variable	N	NCASE	NPARAM	NSELCT	LSPEC	LUNIT	LCHEK	LPLOT
Card Column	5	10	15	20	25	30	35	40
Format Type	I	I	I	I	I	I	I	I

Item no. 4: (LSPEC=0): 1 card (16I5)

Variable	LSELCT(1)	LSELCT(2)	LSELCT(3)	-----	LSELCT(NSELCT)			
Card column	5	10	15	20	25	30	35	
Format Typr	I	I	I	I	I	I	I	

Item no. 5: 1 card (2A1)

Variable	VNAM
Card column	2
Format type	A

Item no. 6 1 card (10A8)

Variable	PARNAM(1)	PARNAM(2)	PARNAM(3)	-----	PARNAM(NPARM)			
Card column	8	16	24	32	40	48	56	
Format type	A	A	A	A	A	A	A	

Item no. 7. 1 card (8F10.6)

Variable A B
 Card column 10 20
 Format type F F

Item no. 8. 1 card (8F10.6)

Variable MU
 Card column 10
 Format type F

Item no. 9. 1 card (8F10.6)

Variable Q0(1) Q0(2) Q0(3) -----
 Card column 10 20 30 40 50 60 70
 Format type F F F F F F F

Q0(NPARAM)

Item no. 10. J cards, J=1+INT(N/8), 8 values per card (8F10.6)

Variable XBASE(1) XBASE(2) XBASE(3) -----
 Card column 10 20 30 40 50 60 70 80
 Format type F F F F F F F F

Item no. 11. J cards, J as above, 8 values per card (8F10.6)

Variable YBASE(1) YBASE(2) Ybase(3) -----
 Card column 10 20 30 40 50 60 70 80
 Format type F F F F F F F F

Item no. 12. (LSPEC=1): 1 card (16I5)

Variable LOC0(1) LOC0(2) LOC0(3) -----
 Card column 5 10 15 20 25 30 35
 Format type I I I I I I I

LOC0(NSELCT)

Item no. 13. 1 card (8F10.6)

Variable

Card column

Format type

M1	Q1	
10	20	
F	F	

Item no. 14. J cards, J as above, 8 values per card (F10.6)

Variable

Card column

Format type

XCALB(1)	XCALB(2)	XCALB(3)	-----					
10	20	30	40	50	60	70	80	
F	F	F	F	F	F	F	F	

Item no. 15. J cards, J as above, 8 values per card (8F10.6)

Variable

Card column

Format type

YCALB(1)	YCALB(2)	YCALB(3)	-----					
10	20	30	40	50	60	70	80	
F	F	F	F	F	F	F	F	

Item no. 16. (LSPEC=1): 1 card (16I5)

Variable

Card column

Format type

LOC1(1)	LOC1(2)	LOC1(3)	-----	LOC1(NSELCT)				
5	10	15	20	25	30	35		
I	I	I	I	I	I	I		

Item no. 17. 1 card (8f10.6)

Variable

Card column

Format type

M2	
10	
F	

Item no. 18. 1 card (8F10.6)

Variable

Card column

Format type

Q2(1)	Q2(2)	Q2(3)	-----	Q2(NPARAM)				
10	20	30	40	50	60	70		
F	F	F	F	F	F	F		

Item no. 19 (LCHEK=1): J cards , J=1+INT(N/8), 8 values per card (8F10.6)

Variable	XCHEK(1)	XCHEK(2)	XCHEK(3)	-----				
Card column	10	20	30	40	50	60	70	80
Format type	F	F	F	F	F	F	F	F

Item no. 20. (LCHEK=1): J cards, J as above, 8 values per card (8F10.6)

Variable	YCHEK(1)	YCHEK(2)	YCHEK(3)	-----				
Card column	10	20	30	40	50	60	70	80
Format type	F	F	F	F	F	F	F	F

A.6 DESCRIPTION OF OUTPUT

The first output item consists of a banner page, and the card images of the input data, the latter only if LECHO = 1.

The second item is a page headed by the job title, listing;

1. the input parameters N, A, B and NPARAM relevant to the actual calculation;
2. the straining option selected and the classifications of the straining points selected.

The third item is the results of the computations on the base solution. These include: the Mach number MO, values of the perturbation parameters of the base solution Q0(k), K = 1, NPARAM, and the critical value of the dependent parameter, the locations of the minimum, maximum and critical points, and the locations of the invariant points. These results are then repeated for each of the NPARAM calibration solutions.

Results for unit straining of XBASE, and unit perturbations of the dependent variables are the fourth output item. This is done only if LUNIT = 1.

The fifth item (repeated for each case computed) summarizes the results of the perturbation calculation. The Mach number, the values of the perturbation parameters, and the critical values of the variable are printed first, followed by the locations of the minimum, maximum, and critical points in the perturbation solution and comparison solution (if any). Next, a legend is printed providing the maximum, minimum, and critical values of the dependent parameter and corresponding point symbols. Print symbols are also provided for the printer plot of the perturbation solutions P, comparison solution C (if any), and a common symbol (\$) when there is agreement within printer plot accuracy between the two. Then follows a table listing XBASE, YBASE, XPERT (the strained coordinate), and YPERT (the computed value of the dependent variable). If LCHEK = 1, three additional columns list XCHEK (the computed solution) at the points given by XCHEK. This allows direct numerical comparison of YPERT with XCHEK, since the values of XPERT and XCHEK will not in general coincide. A printer plot is then provided of the perturbation result, together with the comparison solution (if any).

The final item is a table which summarizes the perturbation parameters for the base, calibration, and all predictive solutions.

A.7 ERROR MESSAGES

NUMBER OF CRITICAL POINTS IN
BASE AND CALIBRATION SOLUTIONS
ARE UNEQUAL - CALCULATION ENDED

This message will be printed if critical points are specified in straining ($LSPEC = 0$) and the number of critical points in base and calibration solutions are unequal. The remedy is to avoid use of critical points in straining, or to use base and calibration solutions having equal numbers of critical points.

NUMBER OF CRITICAL POINTS
SELECTED EXCEEDS NUMBER
ACTUALLY LOCATED - CALCULATION
ENDED

This message will be printed if more critical points are specified in straining ($LSPEC = 0$) than the number located by the program. The remedy is to specify a number of points less than or equal to the actual number.

ORDER OF SPECIFIED POINTS IN
BASE AND CALIBRATION SOLUTIONS
DOES NOT CORRESPOND - CALCULATION ENDED

This message will be printed if the fixed points specified ($LSPEC = 0$) occur in a different sequence in the base and calibration solutions. The remedy is to use base and calibration solutions having the same qualitative features.

A.8 SAMPLE CASE

The sample case presented in this section provides some example results of perturbation calculations and comparisons with 'exact' nonlinear solutions for a multiple-shock flow for which partial results were provided in figure 3 of the main text. The calculation is for the simultaneous four-parameter ($M_\infty, \tau, t, \alpha$) perturbation of strongly-supercritical full potential flows past a cascade of blades having NACA four-digit profiles. The base flow is for oncoming Mach number $M_\infty = 0.780$, thickness ratio $\tau = 0.110$, gap-to-chord ratio $t = 3.2$, and oncoming inflow angle $\alpha = 0.3^\circ$. The calibration flows to account for perturbations in these parameters are at the following values of these

parameters) $(M_\infty, \tau, t, \alpha) = (0.790, 0.110, 3.2, 0.3), (0.780, 0.120, 3.2, 0.3), (0.780, 0.110, 3.0, 0.3), (0.780, 0.110, 2.3, 0.5)$. Perturbation results have been determined for 19 different solutions, which are summarized in the summary table after the print output. Results for several of those cases are presented here.

The input data is tabulated in figure A.1, with item numbers corresponding to those identified in Section A.5.1 and A.5.2. The first card, item 1, provides the title of the run. The next card, item 2, indicates that the input deck will not be printed (LECHO = 0). The next card, item 3, indicates that there are 191 points ($N = 191$) at which data will be input for the base, calibration, and comparison solutions; that there will be 19 cases (NCASE = 19) for which perturbation solutions are to be computed; that the number of parameters to be perturbed are 4 (NPARAM = 4); that there will be three invariant points (NSELCT = 3) in addition to the end points; that the invariant points will be located by the program (LSPEC = 0); that the information regarding the unit perturbation will be printed (LUNIT = 1); that there will be a comparison of the perturbation results with the exact solution (LCHEK = 1); and that there will be plots by a peripheral device of the output (LPLOT = 1). The next card, item 4, specifies that the three invariant points to be selected by the program are to be: the medium point (LSELECT(1) = 2), i.e., the stagnation point; the first critical point (LSELECT(2) = 3), i.e., the 1st shock point found when moving forward on the bottom surface from the leading edge; and the second critical point (LSELECT(3) = 4), i.e., the end shock point.

The next card, item 5, indicates that the dependent variable for print output will be symbolized by a 'CP' denoting the pressure coefficient. The next card, item 6, indicates that the parameters to be varied are "PARNAM(1) = MACH NO., PARNAME(L) = TAU, PARNAME(3) = PITCH, PARNAME(4) = ALPHAL. The next card, item 7, indicates that the coordinates of the data points to be read in will start at $x = 1.0$ on the lower surface and will end at $x = 1.0$ on the upper surface, i.e., $A = -1$, $B = 1$. The next card, item 8, provides the oncoming Mach number of the base flow $M_0 = 0.780$. The next card, item 9, provides the base flow values of the parameters to be perturbed: $Q_0(1) = 0.780$, $Q_0(2) = 0.110$, $Q_0(3) = 3.2$, $Q_0(4) = 0.30$. The following 25 cards, item 10, provide the 191 base flow values of the surface coordinates XBASE(I), $I = 1, N$, while the next 5 cards, item 10, provide the 191 base flow values of the surface coordinates XBASE(I), $I = 1, N$, while the next 25 cards, item 11 provide the 191 base flow values of the dependent variable (pressure coefficient) YBASE(I), $I = 1, N$. The next card, item 13 provides the values of the oncoming Mach number M_1 and the value of the 1st-perturbation parameter in the first calibration solution. Items 14 and 15, which correspond to the arrays

of 191 points of coordinates XCALB(I), I = 1, N and dependent variable YCALB(I), I = 1, N, are provided as for the base flow in two sets of 25 cards each. Items 13, 14, and 15 are then repeated 3 more times corresponding to the total of NPARAM = 4 calibration solutions required. Items 17, 18, 19 and 20, of which there are 19 sets corresponding to the 19 cases to be studied, provide analogous information as items 8, 9, 10, and 11 of the base flow, but now refer to the 'exact' nonlinear results. Items 19 and 20, which correspond to the XCHEK(I), I = 1, N and YCHEK(I), I = 1, N arrays, respectively, have of course been previously computed at the indicated values of the perturbed parameters, and are included here for comparative purposes to enable assessment of the perturbation results.

Figure A.2 provides an abbreviated print output for sample case, while figure A.3 provides an abbreviated plot output of the results for the 19 cases, and display the base (----), calibration (.....), perturbation (****), and 'exact' nonlinear (—) flow solutions. In these plots, the calibration solutions denoted "calibration no. 1 of 4", etc. correspond to those indicated in the summary table provided at the end of the print output for this case shown in Figure A.2.

TEST CASE - 4 PARAMETER PERTURBATION OF A SUPERCRITICAL CASCADE FLOW

CP	MACH NO.	TAU	PITCH	ALPHA1	0	191	19	4	3	0	1	1	1	
-1.0	-1.0	1.0			0	191	19	4	3	0	1	1	1	
.780000	.110000	3.200000	.300000	.995569	*.993212	*.990334	*.986638	*.983028	*.98004	*.949004	*.941022	*.933768	*.933768	
.999495	.998713	*.997404	*.995569	.962325	*.955988	*.949004	*.941022	*.94004	*.887764	*.877764	*.86630	*.855106	*.855106	
.978608	.973681	*.968251	*.967766	*.897818	*.887764	*.877764	*.86630	*.855106	*.793319	*.779955	*.766302	*.752364	*.752364	
.925451	.916680	*.877764	*.877764	*.806349	*.793319	*.779955	*.766302	*.752364	*.730000	*.716638	*.693319	*.679955	*.664895	
.843453	.831634	*.819062	*.819062	*.694061	*.678926	*.663598	*.648995	*.632431	*.568512	*.552310	*.536030	*.519718	*.503382	
.738155	.723692	*.708989	*.708989	*.604859	*.581952	*.561952	*.540538	*.520382	*.438147	*.421952	*.405838	*.389822	*.373922	
.616624	.600691	*.584668	*.584668	*.504859	*.481952	*.464625	*.446625	*.427928	*.327986	*.311819	*.281906	*.267292	*.252928	
.487040	.470708	*.454404	*.454404	*.396714	*.396714	*.379955	*.364625	*.342536	*.211494	*.198286	*.185405	*.172863	*.160616	
.358154	.342536	*.327986	*.327986	*.296714	*.296714	*.281906	*.267292	*.252928	*.225014	*.211494	*.198286	*.185405	*.172863	
.238830	.225014	*.211494	*.211494	*.198286	*.198286	*.185405	*.172863	*.160616	*.126369	*.115725	*.105497	*.095694	*.086328	
.137416	.126369	*.115725	*.115725	*.095694	*.095694	*.086328	*.077006	*.068939	*.053397	*.046336	*.039757	*.033665	*.022265	
.060934	.053397	*.046336	*.046336	*.039757	*.039757	*.028066	*.018360	*.018360	*.010617	*.007481	*.004859	*.002755	*.000000	
.014246	.010617	*.007481	*.007481	*.004859	*.004859	*.001255	*.000321	*.000000	*.001225	*.002752	*.004802	*.007411	*.010536	
.000306	.001225	*.002752	*.002752	*.004802	*.004802	*.001255	*.000321	*.000000	*.027959	*.033557	*.039651	*.053303	*.060550	
.022860	*.027959	*.033557	*.033557	*.039651	*.039651	*.036598	*.032733	*.029144	*.088699	*.086682	*.105510	*.115766	*.126440	
.077735	*.088699	*.086682	*.086682	*.105510	*.115766	*.126440	*.137520	*.148995	*.160851	*.185658	*.198581	*.211831	*.225393	
.160851	*.173077	*.185658	*.185658	*.198581	*.211831	*.225393	*.239552	*.253392	*.287342	*.312446	*.327750	*.343235	*.358886	
.267798	*.282454	*.297342	*.297342	*.312446	*.327750	*.343235	*.358886	*.374684	*.390612	*.42287	*.438999	*.455221	*.487223	
.520603	*.536909	*.553170	*.553170	*.569369	*.585499	*.601512	*.617422	*.633202	*.644307	*.679660	*.694698	*.709586	*.724248	
.648836	*.664307	*.679660	*.679660	*.694698	*.709586	*.724248	*.738669	*.752835	*.766729	*.780338	*.793648	*.806645	*.819315	
.866474	*.877323	*.887774	*.887774	*.897815	*.907436	*.916645	*.934623	*.955237	*.897815	*.907436	*.916645	*.925376	*.933675	
.941515	*.968886	*.955781	*.955781	*.962194	*.968118	*.973549	*.97880	*.982907	*.968628	*.993124	*.995497	*.997348	*.998675	
.660763	*.575142	*.508550	*.508550	*.454292	*.407733	*.365898	*.327733	*.291440	*.254241	*.195204	*.16076	*.141713	*.120201	
.079017	*.069975	*.061687	*.061687	*.052188	*.041329	*.029199	*.016319	*.002961	*.024487	*.038230	*.051655	*.064464	*.076552	
*.010696	*.024487	*.038230	*.038230	*.051655	*.064464	*.076552	*.088017	*.098956	*.109403	*.119377	*.128919	*.138069	*.154608	
*.169807	*.170420	*.167580	*.167580	*.160664	*.146727	*.146727	*.161358	*.166584	*.160323	*.198376	*.202844	*.208692	*.279184	
*.000429	*.118968	*.972139	*.972139	*.974261	*.974261	*.974261	*.973130	*.970687	*.960708	*.962275	*.957354	*.950981	*.943667	
*.962228	*.962275	*.957354	*.957354	*.950981	*.943667	*.935439	*.926325	*.916319	*.905393	*.893513	*.880631	*.866686	*.851623	
*.778826	*.756758	*.733216	*.733216	*.707749	*.680122	*.650175	*.617610	*.581957	*.709565	*.726777	*.742591	*.757044	*.770186	
*.542170	*.598376	*.452347	*.452347	*.02844	*.02844	*.02844	*.026567	*.104356	*.119375	*.981404	*.799047	*.601992	*.426749	
*.802242	*.810509	*.817508	*.817508	*.827167	*.827167	*.827167	*.829662	*.827168	*.821045	*.793036	*.563136	*.314794	*.291525	
*.317122	*.308175	*.297215	*.297215	*.251825	*.303986	*.350902	*.394016	*.434147	*.471923	*.542111	*.572529	*.600042	*.625381	
*.218218	*.204700	*.191285	*.191285	*.178085	*.165156	*.152520	*.139950	*.127253	*.114431	*.101360	*.087898	*.073914	*.059507	
*.082242	*.081046	*.022726	*.022726	*.033541	*.042657	*.050867	*.059558	*.069860	*.299688	*.099296	*.120145	*.146719	*.200771	
*.299688	*.331703	*.368645	*.368645	*.408078	*.450350	*.494136	*.5277147							

Figure A.1- Card input data for sample case

* 790000	* 998713	* 997404	* 995569	* 993212	* 986398
* 998495	* 986251	* 986251	* 962230	* 955008	* 949004
* 978608	* 968181	* 968181	* 897818	* 887747	* 941022
* 925451	* 916680	* 907466	* 806149	* 79309	* 933168
* 843453	* 831434	* 815062	* 749955	* 766302	* 855106
* 738155	* 723692	* 708989	* 694061	* 678926	* 752664
* 616624	* 600691	* 586448	* 566512	* 552300	* 648958
* 481040	* 470708	* 430140	* 40147	* 421525	* 519118
* 359154	* 342536	* 321086	* 311819	* 296554	* 503382
* 238830	* 225014	* 211494	* 198286	* 185405	* 503382
* 137416	* 126369	* 115725	* 105497	* 095694	* 503382
* 160851	* 173077	* 185568	* 198581	* 211831	* 523952
* 267798	* 282454	* 291342	* 312446	* 327550	* 536330
* 014246	* 001617	* 007481	* 004859	* 002275	* 536330
* 000306	* 001225	* 002732	* 004002	* 007411	* 519118
* 022860	* 027159	* 03557	* 046351	* 010536	* 504288
* 077353	* 086293	* 096682	* 105510	* 115766	* 504288
* 766729	* 789338	* 793648	* 806648	* 819350	* 504288
* 866474	* 877323	* 887774	* 891815	* 901436	* 504288
* 915115	* 948886	* 955781	* 962294	* 968118	* 504288
* 986826	* 990233	* 993124	* 994997	* 998875	* 504288
* 625161	* 636677	* 547948	* 496922	* 441873	* 504288
* 285957	* 253438	* 222993	* 194725	* 169258	* 504288
* 106952	* 094280	* 091443	* 081443	* 081645	* 504288
* 024649	* 012825	* 001354	* -009465	* 019256	* 504288
* 046383	* 049859	* -051728	* -051769	* -049558	* 044424
* 000740	* 033790	* -272131	* -791329	* -989308	* -996302
* 995378	* 993891	* -991799	* -989117	* -989865	* -996271
* 966842	* -960329	* -953063	* -940330	* -938219	* -926638
* 893227	* -880406	* -866638	* -851835	* -835915	* -818807
* 759623	* -731668	* -713171	* -681274	* -627841	* -035448
* 519680	* -758561	* -229425	* -379572	* -322975	* -035805
* 022051	* 139662	* 283583	* 465236	* -678740	* -035805
1. * 123192	* 991649	* 804492	* 608914	* -898547	* -035805
* 028452	* -110695	* -180112	* -239458	* -291613	* -916295
* 461504	* -499119	* -533047	* -563342	* -591381	* -04164
* 666010	* -716736	* -724100	* -741189	* -757073	* -04164
* 810203	* -821169	* -831244	* -840398	* -848603	* -04164
* 871795	* -815184	* -877640	* -879096	* -879394	* -04164
* 129401	* -177701	* -151619	* -170257	* -180343	* -04164
* 176313	* -169347	* -161181	* -152229	* -142803	* -04164
* -101001	* -089373	* -071144	* -064229	* -050755	* -037019
* 003304	* 015860	* 027572	* 037833	* 046295	* -053816
* 084631	* 100396	* 120360	* 144252	* 170602	* -062008
* 290546	* 323158	* 357105	* 391082	* 422405	* -028312
* 780000	* 120000				* -258942
* 99496	* 998714	* 997407	* 995575	* 993220	* 986395
* 978636	* 973715	* 968293	* 962374	* 955963	* 941022
* 925533	* 916769	* 907562	* 897920	* 887856	* 941022
* 843587	* 831573	* 819206	* 806497	* 793460	* 941022
* 738311	* 723866	* 709141	* 694210	* 679070	* 941022
* 616735	* 600791	* 584735	* 566584	* 553357	* 941022
* 013421	* 009888	* 006887	* 004222	* 002497	* 001117
* 000268	* 001087	* 002453	* 004364	* 006817	* 000000
* 021952	* 027035	* 032624	* 038714	* 053000	* 013330
* 076471	* 085433	* 094844	* 104695	* 114978	* 018150
* 160176	* 172433	* 1H5445	* 198000	* 2112R2	* 068038

Figure A.1- Continued

*267379	*282067	*296987	*312123	*327458	*342974	*358655	*374482
*390439	*405056	*422668	*438906	*455202	*471539	*487400	*504266
*520620	*536945	*553223	*569438	*585571	*601607	*617529	*633319
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*941585	*948449	*955837	*962243	*968160	*973584	*978508	*982330
*988844	*990246	*993133	*995503	*997351	*998676	*999477	
*778187	*643254	*577572	*514959	*4644661	*418858	*377585	*339440
*303090	*269002	*231151	*207670	*181000	*158080	*139916	*126059
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*004508	*054337	*056667	*059591	*053831	*048579	*039553	*024489
*002015	*038189	*021204	*0199151	*01053651	*0163071	*01063441	*1.063136
*1.062176	-1.066557	-1.05341	-1.055341	-1.051783	-1.042793	-1.037260	
*1.030937	-1.023198	-1.015828	-1.0107010	-0.997326	-0.986775	-0.975357	*0.963040
*0.949770	-0.935492	-0.920148	-0.903678	-0.886027	-0.867129	-0.848875	-0.825114
*-0.801701	-0.776498	-0.749336	-0.700000	-0.688338	-0.65262	-0.624464	
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*0.071113	*0.213735	*0.379255	*0.564986	*0.760419	*0.944831	*1.086243	*1.151976
1.125456	1.026020	0.869853	0.693094	0.518632	0.358587	0.099338	
*0.001756	*0.088718	-0.165230	-0.233282	-0.296060	-0.348640	-0.397256	-0.441646
*-0.483708	-0.525064	-0.553578	-0.597755	-0.628929	-0.657892	-0.684937	-0.710125
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*-0.416111	-0.108785	-0.148390	-0.169889	-0.181857	-0.187368	-0.188309	-0.185828
*-0.180806	-0.096601	-0.156628	-0.156444	-0.146644	-0.136477	-0.12584	-0.10585
*-0.103868	-0.056601	-0.077684	-0.064024	-0.049764	-0.035231	-0.020114	-0.006418
*-0.007479	-0.020791	-0.033204	-0.041119	-0.053225	-0.061398	-0.070178	-0.080759
*-0.094133	-0.111067	-0.132206	-0.157348	-0.151712	-0.147778	-0.145656	-0.277888
*-0.311239	*3.55517	*3.381103	*3.417216	*3.454758	*4.77775	*5.04653	
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*925230	*916440	*907206	*897540	*887452	*876952	*866653	*854765
*843100	*823071	*818690	*805971	*792926	*779570	*765916	*751980
*731776	*723320	*708626	*693710	*678589	*663277	*647790	*632146
*616361	*600450	*584431	*568320	*552134	*535890	*5119605	*503497
*486981	*40677	*454399	*438167	*421997	*405907	*389913	*370333
*356285	*326885	*327250	*311997	*296944	*282106	*267500	*253142
*239049	*225235	*221716	*198506	*185622	*173076	*160884	*143057
*131609	*126552	*115899	*105660	*095846	*086668	*077535	*063056
*0.061039	*0.054491	*0.046419	*0.039828	*0.027860	*0.01287	*0.023006	*0.018393
*0.01272	*0.010637	*0.007494	*0.004866	*0.002780	*0.000322	*0.000000	
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*0.07511	*0.064667	*0.095872	*0.105716	*0.115907	*0.126676	*0.137771	*0.142260
*0.161129	*0.173367	*0.185958	*0.198890	*0.212147	*0.225715	*0.239578	*0.253720
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*0.686683	*0.661132	*0.679404	*0.694484	*0.709355	*0.724002	*0.738410	*0.755656
*0.766451	*0.780054	*0.793360	*0.806354	*0.819024	*0.831357	*0.843339	*0.854959
*0.866204	*0.877063	*0.887524	*0.897578	*0.907212	*0.916417	*0.925182	*0.933497
*0.941352	*0.98740	*0.955651	*0.962079	*0.968019	*0.973664	*0.978049	*0.982849
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*0.672452	*0.583881	*0.515448	*0.460229	*0.412948	*0.370572	*0.313613	*0.295165
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*-0.104490	*-0.113567	*-0.122034	*-0.129895	*-0.136994	*-0.142975	*-0.147404	*-0.149788
*-0.149528	*-0.158860	*-0.137715	*-0.123451	*-0.10101	*-0.085954	*-0.043774	*-0.077544
*-0.979011	*-0.985637	*-0.987080	*-0.987308	*-0.986512	*-0.984670	*-0.982374	*-0.970017
*-0.974765	*-0.989621	*-0.963601	*-0.956710	*-0.948951	*-0.940342	*-0.930900	*-0.926113
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Figure A.1- Continued

-• 780518	-• 758491	-• 734847	-• 709222	-• 681334	-• 651131	-• 618334	-• 582483
-• 542556	-• 498610	-• 452362	-• 402696	-• 346011	-• 278795	-• 197933	-• 103707
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-• 044456	-• 127566	-• 197281	-• 256824	-• 607126	-• 632579	-• 656470	-• 678856
-• 447793	-• 514751	-• 548982	-• 579556	-• 767449	-• 811197	-• 793805	-• 805309
-• 699603	-• 718693	-• 736277	-• 752499	-• 846289	-• 850815	-• 853885	-• 855303
-• 815729	-• 825080	-• 833338	-• 840434	-• 846762	-• 850815	-• 853885	-• 855303
-• 854769	-• 851647	-• 840175	-• 740515	-• 426762	-• 303010	-• 261195	-• 284058
-• 290361	-• 289859	-• 289496	-• 277072	-• 276759	-• 256245	-• 244501	-• 232272
-• 219107	-• 206939	-• 194098	-• 181343	-• 168799	-• 156420	-• 14046	-• 131552
-• 118857	-• 105875	-• 092470	-• 078517	-• 064113	-• 049551	-• 05087	-• 026898
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-• 077547	-• 094072	-• 114948	-• 139593	-• 166841	-• 195931	-• 226507	-• 258379
-• 291544	-• 326079	-• 362254	-• 400337	-• 440466	-• 479680	-• 499458	-• 529458
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-• 708478	-• 726964	-• 743961	-• 759534	-• 773825	-• 788882	-• 798764	-• 808499
-• 819692	-• 827532	-• 834774	-• 840720	-• 845258	-• 848250	-• 859481	-• 868582
-• 844747	-• 832277	-• 740646	-• 437898	-• 244913	-• 282473	-• 299178	-• 304774
-• 303452	-• 297680	-• 289019	-• 278493	-• 266799	-• 254426	-• 21684	-• 228117
-• 215607	-• 202439	-• 189306	-• 176332	-• 163616	-• 151102	-• 138627	-• 126062
-• 113334	-• 100323	-• 086919	-• 072983	-• 058612	-• 044096	-• 029682	-• 015537
-• 001814	-• 011308	-• 023549	-• 034378	-• 043518	-• 01758	-• 00823	-• 026239
-• 083134	-• 100208	-• 121124	-• 145915	-• 173205	-• 202435	-• 233136	-• 265239
-• 298921	-• 334230	-• 371701	-• 411706	-• 454665	-• 499081	-• 529480	-• 560000
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Figure A.1- Continued

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*843239	*831216	*8184*0	*806125	*793084	*786179
*737939	*723481	*708785	*693865	*678738	*6633420
*616477	*600554	*589522	*568396	*552194	*535932
*486962	*470534	*454333	*438075	*421879	*405761
*358051	*342220	*322954	*311669	*296583	*281712
*238556	*224109	*211157	*197916	*185000	*172423
*136865	*125181	*115101	*104836	*094999	*086646
*060117	*052557	*045478	*038887	*032789	*022095
*013445	*009306	*008899	*004430	*002502	*001118
*000269	*001089	*002658	*004374	*006832	*009828
*021999	*027993	*032693	*038795	*045394	*052484
*076624	*085600	*095027	*104894	*115191	*125906
*160445	*172713	*185336	*198299	*212589	*225189
*261688	*282383	*297129	*312429	*327756	*343262
*390682	*406311	*422873	*435089	*455363	*471675
*520678	*539976	*553227	*569415	*585522	*601532
*648814	*664271	*679550	*694635	*709510	*724160
*766611	*780213	*795516	*805057	*819173	*831501
*866331	*877183	*887638	*897684	*907312	*916510
*941425	*948805	*955109	*962130	*968062	*978438
*986798	*990212	*993109	*995486	*997342	*998672
*715525	*623070	*555645	*497218	*446870	*402235
*287908	*287908	*221641	*197479	*144936	*123510
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Figure A.1- Continued

•267915	•282584	•297744	•312597	•327907	•343397	•359050	•374497
•390772	•406807	•420235	•439138	•455379	•455379	•468023	•504352
•520669	•536957	•553200	•569379	•585479	•601482	•617372	•653133
•648748	•664201	•679476	•694558	•709431	•724980	•738490	•752644
•766530	•780132	•793337	•806430	•819098	•831128	•843408	•855024
•866266	•877122	•881580	•897630	•907262	•916463	•925225	•935356
•941388	•948772	•955679	•962104	•968040	•973481	•978423	•982861
•986789	•993025	•993104	•995484	•997340	•998671	•999475	
•-1.016993	•-1.018565	•-1.012929	•-1.012432	•-1.012423	•-1.012423	•-1.012423	•-1.012432
•-0.994302	•-0.987764	•-0.980388	•-0.921666	•-0.963089	•-0.953160	•-0.942382	•-0.930729
•-0.918150	•-0.904595	•-0.890005	•-0.870026	•-0.847326	•-0.839474	•-0.820137	•-0.799347
•-0.776990	•-0.752989	•-0.735226	•-0.694469	•-0.669458	•-0.636926	•-0.601581	•-0.562768
•-0.519590	•-0.472765	•-0.422853	•-0.368337	•-0.306643	•-0.235628	•-0.182827	•-0.156288
•0.05611	•0.186640	•0.343244	•0.528224	•0.732213	•0.930643	•1.084243	•1.152652
•1.191711	•1.996502	•8.19294	•6.924927	•4.51350	•2.97174	•0.601094	•0.513131
•-0.046882	•-0.129432	•-0.20286	•-0.266035	•-0.322394	•-0.372901	•-0.418740	•-0.461181
•-0.502119	•-0.541223	•-0.576613	•-0.608757	•-0.638152	•-0.665193	•-0.693397	•-0.713987
•-1.35954	•-0.756319	•-0.75226	•-0.792826	•-0.809216	•-0.824468	•-0.838636	•-0.851758
•-0.863864	•-0.874989	•-0.891513	•-0.943110	•-0.904691	•-0.906934	•-0.915786	•-0.920910
•-0.924997	•-0.928077	•-0.930140	•-0.931090	•-0.930643	•-0.928032	•-0.926110	•-0.946648
•-0.167509	•-0.159802	•-0.184662	•-0.198446	•-0.204887	•-0.206189	•-0.203851	•-0.198866
•-0.191925	•-0.183550	•-0.174160	•-0.164133	•-0.153755	•-0.143096	•-0.132081	•-0.120668
•-0.108824	•-0.096494	•-0.083570	•-0.069961	•-0.05793	•-0.041369	•-0.026019	•-0.012789
•-0.00983	•-0.014170	•-0.026677	•-0.037322	•-0.046365	•-0.054416	•-0.063019	•-0.073419
•-0.086493	•-0.103247	•-0.124339	•-0.149521	•-0.177204	•-0.206747	•-0.237781	•-0.270073
•-0.303583	•-0.338338	•-0.374562	•-0.412278	•-0.451357	•-0.486687	•-0.496474	
•780000							
3.100000	•500000	•995568	•993210	•990331	•986935	•983025	
•999495	•998712	•997403	•995568	•993210	•990331	•986935	
•978604	•973676	•968247	•962320	•955901	•948997	•933757	
•925438	•916666	•901450	•897800	•887728	•87725	•855086	
•843434	•831416	•819045	•806333	•793294	•779942	•766290	
•738146	•723684	•708883	•694057	•678922	•663595	•648091	
•616619	•600684	•586638	•568499	•552282	•536006	•519688	
•486992	•470649	•453333	•438006	•421851	•405720	•373766	
•357976	•342336	•328860	•311568	•296476	•281599	•266956	
•238433	•224584	•211032	•197792	•184877	•172302	•160082	
•136756	•125677	•112002	•104744	•0.94913	•0.855220	•0.76573	
•0.060057	•0.052504	•0.045331	•0.038846	•0.032754	•0.027161	•0.022071	•0.017492
•0.013431	•0.009895	•0.00892	•0.004425	•0.002499	•0.001117	•0.000283	•0.000000
•0.002668	•0.01088	•0.01255	•0.01368	•0.006823	•0.009816	•0.013342	•0.017396
•0.021971	•0.020759	•0.022552	•0.028748	•0.045339	•0.052420	•0.059986	•0.068027
•0.076535	•0.085503	•0.094221	•0.104779	•0.115067	•0.125775	•0.136889	•0.148229
•1.60291	•1.075553	•1.085170	•1.098129	•1.11414	•1.225011	•1.289095	•1.253079
•267518	•282205	•297124	•312258	•327589	•343101	•358777	•374598
•390547	•4.066607	•4.022760	•4.38989	•4.552275	•4.71602	•4.67952	•5.043036
•520649	•5.039692	•5.532229	•5.669432	•5.855554	•6.01578	•6.17489	•6.33269
•648901	•6.64370	•6.796861	•6.94756	•7.095644	•7.24299	•7.38716	•7.52877
•766767	•780372	•793678	•806670	•819336	•831662	•843637	•855247
•866482	•877329	•887778	•897817	•907437	•916627	•925377	•933676
•941516	•948887	•955872	•962194	•968118	•973547	•978478	•982905
•986824	•990231	•993123	•995496	•997347	•998674	•999476	
•-0.004402	•-0.02819	•-0.05487	•-0.054816	•-0.052471	•-0.047068	•-0.037620	•-0.022892
•-1.054885	•-1.052919	•-1.050513	•-1.04725	•-1.042134	•-1.036912	•-1.031609	•-1.027791
•-1.021187	•-1.013767	•-1.005518	•-0.986421	•-0.986453	•-0.975611	•-0.963891	•-0.951264

Figure A.1- Continued

-.937684	-.923306	-.907481	-.890152	-.872857	-.853107	-.833170	-.811084
-.787336	-.761495	-.734481	-.709114	-.672806	-.638053	-.600780	-.560151
-.514699	-.4664950	-.411349	-.352299	-.286280	-.211972	-.127151	-.026132
.091396	.233180	.398011	.582956	.775553	.956501	1.093184	1.153325
1.125185	1.01614	.856124	.676177	.499076	.339082	.198623	.079197
-.021793	-.108556	-.184638	-.252662	-.313226	-.367414	-.416085	-.460625
-.503433	-.544511	-.582139	-.616233	-.647452	-.676123	-.702802	-.727756
-.751024	-.726552	-.792785	-.811574	-.829121	-.846497	-.860760	-.874953
-.688118	-.900597	-.921512	-.921155	-.930328	-.939328	-.946686	-.953101
-.958592	-.963277	-.967063	-.970094	-.972266	-.974867	-.973624	-.974336
-.953783	-.544528	-.120123	-.106248	-.1130620	-.145161	-.153053	-.156106
-.155544	-.152266	-.146967	-.140231	-.132502	-.123987	-.114720	-.104720
-.094018	-.082917	-.070443	-.057413	-.043684	-.026611	-.015490	-.001511
.012137	.025217	.037417	.048192	.057280	.063462	.074030	.084126
.097121	.114220	.135718	.161004	.188854	.218509	.249622	.281961
.315461	.350089	.386020	.423017	.460380	.491526	.485025	
.780000	.780000	.120000	.000000	.400000	.993196	.990312	.986909
.994495	.998710	.997398	.995559	.995822	.989072	.941512	.933645
.978562	.977626	.968187	.962250	.987565	.877072	.866179	.855898
.925316	.916533	.907306	.897646	.8806125	.793084	.779130	.766078
.843239	.831216	.810840	.8006125	.693865	.678738	.663420	.632273
.737939	.723981	.708785	.693865	.568396	.552194	.535932	.501299
.616477	.600554	.584522	.568396	.438075	.421879	.405761	.389739
.486962	.470534	.454333	.4311669	.326954	.296583	.281712	.252883
.358051	.342220	.326954	.311669	.224719	.211157	.197916	.172423
.238556	.224719	.211157	.200000	.136865	.115101	.104336	.094999
.136865	.125181	.115101	.104336	.038887	.032789	.027190	.022095
.060117	.052557	.045478	.045478	.006899	.00430	.002502	.001118
.013445	.009906	.006899	.006899	.002458	.004314	.006832	.009828
.000269	.001089	.001089	.001089	.027093	.032693	.038795	.045394
.021999	.021999	.021999	.021999	.085500	.095027	.104694	.115191
.0766624	.0766624	.0766624	.0766624	.1.023471	.1.05147	.1.062837	.1.059139
.1.02470	.1.13427	.1.05147	.1.062837	.1.059139	.1.056227	.1.059096	.1.059096
.038477	.0266663	.015354	.004891	.004891	.004359	.012203	.018621
-.026928	-.026928	-.026928	-.026928	-.026928	-.026928	-.0110532	-.023669
-.0622792	-.426055	-.008072	-.025613	-.025613	-.020502	-.0110532	-.024220
-.1.068721	-.1.066665	-.1.066665	-.1.066665	-.1.023471	-.1.014973	-.1.005662	-.0995523
-.1.031174	-.1.031174	-.1.031174	-.1.031174	-.1.023471	-.1.014973	-.1.005662	-.0995523
-.965626	-.931959	-.877183	-.886638	-.897684	-.907312	-.916510	-.925268
-.795852	-.770387	-.742952	-.713336	-.681410	-.670362	-.647145	-.610153
-.544361	-.474318	-.420449	-.361446	-.2907699	-.285520	-.242175	-.213326
-.081839	.223787	.389020	.574157	.768549	.951307	1.090309	1.152922
1.122314	1.020678	.861946	.683339	.507472	.360620	.346936	.3072758
-.013881	-.1.00803	-.1.00803	-.1.00803	-.1.023471	-.1.014973	-.1.005662	-.0995523
-.466591	-.538075	-.575946	-.610205	-.641741	-.670362	-.6461641	-.614641
-.766416	-.768272	-.788649	-.807699	-.842175	-.877721	-.857721	-.827205
-.855676	-.898183	-.909755	-.920387	-.938854	-.946721	-.953690	-.953690
.011614	.024411	.036365	.046863	.055580	.063350	.071654	.081662
.094415	.111091	.132345	.157364	.184883	.214492	.245418	.277320
.3444221	.360022	.413320	.445721	.482948	.503599		

Figure A.1- Continued

Figure A.1- Continued

*.076994	*.095942	*.095348	*.095942	*.105195	*.115470	*.126164	*.137264	*.148759
*.160437	*.172883	*.185484	*.189426	*.211695	*.225276	*.239153	*.253311	*.258959
*.267733	*.282403	*.297306	*.312423	*.327739	*.343235	*.358895	*.374701	*.388954
*.390636	*.406682	*.422822	*.439037	*.455311	*.471626	*.487964	*.504308	*.520640
*.520640	*.536944	*.553202	*.569396	*.585511	*.601529	*.617334	*.633208	*.633208
*.648836	*.664301	*.679588	*.694681	*.709563	*.724220	*.738636	*.752797	*.752797
*.766689	*.780293	*.793600	*.806594	*.819262	*.831591	*.843568	*.855181	*.855181
*.866419	*.877269	*.887721	*.897764	*.907388	*.916581	*.925334	*.933637	*.933637
*.941480	*.948854	*.955753	*.962169	*.968096	*.973529	*.978663	*.982893	*.982893
*.948815	*.990224	*.993118	*.995493	*.997345	*.998674	*.99976	*.99976	*.99976
*.708054	*.626684	*.655502	*.677759	*.697959	*.704851	*.704851	*.704851	*.704851
*.291788	*.258595	*.227420	*.198410	*.172222	*.150032	*.132220	*.110942	*.110942
*.108442	*.099799	*.091488	*.082425	*.072058	*.060655	*.048632	*.036274	*.036274
*.023885	*.011678	*.000174	*.01359	*.021480	*.030350	*.038005	*.044433	*.044433
*.049514	*.053106	*.050339	*.052094	*.052834	*.047569	*.03804	*.024088	*.024088
*.002259	*.035163	*.0345250	*.0345250	*.0345250	*.023550	*.023440	*.022733	*.022733
*.1.021332	-1.019524	-1.016994	-1.013857	-1.010133	-1.005832	-1.000927	-1.000927	-1.000927
-*.989907	-*.981894	-*.973992	-*.965286	-*.955761	-*.945417	-*.934255	-*.922244	-*.922244
-*.909331	-*.895461	-*.880573	-*.864613	-*.847523	-*.829247	-*.809660	-*.788684	-*.788684
-*.766142	-*.741969	-*.716034	-*.688119	-*.658002	-*.625513	-*.590209	-*.551404	-*.551404
-*.508252	-*.461289	-*.411314	-*.356826	-*.295257	-*.224430	-*.141900	-*.045738	-*.045738
*.065662	*.196054	*.351813	*.535659	*.738207	*.935032	*.1.087772	*.1.154647	*.1.154647
1.121995	*.999766	*.822454	*.635189	*.458138	*.304762	*.174308	*.061846	*.061846
*.035783	*.120048	*.192797	*.256299	*.312059	*.363011	*.408866	*.451467	*.451467
*.492619	*.531171	*.567246	*.5949429	*.628809	*.655920	*.681323	*.705146	*.705146
*.727339	*.747981	*.767204	*.785161	*.801916	*.817586	*.832265	*.845717	*.845717
*.858228	*.869924	*.880649	*.890454	*.898833	*.907299	*.914359	*.920516	*.920516
*.925789	*.930241	*.933029	*.936844	*.938932	*.940104	*.940104	*.938894	*.938894
*.934220	*.568149	*.134333	*.090919	*.115918	*.132691	*.142776	*.147634	*.147634
*.148666	*.146268	*.148822	*.135787	*.128658	*.120671	*.111686	*.102351	*.102351
*.092102	*.081139	*.0669397	*.056823	*.049941	*.0429941	*.0316244	*.0202707	*.0202707
*.010486	*.023167	*.035059	*.054545	*.054687	*.061782	*.070159	*.080451	*.080451
*.093374	*.109714	*.130386	*.154983	*.182132	*.211150	*.241598	*.273262	*.273262
*.306912	*.339921	*.374955	*.411358	*.448035	*.480512	*.465608		
*.785000	*.115000	3.000000	*.400000	*.990305	*.993191	*.996900	*.982979	*.982979
*.999495	*.998709	*.997396	*.995556	*.992225	*.995793	*.998874	*.99176	*.933606
*.978647	*.973608	*.968165	*.962257	*.987507	*.877011	*.866115	*.854830	
*.925272	*.916486	*.907255	*.897592	*.887507	*.779304	*.765996	*.752061	
*.843169	*.831142	*.818764	*.806047	*.793663	*.663347	*.647658	*.632209	
*.737857	*.723399	*.708704	*.693787	*.678663	*.535911	*.519617	*.503298	
*.616618	*.600502	*.584476	*.568357	*.542164	*.405835	*.379393		
*.486972	*.470656	*.454367	*.438122	*.421939	*.281912	*.26720	*.252916	
*.358170	*.342554	*.327104	*.311836	*.296766	*.172755	*.160547	*.148705	
*.238806	*.224976	*.211441	*.198216	*.185316	*.126298	*.137406	*.148909	
*.137243	*.126173	*.1155016	*.105255	*.095429	*.086040	*.077097	*.068609	
*.160193	*.173045	*.185652	*.198599	*.211812	*.225456	*.239335	*.253493	
*.060584	*.053130	*.045954	*.039362	*.033261	*.027656	*.022552	*.017951	
*.0138855	*.010265	*.0071190	*.004642	*.002636	*.001185	*.000302	*.000000	
*.000287	*.001156	*.002593	*.004586	*.007122	*.010187	*.013769	*.017859	
*.022455	*.027558	*.033165	*.039270	*.042869	*.052955	*.060523	*.068565	
*.077074	*.086041	*.095556	*.105311	*.115596	*.126298	*.137406	*.148909	
*.766530	*.780132	*.793437	*.806430	*.819098	*.831428	*.843006	*.855024	
*.866266	*.877122	*.887580	*.897630	*.907262	*.916463	*.925225	*.933536	
*.390772	*.406807	*.422935	*.439138	*.454399	*.471699	*.488023	*.504352	
*.520689	*.536954	*.553220	*.569379	*.585579	*.601482	*.617372	*.633133	
*.648848	*.664201	*.679416	*.694558	*.709431	*.724080	*.738950	*.752644	
*.116301	*.109572	*.101570	*.092927	*.082915	*.071857	*.084306	*.095024	
*.036878	*.025417	*.014445	*.004296	*.0046172	*.012277	*.018507	*.023326	
*.026622	*.028233	-0.027969	-0.025577	-0.020594	-0.012238	*.00616	*.020562	
*.056775	-0.442134	-1.002884	-1.040202	-1.040447	-1.039858	-1.037151		

Figure A.1- Continued

-1.035076	-1.032488	-1.029356	-1.025681	-1.021475	-1.016743	-1.011454	-1.005527
-0.998884	-0.991494	-0.983347	-0.974426	-0.964716	-0.954221	-0.942944	-0.930861
-0.917924	-0.904013	-0.873309	-0.846225	-0.818266	-0.791222	-0.759074	-0.721018
-0.774088	-0.750428	-0.724749	-0.696757	-0.666477	-0.6337905	-0.6033842	-0.575333
-0.518083	-0.470680	-0.420476	-0.366002	-0.304548	-0.2733842	-0.2427804	-0.2151417
-0.056027	-0.186457	-0.342455	-0.526932	-0.730690	-0.929410	1.084056	1.154236
1.126336	1.004381	0.830887	0.624663	0.46604	0.312894	-0.182496	-0.06979
-0.027756	-0.112164	-0.185055	-0.248778	-0.305237	-0.355883	-0.01888	-0.444489
-0.485647	-0.525015	-0.560815	-0.593327	-0.623090	-0.650513	-0.676152	-0.700201
-0.722639	-0.743512	-0.762931	-0.781196	-0.798283	-0.806711	-0.814273	-0.82844
-0.855124	-0.867694	-0.878781	-0.888978	-0.898283	-0.906711	-0.914273	-0.920976
-0.928847	-0.931961	-0.936389	-0.940141	-0.943185	-0.95471	-0.964937	-0.97527
-0.946741	-0.940178	-0.511288	-0.080981	-0.91545	-0.12787	-0.25803	-0.33031
-0.135962	-0.135645	-0.132818	-0.128323	-0.122491	-0.115636	-0.107837	-0.09151
-0.089638	-0.079318	-0.068139	-0.056650	-0.043210	-0.029958	-0.016600	-0.003356
-0.09579	-0.022001	-0.033595	-0.043167	-0.052192	-0.067652	-0.077305	-0.07305
-0.095552	-0.105571	-0.126516	-0.151229	-0.178210	-0.206960	-0.236795	-0.267859
-0.300318	-0.333007	-0.365541	-0.400348	-0.433796	-0.480966	-0.312191	-0.312191
-0.785000	-0.120000	3.200000	500000	9903220	990346	986955	983051
-0.999496	-0.998714	-0.997407	-0.995575	-0.993220	-0.990346	-0.986955	-0.983051
-0.98636	-0.973715	-0.96893	-0.962314	-0.955963	-0.949067	-0.941691	-0.933844
-0.925533	-0.916769	-0.907562	-0.897920	-0.887856	-0.877380	-0.866502	-0.855233
-0.843587	-0.831573	-0.819206	-0.806497	-0.793460	-0.780109	-0.766457	-0.752520
-0.78311	-0.738466	-0.709141	-0.694210	-0.679010	-0.663736	-0.648225	-0.632552
-0.61735	-0.60791	-0.584135	-0.566584	-0.536070	-0.519740	-0.503384	-0.49384
-0.487021	-0.46667	-0.453340	-0.438058	-0.421818	-0.405697	-0.389653	-0.373724
-0.377927	-0.322279	-0.326797	-0.311499	-0.296401	-0.281521	-0.266874	-0.252478
-0.283474	-0.24498	-0.210945	-0.197105	-0.174792	-0.172219	-0.160001	-0.148150
-0.136681	-0.125605	-0.114935	-0.106681	-0.094654	-0.085465	-0.076523	-0.068038
-0.060016	-0.052468	-0.05400	-0.038819	-0.032731	-0.027141	-0.022055	-0.017479
-0.013421	-0.008888	-0.008887	-0.004422	-0.002497	-0.001117	-0.000283	-0.000000
-0.002668	-0.010807	-0.012453	-0.004364	-0.006817	-0.009897	-0.013330	-0.017381
-0.021952	-0.020335	-0.026264	-0.038714	-0.045300	-0.052375	-0.059934	-0.067969
-0.076471	-0.05433	-0.08844	-0.104695	-0.114977	-0.125678	-0.136768	-0.148290
-0.160176	-0.172433	-0.165045	-0.19800	-0.21282	-0.234816	-0.238768	-0.252941
-0.267379	-0.282067	-0.26987	-0.312123	-0.327558	-0.342974	-0.358655	-0.374482
-0.390439	-0.406506	-0.422668	-0.438906	-0.455802	-0.471559	-0.487900	-0.504266
-0.520620	-0.536945	-0.53223	-0.504364	-0.496817	-0.489897	-0.481330	-0.473724
-0.648962	-0.644441	-0.679740	-0.694843	-0.709335	-0.724400	-0.738823	-0.752989
-0.766883	-0.780491	-0.793798	-0.806792	-0.819458	-0.831783	-0.843756	-0.853364
-0.866596	-0.874339	-0.887883	-0.897918	-0.90532	-0.916716	-0.925660	-0.933752
-0.941585	-0.949949	-0.955837	-0.962243	-0.968160	-0.975584	-0.978508	-0.982930
-0.986844	-0.990246	-0.993133	-0.99503	-0.997351	-0.998676	-0.999477	-0.999477
-0.661722	-0.678267	-0.52478	-0.476984	-0.452167	-0.417576	-0.371523	-0.333319
-0.316752	-0.283125	-0.251496	-0.222221	-0.195997	-0.173576	-0.155526	-0.141933
-0.131485	-0.12800	-0.114686	-0.105950	-0.096014	-0.085154	-0.073817	-0.062308
-0.050930	-0.039923	-0.029502	-0.020011	-0.011871	-0.005315	-0.003131	-0.002888
-0.044276	-0.035459	-0.00424	-0.015487	-0.014865	-0.028888	-0.049898	-0.06166
-0.343447	-0.931343	-1.058572	-1.063377	-1.060836	-1.059817	-1.058293	-1.056303
-0.053852	-1.050898	-1.047403	-1.043362	-1.038780	-1.033653	-1.027946	-1.021515
-1.014464	-1.006578	-0.997900	-0.988404	-0.98066	-0.968879	-0.941911	-0.91911
-0.928048	-0.913204	-0.897327	-0.880361	-0.862242	-0.842884	-0.822155	-0.799897
-0.493569	-0.534662	-0.572243	-0.606392	-0.637568	-0.666271	-0.693099	-0.718234
-0.775993	-0.750364	-0.722870	-0.693239	-0.661162	-0.626443	-0.589444	-0.548510
-0.503198	-0.453370	-0.397119	-0.340790	-0.274958	-0.208899	-0.161697	-0.177590
-0.101185	-0.242234	-0.406111	-0.589317	-0.761097	-0.90601	-1.036012	-1.155336
1.12744	1.019544	0.860392	-0.681922	-0.564675	-0.364676	-0.206124	-0.08745
-0.012724	-0.091152	-0.175168	-0.242787	-0.303204	-0.37297	-0.405929	-0.450804
-0.493569	-0.534662	-0.572243	-0.606392	-0.637568	-0.666271	-0.693099	-0.718234
-0.741688	-0.763529	-0.783915	-0.802992	-0.80851	-0.87552	-0.853149	-0.867692
-0.881230	-0.893814	-0.905476	-0.916214	-0.926030	-0.94945	-0.942982	-0.950155
-0.956499	-0.962090	-0.967007	-0.971273	-0.974876	-0.97793	-0.980005	-0.981514
-0.982280	-0.981084	-0.961912	-0.602843	-0.31607	-0.01453	-0.066983	-0.082155
-0.092498	-0.097775	-0.09609	-0.098812	-0.096028	-0.091637	-0.085833	-0.078166
-0.070564	-0.061299	-0.059663	-0.054946	-0.027238	-0.013497	-0.01347	-0.011660
-0.024418	-0.036736	-0.048287	-0.058384	-0.066482	-0.074367	-0.082616	-0.092197

Figure A.1 - Continued

*105560	*121717	*142429	*167102	*194234	*223200	*253537	*285364
*318030	*351580	*785000	*120000	*100000	*400000	*311886	
*785000	*999495	*998712	*997403	*995568	*993210	*986935	*983025
*978604	*973676	*966247	*962320	*959051	*948997	*941613	*93357
*925438	*916666	*907800	*897724	*887728	*877245	*866360	*855086
*843434	*83116	*81945	*806333	*793294	*779942	*766290	*752533
*738116	*723684	*708983	*694057	*678922	*663595	*648091	*632427
*616619	*600684	*584638	*568496	*552282	*536006	*519688	*503344
*486992	*470649	*453333	*438061	*421851	*405720	*389686	*373166
*357976	*342236	*322860	*311568	*296476	*281599	*266956	*252562
*238433	*224384	*211032	*197792	*184877	*172302	*160082	*148229
*136756	*125677	*115002	*10744	*094913	*085520	*076573	*068083
*060057	*052504	*045431	*038846	*032754	*022071	*017492	
*013431	*009895	*006192	*004425	*002499	*001117	*000283	
*000268	*001088	*002455	*004368	*006823	*009816	*013342	*017396
*021971	*027059	*036252	*038748	*045339	*052420	*059986	*068027
*076535	*085503	*094921	*104779	*115067	*125775	*136889	*148399
*160291	*172553	*185170	*198129	*211414	*225011	*238905	*253079
*267518	*282205	*297124	*312258	*327589	*343101	*358777	*374598
*390547	*406607	*422760	*436989	*45275	*471602	*487952	*504306
*520649	*538962	*553229	*569432	*585554	*601578	*617489	*633269
*648901	*664310	*679661	*694756	*709640	*724298	*738716	*752877
*766767	*780312	*793678	*806670	*819336	*831662	*843637	*855247
*866462	*877329	*887778	*897817	*907437	*916627	*925377	*933676
*941516	*948887	*955182	*962194	*968118	*973547	*978478	*982905
*986824	*990231	*993123	*994496	*997347	*998674	*999476	
*704073	*639623	*589321	*53033	*495623	*441338	*400967	*362973
*327595	*293336	*262568	*233588	*207588	*185799	*168342	*154819
*144298	*135709	*127946	*119671	*110181	*099787	*089038	*078254
*067686	*07596	*048157	*039815	*02943	*027770	*024352	*022795
*023315	*026205	*031823	*040640	*033317	*070569	*090182	*-504042
*1.078807	*1.076694	*1.077608	*1.076711	*1.075380	*1.073596	*1.071393	*1.068795
*-0.65803	*-0.62357	*-0.58415	*-0.53968	*-0.49017	*-0.43556	*-0.37548	*-1.03906
*-1.023551	*-1.015448	*-1.005680	*-0.996924	*-0.964642	*-0.915191	*-0.863111	*-0.50187
*-9.6358	*.921550	*-9.05883	*-8.88679	*-8.047678	*-8.51025	*-8.30238	*-8.07996
*-7.84159	*-7.58567	*-7.31016	*-7.01291	*-6.69279	*-6.64999	*-5.98113	*-5.57771
*-5.12526	*-4.62284	*-4.08668	*-3.64592	*-2.93857	*-2.09914	*-1.25532	*-0.26512
*.091968	*.233173	*.397430	*.581348	*.714304	*.955541	*1.093194	*1.154931
*1.129584	*1.023851	*.866317	*.688938	*.514167	*.354541	*.214948	*.096079
*-0.04585	*-0.011169	*-1.67361	*-2.35160	*-2.95754	*-3.50016	*-3.98797	*-4.43434
*-4.86398	*-5.27812	*-5.65676	*-6.00070	*-6.31613	*-6.660629	*-6.87686	*-7.13033
*-7.36706	*-7.58770	*-7.79383	*-7.98693	*-8.33729	*-8.933729	*-8.49572	*-8.664370
*-8.78118	*-8.91051	*-9.03023	*-9.10095	*-9.24269	*-9.33568	*-9.42014	*-9.49623
*-9.56435	*-9.62535	*-9.68009	*-9.78889	*-9.77172	*-9.80847	*-9.83909	*-9.86386
*-9.88299	*-9.89607	*-9.90021	*-9.80750	*-9.66028	*-9.646661	*-9.616430	*-9.43833
*-9.06634	*-9.07122	*-9.077066	*-9.079620	*-9.077594	*-9.073809	*-9.068475	
*-0.61767	*-0.53807	*-0.44617	*-0.34196	*-0.22756	*-0.10690	*-0.01662	*-0.14059
*.026282	*.038088	*.049139	*.056838	*.066863	*.074005	*.081680	*.091101
*.103152	*.119014	*.139555	*.163718	*.190415	*.218635	*.247932	*.278242
*.309020	*.340098	*.370895	*.400514	*.422983	*.422556	*.279647	
*.785000	*.120000	*.300000	*.300000	*.300000			

Figure A.1 - Continued

*000269	*001089	*002458	*014374	*006832	*009828	*013354	*017419
*021999	*027093	*032693	*038795	*045394	*052484	*060057	*068106
*076624	*085600	*095027	*104694	*115191	*125906	*137029	*148466
*160445	*172713	*185336	*188299	*211589	*225189	*239084	*253554
*267698	*282383	*297299	*312429	*327756	*342262	*358530	*374742
*390682	*406731	*422873	*439089	*455363	*471675	*488011	*504551
*520678	*536976	*553227	*569415	*585522	*601532	*617428	*633195
*664271	*679550	*696635	*70510	*724160	*738571	*752726	
*766611	*780213	*793516	*806507	*819173	*831501	*843478	*855092
*866331	*877183	*881638	*897684	*907312	*916510	*925268	*933516
*941425	*948805	*955709	*962130	*968062	*973800	*978438	*982873
*986798	*990212	*993109	*995486	*997342	*998612	*999475	
*992502	*655482	*605101	*550634	*500571	*45605	*415351	*377414
*361461	*307860	*275876	*246807	*221916	*200841	*183231	*169860
*159582	*141442	*14193	*135523	*127787	*108135	*108637	*099086
*089958	*081515	*074023	*067872	*063517	*061236	*061140	*063419
*068426	*076635	*088747	*106271	*139393	*203590	*083451	*1091218
*-1.094785	-1.093610	-1.092881	-1.091601	-1.088507	-1.085302	-1.080170	
-1.076686	-1.072792	-1.066438	-1.062613	-1.058318	-1.052567	-1.046261	-1.039373
-1.031800	-1.023507	-1.014472	-1.004673	-0.994088	-0.982710	-0.970539	-0.957549
-0.943688	-0.948894	-0.913088	-0.896178	-0.878069	-0.837816	-0.815596	
-0.917555	-0.766243	-0.738805	-0.709089	-0.679592	-0.642553	-0.605960	-0.566461
-0.821557	-0.470943	-0.418747	-0.351941	-0.292318	-0.218507	-0.134265	-0.035357
-0.83094	-0.224417	-0.389017	-0.573606	-0.767688	-0.950597	-1.090412	-1.154435
1.*131605	1.*028040	1.*072108	1.*695813	1.*521724	1.*362487	1.*230307	1.*104242
*003507	*083211	*155561	*227524	*288277	*342692	*391688	*436211
*779184	*520923	*559063	*593589	*625432	*654814	*682126	*707683
*-731560	-0.753831	-0.774660	-0.794194	-0.812520	-0.829698	-0.845188	-0.860848
-0.974936	-0.988109	-0.90399	-0.911808	-0.923340	-0.932020	-0.940814	-0.948926
-0.956222	-0.962855	-0.988917	-0.974445	-0.979439	-0.983893	-0.987815	-0.991249
-0.994243	-0.996790	-0.998848	-1.000433	-0.998149	-0.993553	-0.992219	-0.991730
-0.104054	-0.02675	-0.044746	-0.052267	-0.053387	-0.057797	-0.056901	-0.054003
-0.049360	-0.031162	-0.035485	-0.026364	-0.016044	-0.004953	-0.006544	-0.018193
-0.029754	-0.04952	-0.051435	-0.060604	-0.068111	-0.074707	-0.090667	
-0.101827	-0.116652	-0.136982	-0.161606	-0.187616	-0.211920	-0.243500	-0.272636
*302397	*331981	*360940	*386762	*404828	*394883	*130523	
*790000	*105000	3.*000000	*500000	*993183	*990293	*986883	*982957
*999494	*998708	*991393	*995551	*992177	*985738	*941406	*933529
*978519	*973573	*968124	*962177	*887489	*887397	*865692	*854700
*925189	*916396	*901159	*897489	*792850	*779492	*765638	*751901
*843033	*811001	*818618	*805896	*693635	*678516	*663207	*647725
*737698	*723242	*708549	*588387	*552105	*535870	*519594	*503295
*616305	*600400	*584387	*588283	*436211	*422054	*405710	*374131
*486991	*4.0697	*4.54431	*3.212392	*2.971154	*2.822994	*2.677004	*2.533362
*358397	*342811	*225485	*211982	*198788	*185918	*173388	*161709
*2329283	*137962	*126919	*116278	*106052	*096250	*086682	*077059
*061481	*053940	*046872	*040284	*034182	*028572	*023457	*018836
*014696	*011020	*007814	*005108	*003936	*001335	*000344	*000000
*000329	*001307	*002894	*005053	*007747	*010943	*016409	*018743
*023359	*028472	*034084	*040190	*046785	*053663	*061418	*069444
*077934	*086881	*086275	*106107	*112366	*112043	*11824	*149599
*161455	*173678	*186255	*199172	*212414	*225566	*239813	*253640
*268331	*282968	*291836	*312919	*328199	*34358	*359281	*375049
*390946	*406954	*433055	*459232	*453468	*471745	*488047	*504355
*520652	*536922	*553147	*589310	*585395	*601985	*617264	*633014
*6488620	*666405	*677334	*694411	*709280	*723926	*738333	*752488
*766374	*771998	*793284	*806281	*818953	*831288	*843272	*854895
*666143	*817005	*881470	*897527	*907164	*916313	*925141	*934359
*941318	*948709	*95623	*96055	*967998	*973447	*978395	*982838
*986772	*990192	*993096	*994478	*997337	*998670	*999475	*993778
*667033	*578248	*510788	*455919	*409411	*367688	*329507	
*120195	*228405	*198506	*110683	*146211	*124224	*107647	*094899
*084752	*076268	*068299	*059533	*042998	*037902	*025844	*013399
*000768	*-0.11868	-0.047431	-0.036291	-0.047471	-0.057732	-0.067131	-0.075714

Figure A.1- Continued

-0.083445	-0.090268	-0.096128	-1.00942	-1.04445	-1.06158	-1.05885	-1.01728
-0.09399	-0.08065	-0.07053	-0.06463	-0.28437	-0.75293	-0.94113	-0.94947
-0.950180	-0.950026	-0.949061	-0.947359	-0.944960	-0.911668	-0.933371	-0.933371
1.101278	-0.923008	-0.916375	-0.908973	-0.907978	-0.891860	-0.871115	-0.871115
-0.88816	-0.885334	-0.853171	-0.821294	-0.801619	-0.779948	-0.753163	-0.753163
-0.86054	-0.733696	-0.712393	-0.689635	-0.665156	-0.638574	-0.609711	-0.578083
-0.504222	-0.464480	-0.420231	-0.375307	-0.323205	-0.259396	-0.179919	-0.087037
-0.802189	-0.812788	-0.816783	-0.746092	-0.421255	-0.616559	-0.871618	1.154841
1.101173	-0.116783	-0.116783	-0.742873	-0.535093	-0.363593	-0.232617	-0.126434
-0.070104	-0.151552	-0.216137	-0.270583	-0.317893	-0.360914	-0.401503	-0.440596
-0.47816	-0.513694	-0.505041	-0.527390	-0.599136	-0.623238	-0.645994	-0.667397
-0.682915	-0.705710	-0.722809	-0.738730	-0.753546	-0.767314	-0.780085	-0.791899
-0.802189	-0.812788	-0.816783	-0.810129	-0.837430	-0.843802	-0.849229	-0.853683
-0.85716	-0.859629	-0.861115	-0.861146	-0.860358	-0.854050	-0.764934	-0.794621
-0.133943	-0.169288	-0.180216	-0.188260	-0.202266	-0.202007	-0.198712	-0.193196
-0.186132	-0.177665	-0.168454	-0.158729	-0.148743	-0.138527	-0.128076	-0.117251
-0.106039	-0.094391	-0.082203	-0.081450	-0.069375	-0.056016	-0.042423	-0.028858
-0.002956	-0.009360	-0.113360	-0.137144	-0.163430	-0.191672	-0.215986	-0.065319
-0.071513	-0.289592	-0.317116	-0.351698	-0.387767	-0.425190	-0.459269	-0.252014
-0.790000	-0.110000	3.100000	5000000	995562	993201	990318	986918
-0.999495	-0.988711	-0.977400	-0.955652	-0.955845	-0.948934	-0.94542	-0.933002
-0.97857	-0.916575	-0.907352	-0.897696	-0.887617	-0.877127	-0.866236	-0.854956
-0.923354	-0.916575	-0.907352	-0.897696	-0.887617	-0.877127	-0.866236	-0.854956
-0.843298	-0.831276	-0.810898	-0.806182	-0.793140	-0.779785	-0.766131	-0.752194
-0.731987	-0.735257	-0.708828	-0.693905	-0.678776	-0.663495	-0.632304	-0.632304
-0.616056	-0.601052	-0.584550	-0.568424	-0.552224	-0.539565	-0.519659	-0.503141
-0.487010	-0.470650	-0.54398	-0.481052	-0.421968	-0.405884	-0.389858	-0.379866
-0.352597	-0.347153	-0.318192	-0.296832	-0.281988	-0.263778	-0.253016	-0.253016
-0.236892	-0.225105	-0.21586	-0.198377	-0.185377	-0.172951	-0.169393	-0.169393
-0.137495	-0.126444	-0.115796	-0.105564	-0.095757	-0.086385	-0.077459	-0.068887
-0.069977	-0.053435	-0.046370	-0.039786	-0.036300	-0.028087	-0.022881	-0.018373
-0.012527	-0.010625	-0.007486	-0.004862	-0.002777	-0.001255	-0.000921	-0.000621
-0.003037	-0.001226	-0.002734	-0.004806	-0.007418	-0.010546	-0.014168	-0.018277
-0.02881	-0.01984	-0.033587	-0.039686	-0.046276	-0.053350	-0.060903	-0.069929
-0.07719	-0.063367	-0.095763	-0.105597	-0.115988	-0.126581	-0.136277	-0.149108
-0.160971	-0.173202	-0.185788	-0.198715	-0.211968	-0.225533	-0.239394	-0.255353
-0.267941	-0.252596	-0.297483	-0.312585	-0.327885	-0.343366	-0.359011	-0.374802
-0.390712	-0.406755	-0.422881	-0.439084	-0.455335	-0.471648	-0.487775	-0.503039
-0.526032	-0.536926	-0.553136	-0.569362	-0.585470	-0.601481	-0.617380	-0.633149
-0.648772	-0.664234	-0.679518	-0.694607	-0.709488	-0.724144	-0.738559	-0.752120
-0.766610	-0.780211	-0.793525	-0.806520	-0.819180	-0.831521	-0.845501	-0.855117
-0.866358	-0.877211	-0.887666	-0.897713	-0.907339	-0.916526	-0.925592	-0.933398
-0.941445	-0.948823	-0.955725	-0.962144	-0.968075	-0.973512	-0.978449	-0.982882
-0.986866	-0.990218	-0.993144	-0.995490	-0.997344	-0.998476	-0.999476	-0.999476
-0.612060	-0.542263	-0.484843	-0.465226	-0.436527	-0.393388	-0.354627	-0.317341
-0.289591	-0.250492	-0.220011	-0.191700	-0.166228	-0.146614	-0.127576	-0.114562
-0.104289	-0.078765	-0.070908	-0.060910	-0.057964	-0.054298	-0.043444	-0.034344
-0.022341	-0.010490	-0.001008	-0.001818	-0.0021607	-0.0030195	-0.0037598	-0.0043804
-0.048113	-0.052183	-0.054057	-0.054126	-0.051973	-0.0466943	-0.038181	-0.022501
-0.003677	-0.031913	-0.038586	-0.058215	-0.065689	-0.0791816	-0.098962	-0.098962
-0.098772	-0.095517	-0.0983496	-0.080498	-0.0776937	-0.0728226	-0.068140	-0.068140
-0.956755	-0.999962	-0.942418	-0.934110	-0.925028	-0.915177	-0.905561	-0.892157
-0.880914	-0.867772	-0.853661	-0.838504	-0.822248	-0.804834	-0.765926	-0.766233
-0.744885	-0.723142	-0.741470	-0.758601	-0.774601	-0.789510	-0.803405	-0.816316
-0.501134	-0.456724	-0.410554	-0.360704	-0.303768	-0.236462	-0.155791	-0.062019
-0.042356	-0.042356	-0.304783	-0.304783	-0.696033	-0.910180	-0.1045137	-0.904864
-0.980227	-0.788544	-0.590153	-0.414890	-0.271618	-0.152183	-0.143271	-0.143271
-0.130952	-0.20108	-0.259138	-0.310915	-0.351695	-0.400833	-0.441708	-0.441708
-0.481331	-0.481331	-0.551996	-0.582171	-0.609755	-0.63534	-0.659223	-0.683338
-0.703439	-0.723142	-0.741470	-0.758601	-0.774601	-0.789510	-0.803405	-0.816316
-0.828284	-0.839386	-0.849616	-0.858973	-0.867491	-0.870587	-0.881198	-0.881767
-0.922070	-0.887466	-0.90436	-0.90436	-0.901537	-0.906189	-0.904864	-0.904864
-0.885096	-0.874732	-0.844556	-0.811546	-0.741377	-0.613177	-0.149558	-0.151691
-0.150672	-0.147253	-0.142036	-0.135536	-0.128156	-0.120073	-0.120073	-0.111307

Figure A.1- Continued

-.091786	-.091044	-.069575	-.051307	-.044381	-.031116	-.017796	-.004627
.008211	.020534	.032661	.042155	.050553	.058039	.066078	.075838
.088254	.104227	.124006	.148368	.174849	.203124	.23205	.263673
.295635	.328846	.362284	.398078	.433502	.461726	.455493	
.790000	.110000	3.000000	4.000000	5.000000	6.000000	7.000000	8.000000
.999494	.998709	.999395	.995553	.993187	.992999	.986691	.982968
.978533	.973590	.968445	.962201	.955765	.948843	.941441	.935677
.925230	.916440	.907206	.89540	.887452	.876952	.866053	.854675
.843100	.831071	.818890	.805971	.792926	.779570	.765916	.751980
.737776	.723320	.708826	.693710	.663277	.647790	.632146	
.616361	.600450	.584431	.568320	.552134	.535890	.519605	.503297
.486981	.470677	.454399	.438167	.421997	.405907	.389913	.374033
.358285	.342685	.327250	.311997	.296944	.282106	.267500	.253142
.239049	.225235	.211116	.198506	.185622	.173076	.160884	.149057
.137609	.126552	.115999	.105660	.095846	.086468	.077535	.069056
.061039	.053491	.046419	.039828	.033726	.028117	.023006	.018393
.016272	.010637	.00794	.004868	.002780	.001257	.000322	.000000
.000307	.001228	.002738	.004812	.007427	.010559	.014186	.018300
.022909	.028019	.033029	.039735	.046333	.053415	.060977	.069011
.077511	.086467	.095812	.105716	.115987	.126676	.137771	.149260
.161129	.173367	.185988	.198890	.212147	.225715	.253120	
.268126	.282779	.297663	.312761	.328055	.343530	.359187	.374950
.390860	.406882	.422996	.43186	.455434	.471723	.488035	.504353
.520660	.536939	.553173	.563344	.585437	.601433	.617317	.633073
.648683	.664132	.679004	.694484	.709355	.724002	.738410	.752565
.766451	.780054	.793380	.806354	.819024	.831359	.854959	
.866204	.877063	.887524	.895758	.907212	.916417	.925182	.933497
.941352	.948740	.955651	.962079	.968019	.973464	.978409	.982849
.986781	.990199	.993300	.995481	.997339	.998671	.999475	
.636410	.640716	.556503	.499191	.450753	.405588	.365623	.328020
.293174	.260239	.229559	.201334	.175962	.154478	.137313	.12093
.113807	.105382	.097654	.089290	.079585	.068861	.057661	.046281
.034932	.023819	.01385	.003352	.005331	.012689	.018776	.023377
.026568	.028133	.027895	.025620	.020866	.012892	.005950	.018256
.052386	.043688	.076636	.1.000209	.1.006338	.1.005756	.1.004676	.1.003128
-1.001128	.998638	.995226	.992097	.980662	.983529	.978469	.972803
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.088931	.0876158	.086299	.0847012	.080837	.0813476	.0794827	.0774803
.753395	.730614	.70601	.680092	.651635	.620822	.587472	.551138
.510749	.4466343	.419007	.369962	.313146	.245958	.165336	.071639
.032671	.150480	.294335	.475497	.687959	.904000	.1.076675	.1.158433
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.912668	.-996396	.-463761	.-666994	.-096097	.-114896	.-126421	.-132768
.135201	.-134626	.-131155	.-122199	.-124136	.-14705	.-10703	.-09592
.089316	.-079262	.-068381	.-056619	.-044129	.-031242	.-018235	.-005382
.007187	.-019251	.-030512	.-043362	.-048504	.-055709	.-063374	.-02708
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.999495	.998712	.99701	.995565	.993205	.990325	.986926	.983013
.978589	.973658	.968225	.962295	.955873	.948965	.941577	.933118
.925396	.916620	.90700	.89747	.887672	.877185	.8666297	.853020
.843365	.831344	.818911	.806257	.793216	.779862	.766210	.752273
.738066	.723605	.708804	.693980	.678848	.663524	.648024	.62365
.616562	.600633	.584593	.564461	.552253	.535985	.519576	.503342
.487001	.470670	.454366	.438107	.421910	.405793	.389773	.313867
.358093	.342468	.327009	.311733	.296656	.281797	.267110	.252793
.238680	.224849	.211313	.198089	.185191	.172632	.160427	.148590
.137132	.126667	.115006	.105160	.095341	.077023	.068541	

Figure A.1- Continued

.060523	.052975	.045906	.039321	.0271626	.022527
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.002477	.001154	.002589	.004580	.007113	.010175
.022427	.027524	.033124	.039222	.045813	.052891
.076984	.05942	.095348	.105195	.126164	.137264
.160637	.172883	.185484	.198426	.211695	.225276
.267733	.282403	.297306	.312423	.327739	.343235
.390636	.466682	.422822	.430037	.455311	.471626
.520640	.536944	.55202	.563396	.585511	.601529
.648836	.664301	.679588	.696681	.709563	.724220
.766688	.780293	.793600	.800594	.819262	.831591
.866419	.872269	.887721	.897764	.907388	.916581
.941480	.988854	.957563	.962169	.968096	.973529
.986815	.902224	.993118	.994493	.997345	.998674
.712384	.639555	.578073	.502795	.454998	.422331
.313145	.280187	.249769	.224487	.173973	.348771
.1323591	.124989	.117111	.108633	.098836	.080030
.053842	.026112	.031804	.027173	.012660	.004780
.-012137	.-0.053300	.-0.016836	.-0.016126	.-0.011659	.-0.00498
.-815832	.-1.034558	.-0.043369	.-1.03158	.-1.041805	.-1.000002
.-1.032307	.-1.038963	.-1.025108	.-1.020795	.-1.016006	.-1.010735
.-991487	.-933704	.-975190	.-965926	.-955895	.-945098
.-907998	.-883907	.-878838	.-862702	.-845417	.-826909
.-763229	.-713026	.-68935	.-654566	.-621943	.-591448
.-506443	.-458854	.-408561	.-35181	.-292877	.-22380
.-848478	.-808019	.-872310	.-882948	.-892732	.-140247
.-923747	.-929658	.-351188	.-535058	.-736799	.-938883
.1.126661	1.07695	.835295	.683332	.472924	.1.087036
.-018431	.-1.1731	.-1.175222	.-2.38142	.-2.95074	.-3.65635
.-475630	.-515034	.-550862	.-583365	.-613093	.-60565
.-713187	.-744263	.-753971	.-772445	.-789753	.-805951
.-848478	.-808019	.-872310	.-882948	.-892732	.-901683
.-923747	.-929658	.-351188	.-535058	.-736799	.-943937
.-955057	.-966461	.-957238	.-951185	.-941559	.-950602
.-057404	.-0.08232	.-0.074462	.-0.071274	.-0.07526	.-0.05747
.-060762	.-0.03122	.-0.044273	.-0.03220	.-0.023168	.-0.01497
.-024297	.-0.05728	.-0.06425	.-0.055798	.-0.065253	.-0.030377
.-0.086666	.-0.14234	.-0.133997	.-0.157153	.-0.183159	.-0.21056
.-295755	.-39169	.-371941	.-413708	.-440443	.-429280
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.990495	.998709	.997396	.995556	.993191	.990305
.978547	.973608	.968165	.962225	.955793	.948874
.925272	.916486	.920255	.989592	.887507	.870711
.842169	.881142	.818764	.886647	.793200	.779649
.737857	.772399	.708704	.663787	.678663	.663347
.616418	.600502	.584476	.568357	.552164	.535911
.486972	.410656	.454367	.438122	.421939	.405835
.358197	.342554	.327104	.311836	.296766	.281912
.238806	.244976	.211441	.198216	.185316	.172755
.137243	.126173	.115506	.105255	.095429	.086040
.060584	.053030	.045954	.039362	.033261	.02656
.013855	.010265	.007190	.004642	.002636	.001185
.000287	.001156	.002593	.00586	.007122	.010187
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.648748	.669201	.679476	.699558	.709431	.724080
.766530	.780132	.793437	.806430	.819098	.831428
.866266	.871122	.887580	.897630	.907262	.916463
.941388	.988772	.955679	.962104	.98040	.973481
.986789	.900205	.93104	.991484	.997340	.998671
.690361	.686356	.596546	.541305	.492031	.447898
.334847	.301909	.271024	.242687	.217650	.196797

Figure A.1- Continued

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*.070375	*.078016	-.089318	-.105996	*.135177	-.325555	-.937842	-.1060325
*1.060803	*1.059618	-1.05843	-1.056541	-1.054481	-1.052070	-1.049330	-1.046276
-.1.042894	*1.039123	-1.03912	-1.030251	-1.025139	-1.019570	-1.013510	-1.006873
1.128870	1.012243	-.841614	*.655645	*.480877	*.9328901	*.952488	*.940854
-.010176	-.094391	-.167231	-.2301919	-.287428	-.338159	-.834366	*.28446
-.468225	-.507922	-.54409	-.576739	-.606829	-.634609	-.629874	-.594864
-.707858	-.729149	-.749073	-.761762	-.785206	-.801715	-.817107	-.831525
-.516931	-.467638	-.417005	-.362583	-.301383	-.231014	-.14903	-.053413
*.057361	*.187132	*.342448	*.526322	*.729709	*.898451	*.1.04227	*.1.05726
1.128870	1.012243	-.841614	*.655645	*.480877	*.9328901	*.952488	*.940854
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*.738438	*.723972	*.709663	*.694329	*.679185	*.663846	*.656124	*.652651
*.616828	*.600876	*.584613	*.568655	*.552420	*.536124	*.519786	*.503422
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*.238294	*.224444	*.210991	*.197652	*.184739	*.172167	*.168102	
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*.649000	*.664487	*.679993	*.694902	*.709800	*.724470	*.738897	*.753067
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*.866678	*.877518	*.887959	*.897990	*.907601	*.916781	*.925559	*.933807
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*.1.064847	*.1.064851	*.1.063664	*.1.062705	*.1.061028	*.1.058946	*.1.056476	*.1.053634
*.1.050407	*.1.046746	*.1.042604	*.1.037969	*.1.032838	*.1.027202	*.1.02021	*.1.014210
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*.110888	*.251190	*.594488	*.566149	*.586578	*.604671	*.1.057352	
*.1.29748	*.1.022688	*.864665	*.681359	*.512968	*.353863	*.214830	*.046516
*.003612	*.089689	*.16521	*.222817	*.293064	*.347038	*.39504	*.40383
*.483482	*.524526	*.562120	*.596212	*.622779	*.656062	*.683028	*.708300
*.731889	*.753891	*.774478	*.793789	*.811900	*.828863	*.844731	*.859557
*.873395	*.886306	*.898325	*.909456	*.919705	*.929096	*.937555	*.945401
*.952372	*.958653	*.964327	*.964431	*.973972	*.977949	*.981371	*.984277

Figure A.1- Continued

-.986706	-.988653	-.990077	-.940993	-.988242	-.818706	-.292617	-.036788
-.014117	-.026716	-.034750	-.018348	-.01187	-.041187	-.038166	-.040682
-.005036	-.028035	-.020661	-.016844	-.01832	-.008984	-.020251	-.031692
-.033896	-.028035	-.020661	-.016844	-.018297	-.008052	-.095449	-.104781
.043063	.054121	.064532	.073707	.081297	.088052	.258817	.288613
.116709	.132019	.151811	.175703	.201995	.229757	.258817	.297725
.319390	.351326	.381599	.410948	.435629	.435114	.435114	.435114
.790000	.120000	.3.200000	.4.000000	.993220	.990346	.986955	.983051
.999496	.998714	.997407	.995575	.955963	.949067	.941691	.933844
.978636	.913715	.968293	.962374	.807856	.817380	.866502	.955233
.925533	.916769	.907562	.897920	.806497	.793460	.780109	.766457
.843587	.831573	.819206	.804681	.694210	.679070	.663736	.648225
.738311	.723846	.709141	.584735	.568584	.52357	.536070	.519740
.616735	.600791	.584735	.454340	.438058	.421638	.405697	.389653
.487021	.470667	.454340	.326197	.311499	.296401	.281521	.262478
.357927	.342279	.210945	.197705	.184792	.172219	.160001	.148150
.238347	.224498	.114935	.104681	.094854	.085465	.076523	.068038
.136681	.125605	.114935	.104681	.098819	.032731	.027141	.022055
.060016	.052468	.045400	.038819	.032731	.027141	.017479	.000000
.013421	.009888	.006887	.004422	.002497	.001117	.000283	.000000
.000268	.001087	.002453	.003634	.006817	.009807	.013330	.017381
.021952	.02035	.032624	.038714	.045380	.052375	.059934	.067969
.076471	.085433	.094844	.104695	.114977	.125678	.136786	.148290
.160176	.172433	.185045	.198000	.211282	.224876	.238768	.252941
.267379	.282067	.296987	.312123	.327458	.342974	.358655	.374482
.390439	.406506	.422668	.438906	.455202	.471539	.487900	.504266
.520620	.536945	.553223	.569438	.585571	.601607	.617529	.633319
.648962	.664441	.679740	.694843	.709735	.724400	.738823	.752989
.766883	.780491	.793798	.806792	.819458	.831783	.843756	.855364
.866596	.877439	.887883	.897918	.907532	.916716	.925460	.933752
.941585	.948949	.958637	.962243	.966160	.973584	.978508	.982930
.986844	.990246	.993133	.995503	.997351	.998676	.999477	.999477
.677429	.655493	.611041	.550169	.506062	.458121	.418405	.379138
.343855	.312197	.282395	.253812	.227282	.205221	.189006	.176819
.166921	.158948	.151879	.144444	.135975	.126783	.117412	.108205
.099439	.091385	.084363	.078728	.075021	.073579	.074652	.078633
.086183	.098316	.116731	.117579	.105555	.614497	.672818	.7060633
.1.079708	.1.076437	.1.076894	.1.075026	.1.072800	.1.070217	.1.067791	.1.064035
-.1.060432	-.1.056426	-.1.051970	-.1.07047	-.1.041653	-.1.05777	-.1.029380	-.1.022375
-.1.014682	-.1.006265	-.997104	-.987173	-.976452	-.964937	-.952626	-.939482
-.925443	-.904345	-.894381	-.877203	-.859842	-.83946	-.818329	-.795965
-.772017	-.746325	-.718687	-.688892	-.656862	-.625251	-.585762	-.545608
-.500427	-.550233	-.396311	-.375889	-.272067	-.198396	-.114356	-.015771
.102132	.242572	.405841	.588537	.780063	.959790	.1.096100	.1.156927
1.131890	1.020189	.870766	.646263	.520961	.362264	.23413	.165120
.004904	-.081319	-.157220	-.224787	-.285203	-.339339	-.388045	-.422747
-.475792	-.511125	-.554957	-.59390	-.620893	-.649900	-.677045	-.702516
-.726320	-.748538	-.769332	-.788843	-.807153	-.824319	-.840398	-.855445
-.869517	-.882675	-.894955	-.903660	-.916894	-.926583	-.935454	-.943530
-.950453	-.957514	-.963603	-.969161	-.974198	-.982716	-.982730	-.986287
-.984439	-.982190	-.994515	-.996384	-.997814	-.998160	-.933117	-.396480
.052275	.022182	.004670	-.006879	-.014455	-.018950	-.020863	-.020473
-.018116	-.013983	-.018169	-.00738	-.017763	-.028033	-.038566	-.038566
.049172	.059523	.069278	.077833	.084857	.091112	.097966	.106601
.118509	.134325	.153382	.175423	.200531	.228447	.258123	.287532
.315615	.345390	.376504	.402899	.417185	.411624	.4228052	
.790000	.120000	3.100000	.300000	.993210	.990331	.986935	.983025
.999495	.998712	.997403	.995568	.993210	.990331	.986935	.983025
.978604	.913676	.968267	.963220	.955901	.948997	.941613	.933157
.925438	.916666	.907450	.897800	.8877245	.8863360	.855086	
.843434	.831416	.819045	.806333	.793294	.7779942	.766290	.75353
.738146	.726684	.708983	.694057	.678922	.663595	.648091	.624247
.616619	.606884	.584638	.564499	.536006	.519688	.50344	
.486992	.476649	.454333	.438061	.421851	.405720	.389686	.377766
.357976	.342336	.326860	.311568	.296476	.281599	.266955	.252562?

Figure A.1- Continued

*224584	*197792	*184877	*172302	*160082	*148229
*125677	*115002	*104744	*094913	*085520	*076573
*136746	*012504	*045431	*038846	*032754	*022071
*06057	*027059	*032652	*038764	*045339	*027161
*013431	*009895	*006892	*004425	*002499	*001117
*000268	*001048	*000455	*000436	*0006823	*0009816
*021971	*027059	*032652	*038764	*045339	*052420
*076555	*085503	*094921	*104779	*115067	*125775
*160291	*172553	*185170	*198129	*211414	*225011
*267518	*282205	*297124	*312525	*327589	*343101
*390547	*406607	*422760	*438989	*455275	*471602
*520649	*536962	*553229	*569432	*585554	*601578
*648901	*664370	*679661	*694756	*709640	*724298
*766767	*780372	*793678	*806670	*819336	*831662
*866682	*877329	*887776	*897817	*907437	*916627
*941516	*948897	*955782	*962194	*968118	*973547
*986824	*990231	*993123	*995496	*997347	*998674
*724661	*700654	*643964	*580662	*529706	*483172
*368888	*335476	*304723	*276311	*251201	*230350
*191558	*183508	*176799	*170313	*162774	*154319
*130674	*124002	*118483	*114578	*112674	*112683
*129641	*146693	*093261	*500491	*105961	*109203
*1.092552	-1.090658	-1.088552	-1.086169	-1.083469	-1.080450
-1.069568	-1.065222	-1.060467	-1.055268	-1.049622	-1.035250
-1.021892	-1.013341	-1.004065	-0.994035	-0.983230	-0.971643
-*932061	-*917099	-*901132	-*884064	-*865804	-*846263
-*779834	-*753444	-*725938	-*696177	-*664021	-*629609
-*508982	-*508304	-*404112	-*345468	-*280667	-*201635
*093698	*234239	*397819	*581132	*773708	*955011
1.133928	1.031386	.876761	.701787	.528868	.370601
*013437	*027011	*148959	*216679	*277245	*331522
-*467980	-*509620	-*547717	-*582304	-*614130	-*643477
-*720485	-*742892	-*763875	-*785577	-*802081	-*819446
-*865310	-*865310	-*902939	-*913764	-*937361	-*855734
-*949995	-*956169	-*962705	-*968743	-*974292	-*979355
-*991960	-*995452	-*998597	-1.001382	-1.03802	-1.05895
-*727051	-*177580	*065036	*040480	*025063	*017473
*003895	*005236	*008719	*014053	*020966	*029122
*056614	*066001	*074961	*082505	*088406	*093964
*118984	*132672	*151370	*173692	*198227	*224457
*307133	*335201	*361003	*383012	*391899	*378809
*790000	*120000	3.000000	*200000	*993196	*990312
*994495	*998710	*997398	*995559	*995822	*948907
*978562	*973626	*968187	*962250	*887565	*877072
*925316	*916533	*907306	*897646	*793084	*779730
*843239	*831216	*818840	*806125	*768825	*766708
*737939	*723461	*708785	*693865	*678738	*663420
*616477	*600554	*584522	*568396	*552194	*535932
*486962	*470634	*454333	*438075	*421879	*388739
*358051	*342420	*326954	*311669	*296583	*281712
*238556	*224709	*211157	*197916	*185000	*172423
*136865	*125781	*115101	*104836	*094999	*085599
*060117	*052557	*045478	*038887	*032789	*027190
*013445	*009906	*006899	*004430	*002502	*001118
*000269	*001089	*002458	*004374	*006832	*009828
*021999	*027093	*032693	*038795	*045394	*052484
*076624	*085600	*095027	*104894	*115191	*125906
*160445	*172713	*185336	*198299	*211589	*225189
*267698	*282383	*297299	*312429	*327756	*343262
*390682	*406731	*422873	*439089	*455363	*471675
*520678	*536976	*553227	*569415	*585522	*601532
*648814	*664271	*679550	*694635	*724160	*73571
*866331	*877183	*887638	*897684	*907312	*916510
*941425	*948805	*955709	*962130	*968062	*973500
*986798	*990212	*993109	*995486	*997342	*998672

Figure A.1- Continued

*.794188	*.734961	*.651940	*.580487	*.536725	*.500650	*.463153	*.422861
*.3H6712	*.353493	*.323188	*.295173	*.269744	*.248643	*.23276	*.219981
*.210926	*.204083	*.197930	*.191193	*.183721	*.175977	*.168081	*.160238
*.153020	*.146792	*.141623	*.137865	*.136224	*.137165	*.140625	*.145853
*.151211	*.144249	*.112778	*.112778	*.1110462	*.108555	*.105547	*.104388
*.102093	*.1099669	*.1097088	*.1094280	*.1091199	*.1087841	*.108419	*.1080337
*.076173	*.1071662	*.1066748	*.1061411	*.1055642	*.1049429	*.104225	*.1035441
*.027492	*.1018842	*.1009473	*.999365	*.988499	*.976874	*.964487	*.951310
*.937288	*.922356	*.906438	*.889449	*.871299	*.851890	*.831100	*.808799
*.784905	*.759358	*.731991	*.702465	*.670459	*.636059	*.59950	*.560154
*.516052	*.465705	*.411174	*.352555	*.287287	*.213918	*.130200	*.031888
*.085890	*.226471	*.390255	*.574131	*.767656	*.950421	*.1090122	*.155748
1.135906	1.035752	*.882932	*.709221	*.537127	*.379365	*.240992	*.122855
*.022566	*.063864	*.140024	*.1207861	*.268543	*.322929	*.371849	*.416449
*.459127	*.501084	*.539652	*.574198	*.606283	*.635984	*.663352	*.689355
*.713497	*.736052	*.757180	*.777029	*.795669	*.813222	*.822990	*.845150
*.859661	*.873280	*.886042	*.897953	*.909033	*.919284	*.92866	*.937495
*.945517	*.92926	*.959816	*.966226	*.972169	*.977653	*.98204	*.987386
*.991770	*.995883	*.999718	*.1003268	*.1006541	*.1009588	*.101279	*.1015237
*.017921	-1.020385	*.872594	*.301626	*.094913	*.071633	*.05874	*.049323
*.044035	*.041505	*.041449	*.043779	*.048090	*.053720	*.060383	*.067648
*.075444	*.083082	*.090228	*.096585	*.101688	*.105751	*.110518	*.116709
*.126274	*.139410	*.156659	*.177537	*.201205	*.226828	*.252274	*.278578
*.302468	*.350649	*.365417	*.394931	*.399119	*.361190	*.00447	

Figure A.1- Concluded

• TEST CASE •

PROGRAM PERTURB
CALCULATES NONLINEAR MULTIPLE-PARAMETER
CONTINUOUS OR DISCONTINUOUS
PERTURBATION SOLUTIONS
WHICH REPRESENT CHANGES IN EITHER
GEOMETRY OR FLOW CONDITIONS
BY EMPLOYING A STRAINED-COORDINATE PROCEDURE
UTILIZING UNIT PERTURBATIONS DETERMINED FROM
PREVIOUSLY CALCULATED
"BASE" AND "CALIBRATION" SOLUTIONS
DISPLACED FROM ONE ANOTHER BY SOME REASONABLE
CHANGE IN GEOMETRY OR FLOW CONDITION
WRITTEN BY
JAMES P. ELLIOTT AND STEPHEN S. STAHLAWA
NIELSEN ENGINEERING AND RESEARCH, INC.
MOUNTAIN VIEW, CALIFORNIA

Figure A.2– Abbreviated print output for sample case

```
*****  
* TEST CASE - 4 PARAMETER PERTURBATION OF A SUPERCRITICAL CASCADE FLOW *  
*****  
  
.....LIST OF INPUT PARAMETERS  
  
N = 191  
A = -1.0    B =  1.0  
NPARAM = 4  
  
.....STRAINING OPTIONS  
  
NUMBER OF FIXED POINTS: 5  
  
FIXED POINTS WILL BE AUTOMATICALLY DETERMINED  
BY THE PROGRAM FOR ALL SOLUTIONS AS FOLLOWS:  
  
TWO END POINTS  
POINT OF MAXIMUM CP  
CPCRIT (1ST POINT)  
CPCRIT (2ND POINT)
```

Figure A.2- Continued

RESULTS OF COMPUTATIONS ON BASE SOLUTION:

.....MACH NUMBER,
VALUES OF PERTURBATION PARAMETERS,
CRITICAL VALUE OF CP:

M0 = .7800
Q0(1) = .7800 (MACH NO.)
Q0(2) = .1100 (TAU)
Q0(3) = 3.2000 (PITCH)
Q0(4) = .3000 (ALPHA)
CPCRIT = -.4940

.....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.
(* DENOTES POINT ON LOWER SURFACE)

MINIMUM AT X = .4220* (POINT NO. 53)
MAXIMUM AT X = 0.0000 (POINT NO. 96)
2 CRITICAL POINT(S):
1ST AT X = .5001* (AFTER POINT NO. 48)
2ND AT X = .4273 (AFTER POINT NO. 139)

.....LOCATION OF FIXED POINTS
(* DENOTES POINT ON LOWER SURFACE)

XFIX(1) = 1.0000*
XFIX(2) = .5001*
XFIX(3) = 0.0000
XFIX(4) = .4273
XFIX(5) = 1.0000

Figure A.2- Continued

RESULTS OF COMPUTATIONS ON 1ST CALIBRATION SOLN:

.....MACH NUMBER,
VALUES OF PERTURBATION PARAMETERS,
CRITICAL VALUE OF CP:

M1 = .7900

(** DENOTES PERTURBATION FROM BASE VALUE)

**Q1(1) = .7900 (MACH NO.)

Q1(2) = .1100 (TAU)

Q1(3) = 3.2000 (PITCH)

Q1(4) = .3000 (ALPHA1)

CPCRIT = -.4638

.....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.
(* DENOTES POINT ON LOWER SURFACE)

MINIMUM AT X = .5197* (POINT NO. 47)

MAXIMUM AT X = 0.0000 (POINT NO. 96)

2 CRITICAL POINT(S):

1ST AT X = .5788* (AFTER POINT NO. 43)

2ND AT X = .5149 (AFTER POINT NO. 144)

.....LOCATION OF FIXED POINTS
(* DENOTES POINT ON LOWER SURFACE)

XFIX(1) = 1.0000*

XFIX(2) = .5788*

XFIX(3) = 0.0000

XFIX(4) = .5149

XFIX(5) = 1.0000

Figure A.2 - Continued

RESULTS OF COMPUTATIONS ON 2ND CALIBRATION SOLN:

.....MACH NUMBER,
VALUES OF PERTURBATION PARAMETERS,
CRITICAL VALUE OF CP:

M1 = .7800

(** DENOTES PERTURBATION FROM BASE VALUE)

Q1(1) = .7800 (MACH NO.)

**Q1(2) = .1200 (TAU)

Q1(3) = 3.2000 (PITCH)

Q1(4) = .3000 (ALPHA1)

CPCRIT = -.4940

.....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.
(* DENOTES POINT ON LOWER SURFACE)

MINIMUM AT X = .5197* (POINT NO. 47)

MAXIMUM AT X = 0.0000 (POINT NO. 96)

2 CRITICAL POINT(S):

1ST AT X = .5774* (AFTER POINT NO. 43)

2ND AT X = .5175 (AFTER POINT NO. 144)

.....LOCATION OF FIXED POINTS
(* DENOTES POINT ON LOWER SURFACE)

XFIX(1) = 1.0000*

XFIX(2) = .5774*

XFIX(3) = 0.0000

XFIX(4) = .5175

XFIX(5) = 1.0000

Figure A.2- Continued

Figure A.2- Continued

RESULTS OF COMPUTATIONS ON 3RD CALIBRATION SOLN:

.....MACH NUMBER,
VALUES OF PERTURBATION PARAMETERS,
CRITICAL VALUE OF CP:

M1 = .7800

(** DENOTES PERTURBATION FROM BASE VALUE)

Q1(1) = .7800 (MACH NO.)

Q1(2) = .1100 (TAU)

**Q1(3) = 3.0000 (PITCH)

Q1(4) = .3000 (ALPHA1)

CPCRIT = -.4940

.....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.
(* DENOTES POINT ON LOWER SURFACE)

MINIMUM AT X = .4382* (POINT NO. 52)

MAXIMUM AT X = 0.0000 (POINT NO. 96)

2 CRITICAL POINT(S):

1ST AT X = .5165* (AFTER POINT NO. 47)

2ND AT X = .4520 (AFTER POINT NO. 140)

.....LOCATION OF FIXED POINTS
(* DENOTES POINT ON LOWER SURFACE)

XFIX(1) = 1.0000*

XFIX(2) = .5165*

XFIX(3) = 0.0000

XFIX(4) = .4520

XFIX(5) = 1.0000

RESULTS OF COMPUTATIONS ON 4TH CALIBRATION SOLN:

.....MACH NUMBER.
VALUES OF PERTURBATION PARAMETERS.
CRITICAL VALUE OF CP:

M1 = .7800

(** DENOTES PERTURBATION FROM BASE VALUE)

Q1(1) = .7800 (MACH NO.)

Q1(2) = .1100 (TAU)

Q1(3) = 3.2000 (PITCH)

**Q1(4) = .5000 (ALPHA1)

CPCRIT = -.4940

.....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.
(* DENOTES POINT ON LOWER SURFACE)

MINIMUM AT X = .4058* (POINT NO. 54)

MAXIMUM AT X = 0.0000 (POINT NO. 96)

2 CRITICAL POINT(S):

1ST AT X = .4903* (AFTER POINT NO. 48)

2ND AT X = .4360 (AFTER POINT NO. 139)

.....LOCATION OF FIXED POINTS
(* DENOTES POINT ON LOWER SURFACE)

XFIX(1) = 1.0000*

XFIX(2) = .4903*

XFIX(3) = 0.0000

XFIX(4) = .4360

XFIX(5) = 1.0000

Figure A.2- Continued

UNIT PERTURBATION OF CP AND UNIT STRAINING OF XBASE
FOR CALIBRATION SOLUTIONS 1 THROUGH 4

POINT	XBASE	* 1ST CALB SOLN *		* 2ND CALB SOLN *		* 3RD CALB SOLN *		* 4TH CALB SOLN *	
		XSTRUNIT	CPUNIT	XSTRUNIT	CPUNIT	XSTRUNIT	CPUNIT	XSTRUNIT	CPUNIT
1	.9995	.9996	-3.6772	.9996	13.0723	.9995	-0.0683	.9995	-.0125
2	.9987	.9989	5.8554	.9989	10.2272	.9988	-.0697	.9987	-.0134
3	.9974	.9978	6.7015	.9975	8.9041	.9975	-.0590	.9974	-.0264
4	.9956	.9963	5.8289	.9963	8.3875	.9957	-.0538	.9956	-.0263
5	.9932	.9943	5.6358	.9943	7.9171	.9934	-.0508	.9931	-.0260
6	.9903	.9919	5.6107	.9918	7.6585	.9907	-.0492	.9901	-.0255
7	.9869	.9890	5.4424	.9890	7.4524	.9874	-.0480	.9887	-.0258
8	.9830	.9857	5.4475	.9857	7.3280	.9836	-.0472	.9827	-.0261
9	.9786	.9820	5.4801	.9819	7.2282	.9793	-.0464	.9742	-.0268
10	.9737	.9778	5.5339	.9778	7.1532	.9745	-.0454	.9732	-.0273
11	.9683	.9732	5.5788	.9732	7.0535	.9693	-.0439	.9676	-.0270
12	.9623	.9683	5.5916	.9681	6.9361	.9636	-.0421	.9616	-.0262
13	.9559	.9628	5.5684	.9627	6.7704	.9574	-.0393	.9550	-.0242
14	.9490	.9570	5.3490	.9569	6.4648	.9507	-.0349	.9480	-.0223
15	.9416	.9508	5.0069	.9506	6.0231	.9435	-.0298	.9405	-.0212
16	.9338	.9442	4.6607	.9440	5.5879	.9359	-.0256	.9325	-.0200
17	.9255	.9372	4.3109	.9370	5.2574	.9279	-.0227	.9240	-.0193
18	.9167	.9298	4.2266	.9296	5.0365	.9194	-.0211	.9150	-.0195
19	.9075	.9220	4.2128	.9218	4.9532	.9105	-.0213	.9056	-.0209
20	.8978	.9139	4.3770	.9136	5.0572	.9012	-.0238	.8988	-.0233
21	.8877	.9054	4.7068	.9051	5.3353	.8914	-.0279	.8855	-.0257
22	.8773	.8966	5.0993	.8962	5.6726	.8813	-.0317	.8748	-.0278
23	.8664	.8874	5.4668	.8870	5.9736	.8708	-.0347	.8637	-.0297
24	.8551	.8779	5.7992	.8775	6.2320	.8598	-.0374	.8522	-.0316
25	.8435	.8681	6.1138	.8677	6.4713	.8486	-.0399	.8435	-.0335
26	.8314	.8579	6.4557	.8575	6.7004	.8370	-.0422	.8281	-.0353
27	.8191	.8475	6.7070	.8470	6.9180	.8250	-.0444	.8155	-.0369
28	.8063	.8368	6.9167	.8363	7.1126	.8127	-.0461	.8035	-.0382
29	.7933	.8258	7.2083	.8253	7.2699	.8001	-.0472	.7892	-.0394
30	.7800	.8146	7.4019	.8140	7.3994	.7872	-.0483	.7756	-.0410
31	.7663	.8031	7.6082	.8024	7.5350	.7740	-.0500	.7617	-.0429
32	.7524	.7913	7.8418	.7906	7.7081	.7605	-.0523	.7451	-.0451
33	.7382	.7793	8.1228	.7796	7.9326	.7467	-.0551	.7330	-.0477
34	.7237	.7672	8.4590	.7664	8.2092	.7327	-.0586	.7182	-.0508
35	.7090	.7548	8.8554	.7540	8.5568	.7185	-.0629	.7032	-.0544
36	.6941	.7422	9.3080	.7414	8.9729	.7041	-.0682	.6880	-.0580
37	.6789	.7294	9.8246	.7286	9.4572	.6894	-.0741	.6726	-.0625
38	.6636	.7165	10.3836	.7156	9.9922	.6746	-.0803	.6550	-.0665
39	.6481	.7034	10.9615	.7025	10.5565	.6596	-.0867	.6411	-.0722
40	.6324	.6902	11.5312	.6893	11.1326	.6445	-.0934	.6252	-.0743
41	.6166	.6769	12.0919	.6759	11.7063	.6292	-.1003	.6091	-.0774
42	.6007	.6635	12.6054	.6622	12.2629	.6138	-.1075	.5988	-.0793
43	.5846	.6500	13.0992	.6489	12.7943	.5982	-.1142	.5744	-.0793
44	.5685	.6364	13.4932	.6352	13.2977	.5826	-.1197	.5600	-.0772
45	.5523	.6227	13.783	.6215	13.6960	.5670	-.1230	.5455	-.0755
46	.5360	.6090	13.8275	.6078	14.0485	.5512	-.1238	.5289	-.0743
47	.5197	.5953	14.0263	.5940	14.2971	.5354	-.1061	.5102	-.0745
48	.5034	.5815	14.1470	.5802	14.2397	.5196	-.0909	.4936	1.2524
49	.4870	.5636	6.1472	.5623	5.9650	.4794	-.0937	.4744	1.4047
50	.4707	.5447	2.8708	.5436	9.4729	.4861	-.0764	.4674	0.7666
51	.4544	.5258	-2.4337	.5246	-9.1191	.4693	-.0681	.4454	0.1006
52	.4381	.5070	-2.2079	.5058	-8.8921	.4525	-.0642	.4225	0.0969
53	.4220	.4883	-2.0468	.4872	-8.7206	.4358	-.0611	.4136	0.0953
54	.4058	.4696	-1.9189	.4685	-8.5696	.4191	-.0583	.3947	0.0947
55	.3898	.4511	-1.8124	.4501	-8.4372	.4026	-.0561	.3821	0.0946

Figure A.2- Continued

5.6	*3719	-1.7337	*4317	-8.3259	*3862	*0545	*3665	*0444	
5.7	*3582	*4145	-1.6884	*4135	-8.2398	*3699	*0533	*3511	*0555
5.8	*3425	*3964	-1.6684	*3955	-8.1763	*3537	*0525	*3358	*0628
5.9	*3271	*3785	-1.6712	*3776	-8.1262	*3378	*0518	*3206	*0969
6.0	*3118	*3608	-1.6843	*3600	-8.0786	*3220	*0513	*3057	*0978
6.1	*2968	*3434	-1.7024	*3426	-8.0281	*3065	*0510	*2909	*0986
6.2	*2819	*3262	-1.7170	*3255	-7.9622	*2911	*0507	*2763	*0994
6.3	*2673	*3093	-1.7238	*3086	-7.8822	*2760	*0503	*2620	*1003
6.4	*2529	*2927	-1.7277	*2920	-7.7894	*2612	*0503	*2479	*1012
6.5	*2388	*2764	-1.7330	*2757	-7.6871	*2466	*0503	*2341	*1021
6.6	*2250	*2604	-1.7435	*2598	-7.5777	*2324	*0504	*2206	*1031
6.7	*2115	*2447	-1.7605	*2442	-7.4611	*2184	*0505	*2073	*1041
6.8	*1983	*2295	-1.7843	*2289	-7.3370	*2048	*0505	*1944	*1052
6.9	*1854	*2146	-1.8128	*2141	-7.2045	*1915	*0503	*1817	*1065
7.0	*1729	*2000	-1.8416	*1996	-7.0621	*1785	*0499	*1695	*1081
7.1	*1607	*1859	-1.8697	*1855	-6.9127	*1659	*0493	*1575	*1101
7.2	*1489	*1723	-1.8914	*1719	-6.7550	*1537	*0486	*1459	*1123
7.3	*1374	*1590	-1.9037	*1586	-6.5767	*1419	*0482	*1347	*1143
7.4	*1264	*1462	-1.9117	*1459	-6.3754	*1305	*0480	*1239	*1161
7.5	*1157	*1339	-1.9295	*1336	-6.1490	*1195	*0482	*1134	*1174
7.6	*1055	*1021	-1.9753	*1218	-5.9059	*1089	*0485	*1034	*1187
7.7	*0957	*1107	-2.0420	*1105	-5.6470	*0988	*0484	*0938	*1204
7.8	*0863	*0999	-2.1030	*0997	-5.3671	*0892	*0480	*0846	*1235
7.9	*0774	*0896	-2.1597	*0894	-5.0885	*0799	*0477	*0759	*1282
8.0	*0689	*0798	-2.2410	*0796	-4.8528	*0712	*0478	*0678	*1350
8.1	*0609	*0705	-2.4085	*0703	-4.6923	*0629	*0490	*0597	*1398
8.2	*0534	*0618	-2.5574	*0616	-4.4183	*0551	*0493	*0523	*1379
8.3	*0463	*0536	-2.4802	*0535	-3.7783	*0479	*0469	*0454	*1362
8.4	*0388	*0460	-2.4087	*0459	-3.0266	*0411	*0456	*0390	*1392
8.5	*0337	*0390	-2.5850	*0389	-2.4638	*0348	*0469	*0330	*1443
8.6	*0281	*0325	-2.9590	*0324	-2.1314	*0290	*0500	*0275	*1498
8.7	*0230	*0266	-3.4047	*0265	-2.2073	*0237	*0537	*0225	*1533
8.8	*0184	*0212	-3.6285	*0212	-2.3516	*0190	*0547	*0180	*1542
8.9	*0142	*0165	-3.5027	*0164	-1.7427	*0147	*0519	*0140	*1553
9.0	*0106	*0123	-3.3397	*0123	-0.0485	*0110	*0486	*0104	*1591
9.1	*0075	*0087	-3.4626	*0086	-1.8499	*0077	*0485	*0073	*1614
9.2	*0049	*0056	-3.5612	*0056	-2.7560	*0056	*0472	*0048	*1540
9.3	*0028	*0032	-3.0212	*0032	-2.4403	*0029	*0388	*0027	*1317
9.4	*0013	*0015	-1.7192	*0014	1.5007	*0013	*0222	*0112	*0849
9.5	*0003	*0004	-1.1863	*0004	0.7268	*0003	*0033	*0003	*0504
9.6	*0.0000	*0.0000	*0.6666	*0.0000	*0.2299	*0.0000	*0.0228	*0.0000	*0.076
9.7	*0.0003	*0.0004	*-5163	*0.0004	*-2.7975	*0.0003	*0.1444	*0.0003	*0.030
9.8	*0.0012	*0.0015	*-2.6946	*0.0015	*-6.7113	*0.0013	*0.1513	*0.0112	*0.0754
9.9	*0.0027	*0.0033	*-4.7477	*0.0033	*-8.8324	*0.0029	*0.0869	*0.0028	*0.1090
10.0	*0.0048	*0.0058	*-5.8699	*0.0058	*-1.2241	*0.0051	*0.1081	*0.0049	*0.1295
10.1	*0.0074	*0.0089	*-6.1156	*0.0090	*-2.3637	*0.0078	*0.1151	*0.0076	*1382
10.2	*0.0105	*0.0127	*-6.1800	*0.0128	*-4.1376	*0.0111	*0.1184	*0.0108	*1412
10.3	*0.0142	*0.0171	*-6.4569	*0.0171	*-5.5968	*0.0150	*0.1242	*0.0144	*1429
10.4	*0.0183	*0.0220	*-6.6293	*0.0221	*-6.0056	*0.0193	*0.1283	*0.0186	*1436
10.5	*0.0229	*0.0275	*-6.3905	*0.0277	*-5.7458	*0.0242	*0.1263	*0.0233	*1420
10.6	*0.0280	*0.0337	*-5.8672	*0.0339	*-5.6281	*0.0296	*0.1206	*0.0285	*1387
10.7	*0.0336	*0.0404	*-5.3306	*0.0406	*-5.8660	*0.0355	*0.1154	*0.0342	*1351
10.8	*0.0397	*0.0478	*-5.0064	*0.0480	*-5.3126	*0.0419	*0.1123	*0.0405	*1319
10.9	*0.0462	*0.0557	*-4.8466	*0.0560	*-6.7808	*0.0489	*0.1102	*0.0472	*1293
11.0	*0.0533	*0.0642	*-4.8099	*0.0646	*-7.1860	*0.0564	*0.1111	*0.0544	*1273
11.1	*0.0608	*0.0733	*-4.8938	*0.0737	*-7.5935	*0.0644	*0.1121	*0.0621	*1259
11.2	*0.0699	*0.0830	*-5.1069	*0.0834	*-8.1542	*0.0728	*0.1142	*0.0703	*1260
11.3	*0.0774	*0.0932	*-5.2199	*0.0937	*-8.6874	*0.0818	*0.1186	*0.0789	*1298
11.4	*0.0863	*1040	*-5.0457	*1045	*-8.8902	*0.0913	*1203	*0.0880	*1314
11.5	*0.0957	*1153	*-4.7993	*1159	*-8.9242	*0.112	*1171	*0.0976	*1257
11.6	*1055	*1271	*-4.6563	*1278	*-9.0467	*1116	*1141	*1077	*1198
11.7	*1158	*1395	*-4.5874	*1402	*-9.2338	*1224	*1181	*1169	
11.8	*1264	*1524	*-4.5443	*1531	*-9.4244	*1337	*1120	*1290	*1553
11.9	*1375	*1657	*-4.5032	*1665	*-9.5943	*1455	*1131	*1403	*1149
12.0	*1490	*1795	*-4.4525	*1804	*-9.7379	*1576	*1134	*1520	*1142
12.1	*1609	*1918	*-4.4019	*1948	*-9.8681	*1701	*1134	*1641	*1127

Figure A.2- Continued

173	*1751	*<0.996	-4.1789	-9.9930	*1153	*1166
172	*173	-4.1241	.2248	-10.1121	*1964	*1694
174	*1986	*2393	-4.3021	*2405	-10.2126	*2026
175	*2118	*2553	-4.2842	*2565	-10.3195	*2100
176	*2224	*2716	-4.2865	*2730	-10.4349	*2284
177	*2393	*2883	-4.3048	*2897	-10.5554	*2300
178	*2554	*3053	-4.3331	*3069	-10.6749	*2441
179	*2678	*3227	-4.1655	*3243	-10.7876	*2585
180	*2855	*3604	-4.4000	*3621	-10.8919	*2832
181	*3124	*3913	-4.403	*3938	-10.9813	*2987
182	*4067	*4909	-4.4843	*4925	-11.0694	*3145
183	*4228	*5095	-4.5544	*5120	-11.1784	*3305
184	*4390	*5248	-4.136	*4136	*11.3323	*3467
185	*4553	*5386	-4.325	*4325	-11.8839	*3630
186	*3147	*4515	-5.2157	*4537	-11.9043	*3796
187	*3916	*4707	-5.8147	*4730	-12.4596	*4131
188	*4067	*4907	-5.6585	*4925	-11.7958	*4301
189	*4228	*5095	-15.5885	*5120	-27.5664	*4472
190	*4390	*5248	-1.4607	*5273	-9.6817	*4631
191	*4553	*5386	-17.3301	*5210	-11.5598	*4798
192	*4716	*5524	16.5106	*5548	16.4700	*4943
193	*4879	*5663	15.6672	*5685	15.6641	*5100
194	*5043	*5801	14.3876	*5812	14.3404	*5256
195	*5206	*5939	13.4548	*5917	13.1640	*5129
196	*5369	*6077	12.3582	*6098	12.0312	*5568
197	*5522	*6215	11.3377	*6235	10.9847	*6232
198	*5684	*6352	10.4095	*6375	10.0417	*5879
199	*5855	*6489	9.5781	*6507	9.2059	*6033
200	*6015	*6625	8.8578	*6643	8.4769	*6187
201	*6174	*6759	8.2158	*6777	7.8536	*6339
202	*6332	*6893	7.6343	*6910	7.2955	*6490
203	*6488	*7026	7.0962	*7041	6.7854	*6640
204	*6643	*7157	6.5956	*7172	6.3182	*6789
205	*6796	*7286	6.1374	*7300	5.8992	*6934
206	*6947	*7414	5.7353	*7428	5.5417	*7078
207	*7096	*7540	5.4001	*7553	5.2557	*7221
208	*7242	*7664	5.1279	*7677	5.0355	*7361
209	*7387	*7786	4.9100	*7798	4.8708	*7499
210	*7548	*7906	4.7349	*7917	4.7580	*7635
211	*7697	*8024	4.5996	*8035	4.6899	*7768
212	*7803	*8139	4.4888	*8149	4.6489	*7898
213	*7936	*8252	4.3714	*8261	4.6009	*8025
214	*8066	*8362	4.2139	*8371	4.5114	*8150
215	*8193	*8470	4.0082	*8478	4.3727	*8271
216	*8316	*8574	3.7682	*8582	4.1991	*8389
217	*8436	*8675	3.5065	*8682	4.0024	*8504
218	*8552	*8774	3.2289	*8780	3.7856	*8615
219	*8665	*8869	2.9227	*8876	3.5380	*8722
220	*8773	*8961	2.6581	*8966	3.2238	*8826
221	*8878	*9049	2.3375	*9054	2.8520	*8926
222	*8978	*9134	1.7682	*9139	2.5354	*9022
223	*9074	*9216	1.5812	*9220	2.4058	*9114
224	*9166	*9294	1.5995	*9298	2.4788	*9202
225	*9281	*9730	1.4157	*9731	4.0564	*9638
226	*9337	*9776	1.9702	*9441	3.0096	*9695
227	*9415	*9505	2.2206	*9507	3.3759	*9460
228	*9489	*9818	1.9768	*9819	3.8828	*9794
229	*9558	*9855	1.6741	*9856	3.7336	*9836
230	*9622	*9888	1.3148	*9889	3.5268	*9874
231	*9681	*9917	1.899	*9918	3.2608	*9907
232	*9735	*9942	1.3520	*9942	2.8816	*9934

Figure A.2- Continued

148	.9955	.9962	-.4223	.9962	2.3418	.9957	.0160	.9956	.0261
149	.9973	.9978	-1.4918	.9978	1.1609	.9975	.0313	.9974	.0283
150	.9987	.9989	-7.6871	.9989	-.9395	.9987	.0648	.9987	.0275
191	.9995	.9996	-24.6958	.9996	-1.9799	.9995	.1357	.9995	-.0299

Figure A.2- Continued

Figure A.2- Continued

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*****  
* OUTPUT FOR CASE NO. 1 OF 19 *  
*****  
  
.....MACH NUMBER,  
VALUES OF PERTURBATION PARAMETERS,  
CRITICAL VALUE OF CP:  
  
M2 = .7750  
  
Q2(1) = .7750 (MACH NO.)  
Q2(2) = .1200 ( TAU )  
Q2(3) = 3.0000 ( PITCH )  
Q2(4) = .5000 ( ALPHA1 )  
  
CPCRIT = -.5095  
  
.....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.  
(* DENOTES POINT ON LOWER SURFACE)  
  
PERTURBATION SOLN:  
  
MINIMUM AT X = .4771* (POINT NO. 52)  
MAXIMUM AT X = 0.0000 (POINT NO. 96)  
2 CRITICAL POINT(S):  
1ST AT X = .5420* (AFTER POINT NO. 48)  
2ND AT X = .5065 (AFTER POINT NO. 139)  
  
COMPARISON SOLN:  
  
MINIMUM AT X = .4870* (POINT NO. 49)  
MAXIMUM AT X = 0.0000 (POINT NO. 96)  
2 CRITICAL POINT(S):  
1ST AT X = .5493* (AFTER POINT NO. 45)  
2ND AT X = .5086 (AFTER POINT NO. 144)  
  
.....FINAL PRINTOUT AND GRAPHICAL DISPLAY OF CP  
  
H = MAXIMUM VALUE OF CP = 1.1517  
L = MINIMUM VALUE OF CP = -1.0527  
* = CRITICAL VALUE OF CP = -.5095  
P = VALUE OF CP PREDICTED BY PERTURBATION SOLUTION  
C = VALUE OF CP IN COMPARISON SOLUTION  
S = AGREEMENT BETWEEN P AND C  
  
PT XBASE CPBASE XPERT CPPERT XCHEK CPCHEK CPPINT H-----*-----L  
1 .9995 .6608 .9995 .8210 .9995 .7155 .8108 P C *  
2 .9987 .5751 .9988 .6594 .9987 .6231 .6506 PC *  
3 .9974 .5086 .9976 .5706 .9974 .5555 .5626 PC *  
4 .9956 .4543 .9960 .5145 .9956 .4972 .5050 PC *  
5 .9932 .4077 .9938 .4637 .9932 .4469 .4533 S *  
6 .9903 .3659 .9912 .4197 .9903 .4022 .4081 PC *
```

*9869 *3274 *9861 *3792 *9869 *3619 *3667
 *9830 *2914 *9405 *3417 *9830 *3238 *3281
 *9786 *2575 *9805 *3063 *9786 *2879 *2918
 *9737 *2254 *9760 *2729 *9736 *2538 *2576
 *9683 *1952 *9711 *2412 *9682 *2216 *2254
 *9623 *1671 *9598 *1847 *9558 *1649 *1694
 *9559 *1417 *9535 *1606 *9489 *1419 *1464
 *9490 *1202 *9022 *9069 *9000 *8973 *8900
 *9416 *1031 *9668 *1400 *9415 *1235 *1278
 *9338 *0897 *9397 *1235 *9336 *1093 *1131
 *9255 *0790 *9321 *1104 *9253 *0982 *1012
 *9147 *0700 *9241 *0996 *9165 *0889 *0910
 *9075 *0615 *9157 *0900 *9073 *0800 *0814
 *8978 *0522 *9024 *0810 *8976 *0703 *0715
 *8877 *0413 *8977 *0716 *8876 *0591 *0605
 *8773 *0292 *AB82 *0612 *8771 *0467 *0483
 *8664 *0163 *8783 *0497 *8662 *0335 *0353
 *8551 *0030 *8680 *0374 *8549 *0199 *0218
 *8435 *-0107 *8574 *0247 *8432 *0061 *0082
 *8314 *-0245 *8464 *0118 *8312 *-0076 *-0054
 *8191 *-0382 *8352 *-0011 *8188 *-0211 *-0188
 *8063 *-0517 *8236 *-0136 *8061 *-0341 *-0317
 *7933 *-0645 *8117 *-0262 *7931 *-0463 *-0439
 *7800 *-0766 *7995 *-0381 *7797 *-0574 *-0553
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 *7524 *-0990 *7744 *-0597 *7521 *-0768 *-0754
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 *6789 *-1467 *7075 *-0989 *6787 *-1067 *-1075
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 *6324 *-1666 *6651 *-1091 *6323 *-1020 *-1033
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 *6007 *-1704 *6362 *-1052 *6006 *-0774 *-0784
 *5846 *-1676 *6216 *-0981 *5845 *-0534 *-0555
 *5685 *-1601 *6069 *-0860 *5684 *-0106 *-1007
 *5523 *-1461 *5921 *-0683 *5522 *-4144 *-3249
 *5360 *-1225 *5773 *-0435 *5359 *-9481 *-6201
 *5197 *-0800 *5624 *-1390 *5196 *-1.0476 *-9160
 *5034 *-4036 *5476 *-4091 *5033 *-1.0523 *-1.0445
 *4870 *-8588 *5303 *-7216 *4870 *-1.0527 *-1.0451
 *4707 *-9641 *5126 *-1.0444 *4706 *-1.0523 *-1.0454
 *4544 *-9721 *4948 *-1.0447 *4543 *-1.0509 *-1.0446
 *4381 *-9743 *4771 *-1.0456 *4381 *-1.0488 *-1.0428
 *4220 *-9750 *4595 *-1.0451 *4219 *-1.0459 *-1.0402
 *4058 *-9746 *4419 *-1.0434 *4058 *-1.0422 *-1.0368
 *3898 *-9731 *4245 *-1.0407 *3897 *-1.0379 *-1.0327
 *3739 *-9707 *4072 *-1.0372 *3738 *-1.0328 *-1.0277
 *3582 *-9672 *3900 *-1.0328 *3581 *-1.0268 *-1.0219
 *3425 *-9628 *3730 *-1.0275 *3424 *-1.0200 *-1.0153
 *3271 *-9574 *3562 *-1.0212 *3270 *-1.0122 *-1.0078
 *3118 *-9510 *3395 *-1.0140 *3117 *-1.0036 *-0.9944
 *2968 *-9437 *3231 *-1.0059 *2966 *-0.9941 *-0.9901
 *2819 *-9354 *3070 *-0.9967 *2817 *-0.9837 *-0.9799
 *2673 *-9263 *2911 *-0.9866 *2671 *-0.9723 *-0.9688
 *2529 *-9163 *2754 *-0.9754 *2527 *-0.9601 *-0.9568
 *2388 *-9054 *2601 *-0.9632 *2386 *-0.9468 *-0.9438
 *2250 *-8935 *2450 *-0.9500 *2247 *-0.9325 *-0.9297
 *2115 *-8806 *2303 *-0.9357 *2112 *-0.9172 *-0.9145
 *1983 *-8667 *2159 *-0.9202 *1979 *-0.9008 *-0.8981
 *1854 *-8516 *2019 *-0.9034 *1850 *-0.8831 *-0.8804
 *1729 *-8354 *1882 *-0.8655 *1750 *-0.8439 *-0.8409
 *1607 *-8179 *1750 *-0.8655 *1602 *-0.8439 *-0.8409
 *1489 *-7999 *1750 *-0.8655 *1621 *-0.8443 *-0.8422

Figure A.2- Continued

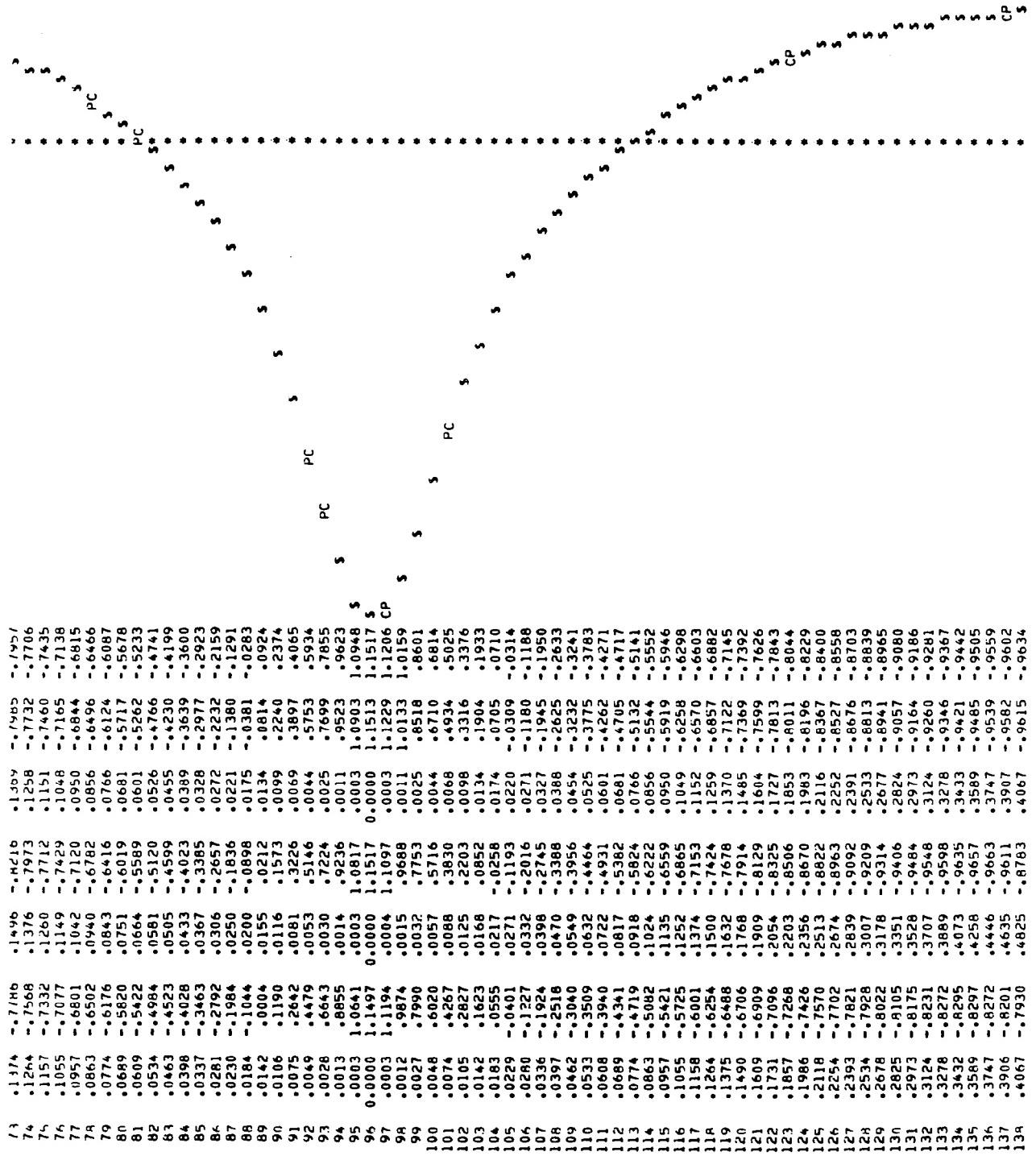


Figure A.2- Continued

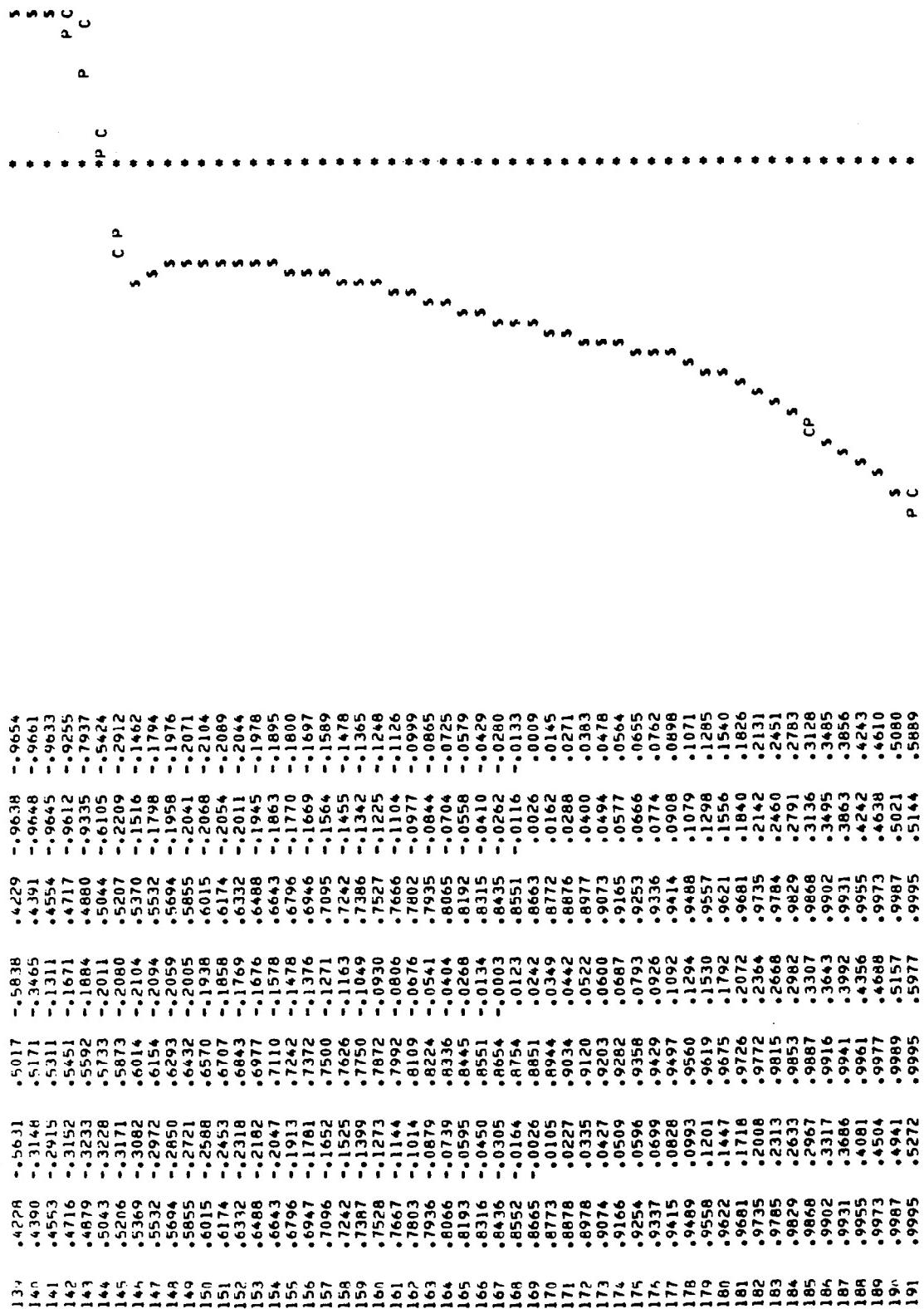


Figure A.2- Continued

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***** OUTPUT FOR CASE NO. 19 OF 19 ****
***** MACH NUMBER.
VALUES OF PERTURBATION PARAMETERS.
CRITICAL VALUE OF CP:
M2 = .7900
Q2(1) = .7900 (MACH NO.)
Q2(2) = .1200 ( TAU )
Q2(3) = 3.0000 ( PITCH )
Q2(4) = .2000 ( ALPHA1 )
CPCRIT = -.4638

***** LOCATIONS OF MIN., MAX., AND CRITICAL PTS.
(* DENOTES POINT ON LOWER SURFACE)

PERTURBATION SOLN:
MINIMUM AT X = .6154* (POINT NO. 51)
MAXIMUM AT X = .0000* (POINT NO. 96)
2 CRITICAL POINT(S):
1ST AT X = .6726* (AFTER POINT NO. 48)
2ND AT X = .6186 (AFTER POINT NO. 138)

COMPARISON SOLN:
MINIMUM AT X = .6939* (POINT NO. 36)
MAXIMUM AT X = 0.0000 (POINT NO. 96)
2 CRITICAL POINT(S):
1ST AT X = .7230* (AFTER POINT NO. 34)
2ND AT X = .6903 (AFTER POINT NO. 155)

***** FINAL PRINTOUT AND GRAPHICAL DISPLAY OF CP
H = MAXIMUM VALUE OF CP = 1.1565
L = MINIMUM VALUE OF CP = -1.1128
* = CRITICAL VALUE OF CP = -.4638
P = VALUE OF CP PREDICTED BY PERTURBATION SOLUTION
C = VALUE OF CP IN COMPARISON SOLUTION
$ = AGREEMENT BETWEEN P AND C
-----L
PT XBASE CPBASE XPERT CPPERT XCHEK CPCHEK CPPINT H-----*
1 .9995 .6608 .9997 .7696 .9995 .7942 .7631 CP $ *
2 .9987 .5751 .9992 .7513 .9987 .7350 .7120 CP * *
3 .9974 .5086 .9983 .6791 .9974 .6519 .6249 CP $ *
4 .9956 .4543 .9971 .6098 .9956 .5805 .5545 CP * *
5 .9932 .4077 .9956 .5560 .9932 .5367 .4993 CP * *
6 .9903 .3659 .9938 .5100 .9903 .5007 .4500 CP C P * *

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Figure A.2- Continued

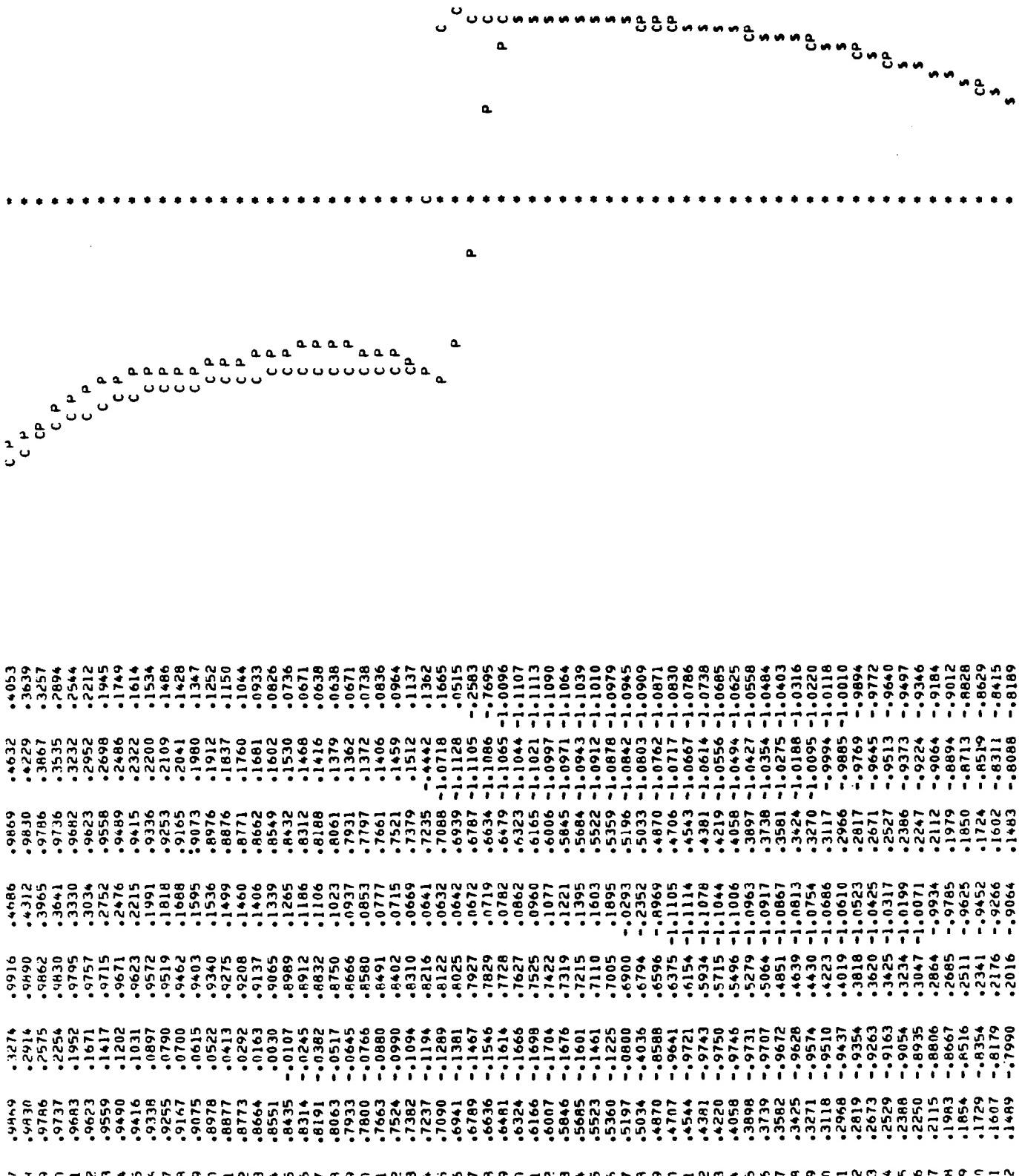


Figure A.2- Continued

14	-0.77446	-0.8845	-0.7849	-0.7446
15	-0.7564	-0.86048	-0.7594	-0.7691
15	-0.7332	-0.8354	-0.7320	-0.7417
15	-0.7055	-0.8081	-0.7025	-0.7125
15	-0.7077	-0.7849	-0.7048	-0.7120
15	-0.6867	-0.6801	-0.6856	-0.6812
15	-0.6867	-0.6801	-0.6705	-0.6812
15	-0.6502	-0.6169	-0.6468	-0.6480
15	-0.6176	-0.1048	-0.7125	-0.5995
15	-0.3463	-0.0456	-0.4206	-0.3228
15	-0.5820	-0.934	-0.6760	-0.5995
15	-0.6669	-0.5422	-0.0825	-0.5683
15	-0.0281	-0.2792	-0.380	-0.3551
15	-0.0230	-0.1984	-0.0311	-0.2506
15	-0.0574	-0.4984	-0.0249	-0.5918
15	-0.0463	-0.4523	-0.0628	-0.5379
15	-0.0398	-0.4028	-0.0538	-0.402
15	-0.0317	-0.3463	-0.0456	-0.4206
15	-0.0689	-0.5820	-0.934	-0.6760
15	-0.0689	-0.5422	-0.0825	-0.6369
15	-0.0230	-0.1984	-0.0311	-0.2506
15	-0.0184	-0.1044	-0.0249	-0.1905
15	-0.0142	-0.004	-0.0193	-0.0779
15	-0.0106	-0.1190	-0.0144	-0.0594
15	0.0003	1.1497	0.0004	0.0594
15	0.0003	1.1194	0.0004	0.1565
15	0.0049	0.4479	0.0264	0.2223
15	0.0049	0.6643	0.0038	0.4150
15	0.0028	0.8855	0.0017	0.8694
15	0.0013	1.0641	0.0004	1.0538
15	0.0003	1.1497	0.0000	0.0000
15	0.0003	1.1194	0.0003	0.0000
15	0.0012	0.9874	0.0018	0.9510
15	0.0027	0.7990	0.0040	0.7368
15	0.0048	0.6020	0.0070	0.5224
15	0.0074	0.4267	0.0108	0.3327
15	0.0105	0.2827	0.0154	0.1700
15	0.0142	0.1623	0.0207	0.0312
15	0.0183	0.0555	0.0267	0.0821
15	0.0229	0.0401	0.0335	0.1726
15	0.0280	-0.1227	0.0409	-0.2679
15	0.0336	-0.1924	0.0491	-0.3139
15	0.0397	-0.2518	0.0580	-0.3143
15	0.0463	-0.3040	0.0677	-0.4295
15	0.0533	-0.3509	0.0780	-0.4803
15	0.0608	-0.3940	0.0891	-0.5287
15	0.0689	-0.4340	0.1008	-0.5770
15	0.0774	-0.4719	0.1132	-0.6217
15	0.0863	-0.5082	0.1263	-0.6585
15	0.0957	-0.5421	0.1400	-0.6902
15	0.1055	-0.5725	0.1544	-0.7204
15	0.1158	-0.6001	0.1694	-0.7490
15	0.1264	-0.6254	0.1851	-0.7759
15	0.1375	-0.6488	0.2013	-0.8009
15	0.1490	-0.6706	0.2181	-0.8239
15	0.1609	-0.6909	0.2354	-0.8451
15	0.1731	-0.7096	0.2533	-0.8647
15	0.1857	-0.7266	0.2717	-0.8828
15	0.1966	-0.7426	0.2906	-0.8996
15	0.2118	-0.7570	0.3100	-0.9151
15	0.2254	-0.7702	0.3299	-0.9296
15	0.2393	-0.7821	0.3502	-0.9432
15	0.2534	-0.7928	0.3709	-0.9556
15	0.2678	-0.8022	0.3919	-0.9669
15	0.2825	-0.8105	0.4134	-0.9767
15	0.2973	-0.8175	0.4352	-0.9855
15	0.3124	-0.8231	0.4573	-0.9928
15	0.3277	-0.8272	0.4797	-0.9991
15	0.3432	-0.8295	0.5023	-1.0047
15	0.3569	-0.8297	0.5253	-1.0099
15	0.3747	-0.8272	0.5484	-1.0151
15	0.3906	-0.8201	0.5717	-1.0207
15	0.4067	-0.7930	0.5952	-0.9748
15	-0.77446	-0.8845	-0.7849	-0.7446
15	-0.7564	-0.86048	-0.7594	-0.7691
15	-0.7332	-0.8354	-0.7320	-0.7417
15	-0.7055	-0.8081	-0.7025	-0.7125
15	-0.7077	-0.7849	-0.7048	-0.7120
15	-0.6867	-0.6801	-0.6856	-0.6812
15	-0.6867	-0.6801	-0.6705	-0.6812
15	-0.6502	-0.6169	-0.6468	-0.6480
15	-0.6176	-0.1048	-0.7125	-0.5995
15	-0.3463	-0.0456	-0.4206	-0.3228
15	-0.5820	-0.934	-0.6760	-0.5995
15	-0.6669	-0.5422	-0.0825	-0.5683
15	-0.0281	-0.2792	-0.380	-0.3551
15	-0.0230	-0.1984	-0.0311	-0.2506
15	-0.0574	-0.4984	-0.0249	-0.4657
15	-0.0463	-0.4523	-0.0628	-0.4196
15	-0.0398	-0.4028	-0.0538	-0.3526
15	-0.0317	-0.3463	-0.0456	-0.3245
15	-0.0689	-0.5820	-0.934	-0.2988
15	-0.0689	-0.5422	-0.0825	-0.2241
15	-0.0230	-0.1984	-0.0311	-0.1345
15	-0.0184	-0.1044	-0.0249	-0.0281
15	-0.0142	-0.004	-0.0193	-0.0175
15	-0.0106	-0.1190	-0.0144	-0.0175
15	0.0003	1.1497	0.0004	0.0003
15	0.0003	1.1194	0.0004	0.0003
15	0.0049	0.4479	0.0264	0.0134
15	0.0049	0.6643	0.0038	0.0044
15	0.0028	0.8855	0.0017	0.0044
15	0.0013	1.0641	0.0004	0.0003
15	0.0003	1.1497	0.0000	0.0000
15	0.0003	1.1194	0.0003	0.0000
15	0.0012	0.9874	0.0018	0.0011
15	0.0027	0.7990	0.0040	0.0011
15	0.0048	0.6020	0.0070	0.0011
15	0.0074	0.4267	0.0108	0.0011
15	0.0105	0.2827	0.0154	0.0011
15	0.0142	0.1623	0.0207	0.0012
15	0.0183	0.0555	0.0267	0.0017
15	0.0229	0.0401	0.0335	0.0021
15	0.0280	-0.1227	0.0409	0.0021
15	0.0336	-0.1924	0.0491	0.0027
15	0.0397	-0.2518	0.0580	0.0032
15	0.0463	-0.3040	0.0677	0.0037
15	0.0533	-0.3509	0.0780	0.0042
15	0.0608	-0.3940	0.0891	0.0047
15	0.0689	-0.4340	0.1008	0.0051
15	0.0774	-0.4719	0.1132	0.0056
15	0.0863	-0.5082	0.1263	0.0061
15	0.0957	-0.5421	0.1400	0.0066
15	0.1055	-0.5725	0.1544	0.0070
15	0.1158	-0.6001	0.1694	0.0074
15	0.1264	-0.6254	0.1851	0.0078
15	0.1375	-0.6488	0.2013	0.0082
15	0.1490	-0.6706	0.2181	0.0086
15	0.1609	-0.6909	0.2354	0.0090
15	0.1731	-0.7096	0.2533	0.0094
15	0.1857	-0.7266	0.2717	0.0098
15	0.1966	-0.7426	0.2906	0.0102
15	0.2118	-0.7570	0.3100	0.0106
15	0.2254	-0.7702	0.3299	0.0110
15	0.2393	-0.7821	0.3502	0.0114
15	0.2534	-0.7928	0.3709	0.0118
15	0.2678	-0.8022	0.3919	0.0122
15	0.2825	-0.8105	0.4134	0.0126
15	0.2973	-0.8175	0.4352	0.0130
15	0.3124	-0.8231	0.4573	0.0134
15	0.3277	-0.8272	0.4797	0.0138
15	0.3432	-0.8295	0.5023	0.0142
15	0.3569	-0.8297	0.5253	0.0146
15	0.3747	-0.8272	0.5484	0.0150
15	0.3906	-0.8201	0.5717	0.0154
15	0.4067	-0.7930	0.5952	0.0158
15	-0.77446	-0.8845	-0.7849	-0.7446
15	-0.7564	-0.86048	-0.7594	-0.7691
15	-0.7332	-0.8354	-0.7320	-0.7417
15	-0.7055	-0.8081	-0.7025	-0.7125
15	-0.7077	-0.7849	-0.7048	-0.7120
15	-0.6867	-0.6801	-0.6856	-0.6812
15	-0.6867	-0.6801	-0.6705	-0.6812
15	-0.6502	-0.6169	-0.6468	-0.6480
15	-0.6176	-0.1048	-0.7125	-0.5995
15	-0.3463	-0.0456	-0.4206	-0.3228
15	-0.5820	-0.934	-0.6760	-0.5995
15	-0.6669	-0.5422	-0.0825	-0.5683
15	-0.0281	-0.2792	-0.380	-0.3551
15	-0.0230	-0.1984	-0.0311	-0.2506
15	-0.0574	-0.4984	-0.0249	-0.4657
15	-0.0463	-0.4523	-0.0628	-0.4196
15	-0.0398	-0.4028	-0.0538	-0.3526
15	-0.0317	-0.3463	-0.0456	-0.3245
15	-0.0689	-0.5820	-0.934	-0.2988
15	-0.0689	-0.5422	-0.0825	-0.2241
15	-0.0230	-0.1984	-0.0311	-0.1345
15	-0.0184	-0.1044	-0.0249	-0.0281
15	-0.0142	-0.004	-0.0193	-0.0175
15	-0.0106	-0.1190	-0.0144	-0.0175
15	0.0003	1.1497	0.0004	0.0003
15	0.0003	1.1194	0.0004	0.0003
15	0.0049	0.4479	0.0264	0.0134
15	0.0049	0.6643	0.0038	0.0044
15	0.0028	0.8855	0.0017	0.0044
15	0.0013	1.0641	0.0004	0.0003
15	0.0003	1.1497	0.0000	0.0000
15	0.0003	1.1194	0.0003	0.0000
15	0.0012	0.9874	0.0018	0.0011
15	0.0027	0.7990	0.0040	0.0011
15	0.0048	0.6020	0.0070	0.0011
15	0.0074	0.4267	0.0108	0.0011
15	0.0105	0.2827	0.0154	0.0011
15	0.0142	0.1623	0.0207	0.0012
15	0.0183	0.0555	0.0267	0.0014
15	0.0229	0.0401	0.0335	0.0017
15	0.0280	-0.1227	0.0409	0.0021
15	0.0336	-0.1924	0.0491	0.0027
15	0.0397	-0.2518	0.0580	0.0032
15	0.0463	-0.3040	0.0677	0.0037
15	0.0533	-0.3509	0.0780	0.0042
15	0.0608	-0.3940	0.0891	0.0047
15	0.0689	-0.4340	0.1008	0.0051
15	0.0774	-0.4719	0.1132	0.0056
15	0.0863	-0.5082	0.1263	0.0061
15	0.0957	-0.5421	0.1400	0.0066
15	0.1055	-0.5725	0.1544	0.0070
15	0.1158	-0.6001	0.1694	0.0074
15	0.1264	-0.6254	0.1851	0.0078
15	0.1375	-0.6488	0.2013	0.0082
15	0.1490	-0.6706	0.2181	0.0086
15	0.1609	-0.6909	0.2354	0.0090
15	0.1731	-0.7096	0.2533	0.0094
15	0.1857	-0.7266	0.2717	0.0098
15	0.1966	-0.7426	0.2906	0.0102
15	0.2118	-0.7570	0.3100	0.0106
15	0.2254	-0.7702	0.3299	0.0110
15	0.2393	-0.7821	0.3502	0.0114
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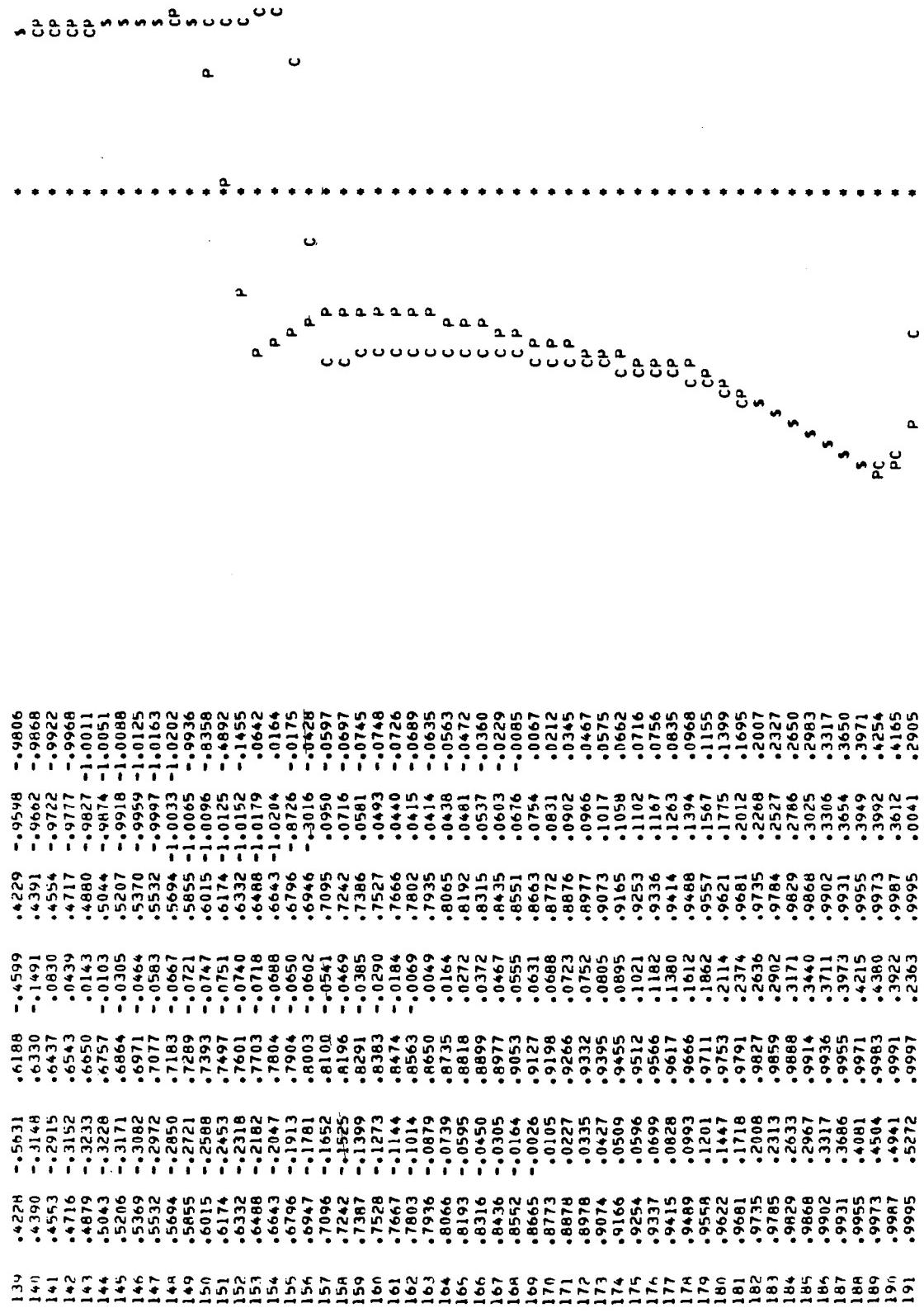


Figure A.2- Continued

* TEST CASE *

	* MACH NO.	* 1ST PAR.	* 2ND PAR.	* 3RD PAR.	* 4TH PAR.	
						* TAU * PITCH * ALPHA1
* BASE SOLUTION:	* .780	* .780	* .110	* .3.200	* .300	
* 1ST CALIBRATION SOLN:	* .790	* .790	* .110	* .3.200	* .300	
* 2ND CALIBRATION SOLN:	* .780	* .780	* .120	* .3.200	* .300	
* 3RD CALIBRATION SOLN:	* .780	* .780	* .110	* .3.000	* .300	
* 4TH CALIBRATION SOLN:	* .780	* .780	* .110	* .3.200	* .500	
* 1ST PERTURBATION SOLN:	* .775	* .775	* .120	* .3.000	* .500	
* 2ND PERTURBATION SOLN:	* .780	* .780	* .115	* .3.000	* .500	
* 3RD PERTURBATION SOLN:	* .780	* .780	* .120	* .3.100	* .500	
* 4TH PERTURBATION SOLN:	* .780	* .780	* .120	* .3.000	* .400	
* 5TH PERTURBATION SOLN:	* .785	* .785	* .110	* .3.000	* .500	
* 6TH PERTURBATION SOLN:	* .785	* .785	* .115	* .3.100	* .500	
* 7TH PERTURBATION SOLN:	* .785	* .785	* .115	* .3.000	* .400	
* 8TH PERTURBATION SOLN:	* .785	* .785	* .120	* .3.200	* .500	
* 9TH PERTURBATION SOLN:	* .785	* .785	* .120	* .3.100	* .400	
* 10TH PERTURBATION SOLN:	* .785	* .785	* .120	* .3.000	* .300	
* 11TH PERTURBATION SOLN:	* .790	* .790	* .105	* .3.000	* .500	
* 12TH PERTURBATION SOLN:	* .790	* .790	* .110	* .3.100	* .500	
* 13TH PERTURBATION SOLN:	* .790	* .790	* .110	* .3.000	* .400	
* 14TH PERTURBATION SOLN:	* .790	* .790	* .120	* .3.300	* .500	
* 15TH PERTURBATION SOLN:	* .790	* .790	* .115	* .3.000	* .300	
* 16TH PERTURBATION SOLN:	* .790	* .790	* .120	* .3.300	* .500	
* 17TH PERTURBATION SOLN:	* .790	* .790	* .115	* .3.100	* .400	
* 18TH PERTURBATION SOLN:	* .790	* .790	* .120	* .3.100	* .300	
* 19TH PERTURBATION SOLN:	* .790	* .790	* .120	* .3.000	* .200	

Figure A.2- Concluded

PLOT of C_p
TEST CASE: CASE NO. 1 OF 19
CALIBRATION NO. 1 OF 4

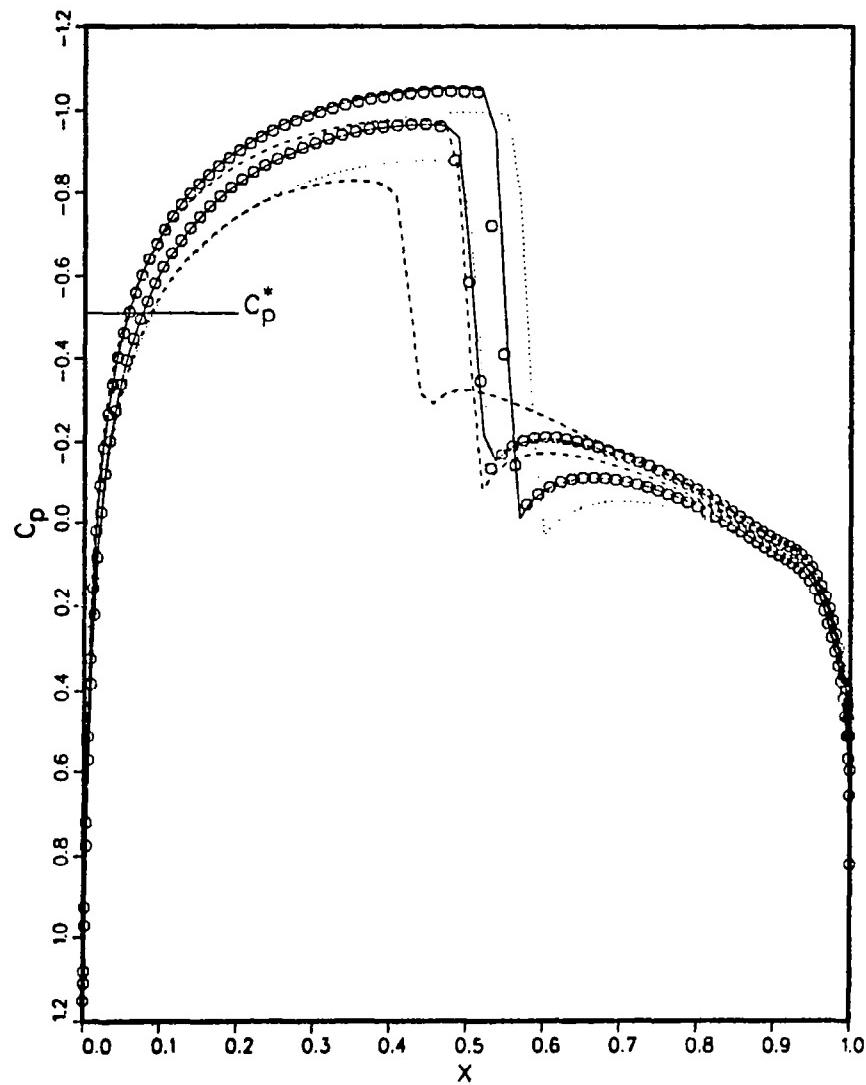


Figure A.3- Abbreviated plot output for sample case

PLOT of C_p
TEST CASE: CASE NO. 1 OF 19
CALIBRATION NO. 2 OF 4

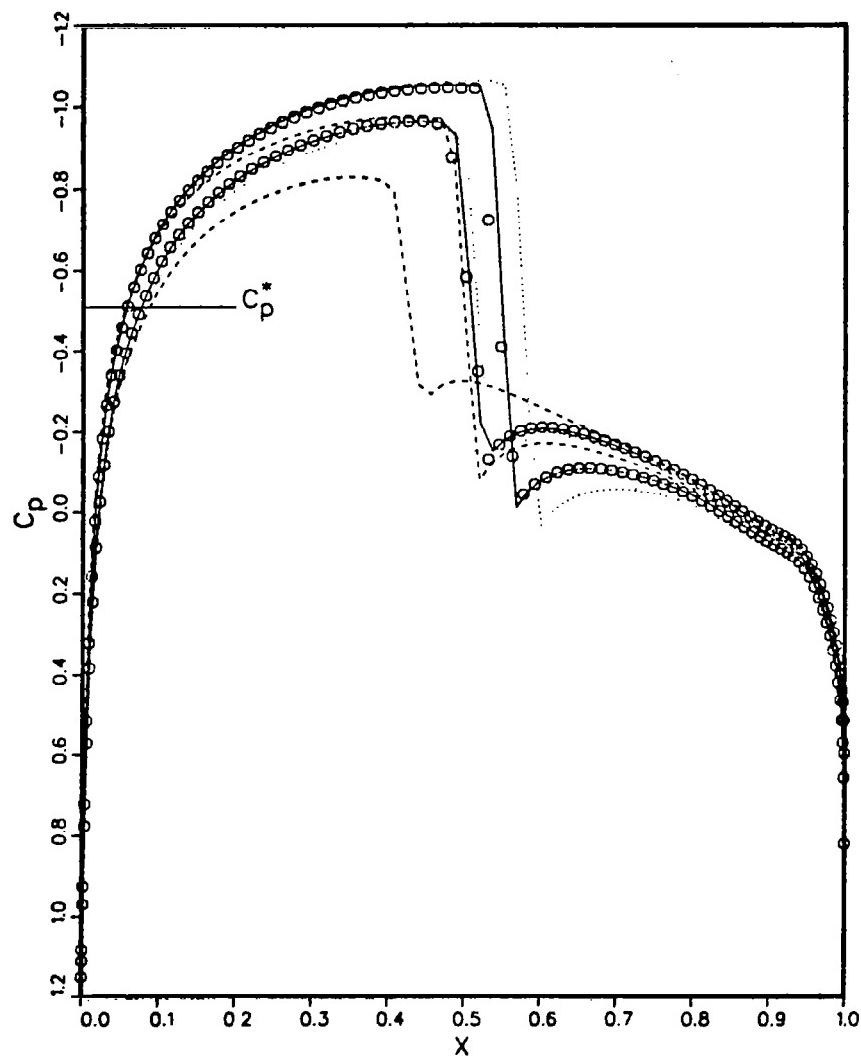


Figure A.3- Continued

PLOT of C_p
TEST CASE: CASE NO. 1 OF 19
CALIBRATION NO. 3 OF 4

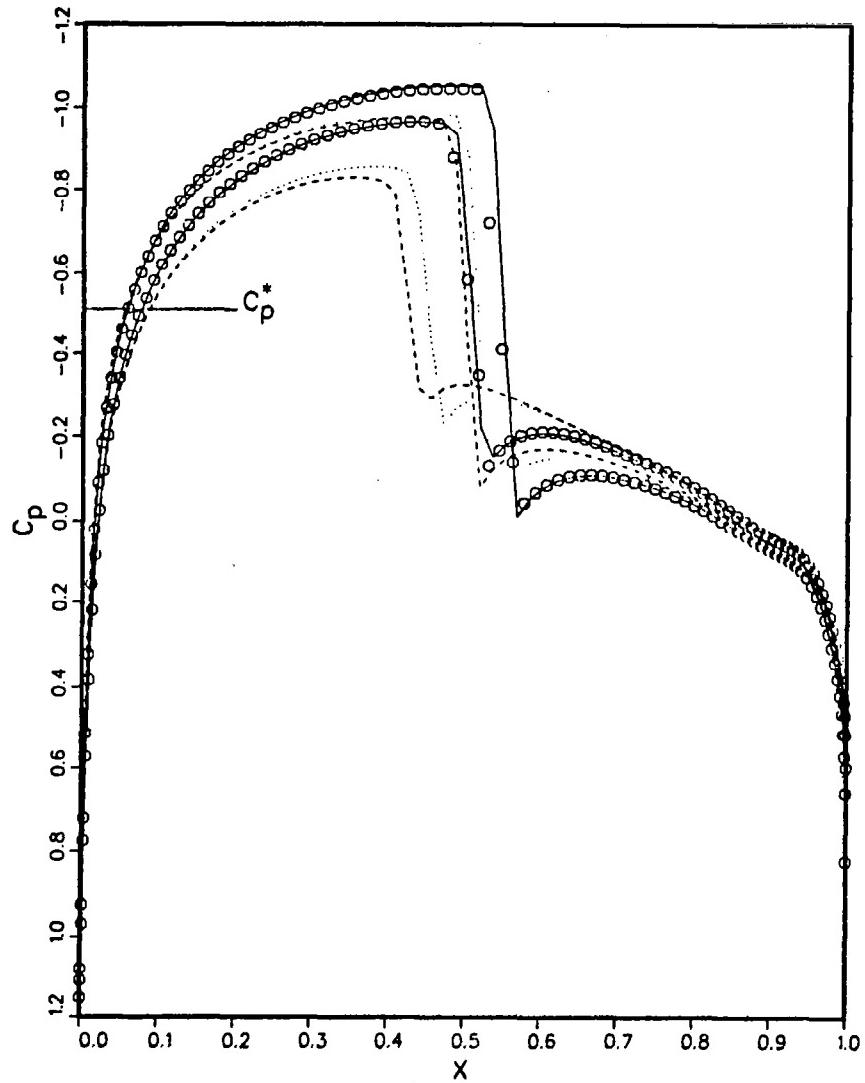


Figure A.3- Continued

PLOT of C_p
TEST CASE: CASE NO. 1 OF 19
CALIBRATION NO. 4 OF 4

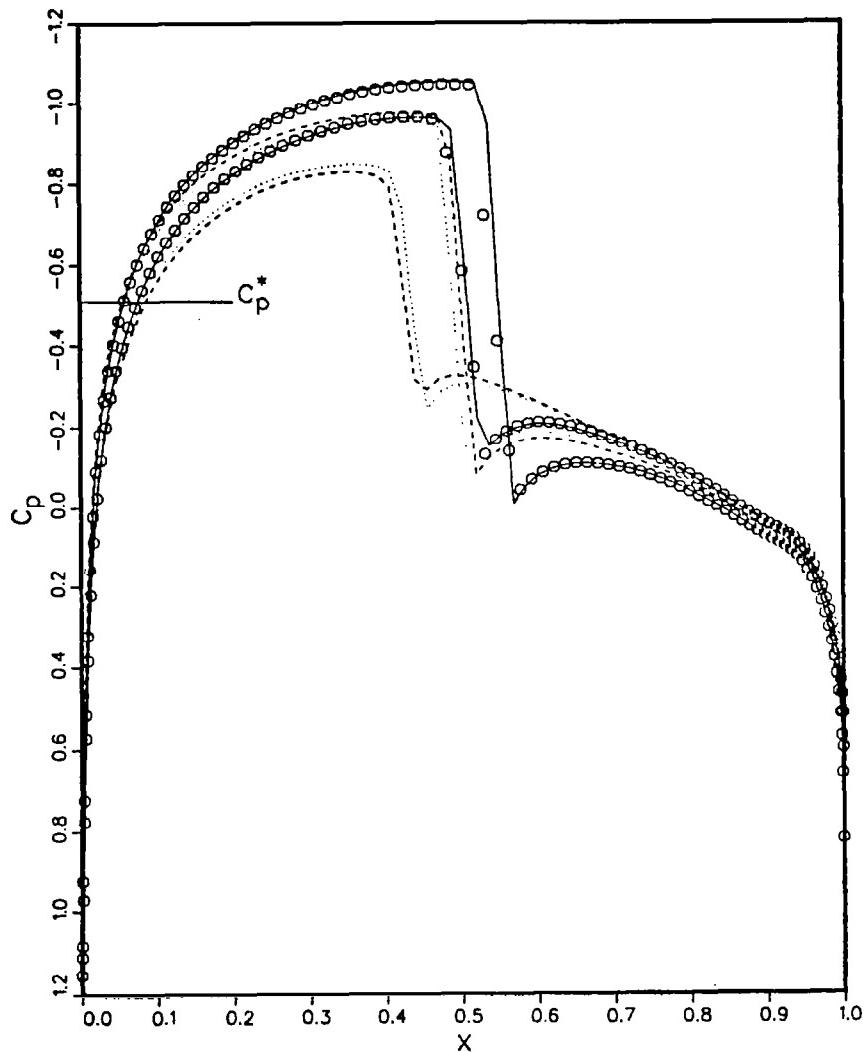


Figure A.3- Concluded

APPENDIX B - LISTING OF COMPUTER PROGRAM PERTRB

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PROGRAM PERTRB (INPUT,OUTPUT,TAPE1,TAPE5=INPUT,TAPE6=OUTPUT,
  & TAPE15)
C***** MAIN PROGRAM PERTURB *****
C
C CALCULATES CONTINUOUS OR DISCONTINUOUS NONLINEAR PERTURBATION
C SOLUTIONS WHICH REPRESENT A MULTIPLE-PARAMETER CHANGE IN EITHER
C GEOMETRY OR FLOW CONDITIONS BY EMPLOYING A STRAINED-COORDINATE
C PROCEDURE. THE METHOD UTILIZES UNIT PERTURBATIONS, DETERMINED
C FOR EACH PARAMETER FROM A PREVIOUSLY CALCULATED COMMON BASE
C SOLUTION, AND A CALIBRATION SOLUTION DISPLACED FROM IT BY SOME
C REASONABLE CHANGE IN THE RELEVANT PARAMETERS, TO PREDICT NEW
C NONLINEAR SOLUTIONS OVER A RANGE OF PARAMETER VARIATION.
C
C THIS CURRENT VERSION OF THE PROCEDURE IS CONFIGURED TO PREDICT AND
C PLOT PRESSURE COEFFICIENTS ON A BLADE OR AIRFOIL SURFACE, AND CAN
C ACCOUNT FOR THE MOTION OF
C   (1) ONE OR MORE CRITICAL POINTS (SHOCK POINTS),
C   (2) A STAGNATION POINT,
C   (3) A MAXIMUM-SUCTION-PRESSURE POINT,
C OR SIMULTANEOUSLY FOR ANY COMBINATION OF THESE.
C
C THE PROGRAM IS ALSO CONFIGURED TO COMPARE THE PERTURBATION-
C PREDICTED SOLUTIONS WITH THE CORRESPONDING EXACT SOLUTIONS
C OBTAINED BY EMPLOYING THE SAME EXPENSIVE COMPUTATIONAL
C PROCEDURE USED TO DETERMINE THE BASE AND CALIBRATION SOLUTIONS.
C SEE THE SUBROUTINE INPUT FOR DETAILS.
C
C N = NO. OF POINTS IN SURFACE PRESSURE DISTRIBUTION - ASSUMED EQUAL
C FOR BASE, CALIBRATION, AND PREDICTED DISTRIBUTIONS.
C   NOTE: N .LE. 200.
C
C NPARAM = NO. OF PARAMETERS VARIED. NOTE: NPARAM .LE. 8.
C
C          BASE           KTH CALIBRATION
C MACH NO.    M0            M1
C PERT. PARAMS. Q0(1),Q0(2),..., Q0(I),...,Q0(K-1),Q1,Q0(K+1),...
C
C M2 = ONCOMING MACH NO. OF PREDICTED FLOW.
C Q2(1),Q2(2),... = VALUES OF PERTURBED PARAMETERS IN PREDICTED
C FLOW.
C
C COORDINATE STRAINING IS PIECEWISE LINEAR WITH END POINTS AND ONE
C OR MORE USER-SELECTED INTERIOR POINTS HELD INVARIANT.
C
C THE PROGRAM LOCATES MINIMUM, MAXIMUM, AND ALL CRITICAL POINTS
C (SHOCK POINTS) IN THE BASE AND CALIBRATION SOLUTIONS, AND STORES
C THESE IN THE ARRAYS XLOC0 AND XLOC1 (IT IS ASSUMED THAT THE NUMBER
C OF CRITICAL POINTS DOES NOT EXCEED FOUR) AS FOLLOWS
C
C          BASE           CALIBRATION
C
C XLOC0(1) = MINIMUM PT.      XLOC1(1) = MINIMUM PT.      MAIN 50
C XLOC0(2) = MAXIMUM PT.      XLOC1(2) = MAXIMUM PT.      MAIN 51
C XLOC0(3) = CRITICAL PT. NO. I XLOC1(3) = CRITICAL PT. NO. I MAIN 52
C ...     = ...
C
C          XLOC0(6) = CRITICAL PT. NO. 4      XLOC1(6) = CRITICAL PT. NO. 4 MAIN 58
C          THE NUMBER OF POINTS SELECTED FROM THESE IS SPECIFIED BY NSELCT. MAIN 59
C          THE CORRESPONDING SUBSCRIPTS OF XLOC0 AND XLOC1 ARE SPECIFIED IN MAIN 60
C          THE FIRST NSELCT ELEMENTS OF THE ARRAY LSELCT, E.G., TO SELECT THE MAIN 61
C          MAXIMUM POINT AND THE FIRST AND THIRD CRITICAL POINTS, ONE MAIN 62
C          SPECIFIES MAIN 63
C
C          NSELCT = 3
C          LSELCT(1) = 2
C          LSELCT(2) = 3
C          LSELCT(3) = 5
C
C          PROVISION IS ALSO MADE TO ALLOW THE USER TO SPECIFY THE POINTS IN MAIN 70
C          THE BASE AND CALIBRATION SOLUTIONS THAT WILL BE HELD INVARIANT, TO MAIN 71
C          A MAXIMUM OF SIX. SEE THE SUBROUTINE INPUT FOR DETAILS. MAIN 72
C
C          DIMENSION XCSAVE(8,200),YCSAVE(8,200),XUSAVE(8,200),YUNIT(8,200) MAIN 73
C          DIMENSION XBASE(200),XCALB(200),XCHEK(200),XPERT(200),XUNIT(200), MAIN 74
C          % DELX(200)
C          % DIMENSION YBASE(200),YCALB(200),YPERT(200),YINTP(200), MAIN 75
C          % YPRTI(200)
C          % DIMENSION XLOC0(6),XLOC1(6),XLOC3(6),XFIX0(8),XFIX1(8) MAIN 76
C          % DIMENSION LCR0(4),LCR1(4),LCR2(4),LCR3(4),LOC0(6),LOC1(6), MAIN 77
C          % TSE0(8),ISE0(8),LSELCT(16)
C          % DIMENSION Q0(8),Q2(8),DEL1(8),XOUT(8),ORD(8),PARNAH(8) MAIN 78
C          % DIMENSION HEAD0(5),HEAD1(5),HEAD2(5),HEAD3(5)
C          % DIMENSION VNAM(21),FLAG(8),STRING(96),STRUN(196)
C          % DIMENSION MSAVE(8),M2SAVE(25),Q1SAVE(8),Q2SAVE(8+25)
C          REAL M0,M1,M2,MSAVE,M2SAVE
C          COMMON /COEFF/C(8,7),D(8,7)
C          COMMON /HEAD/TITLE(80)
C          COMMON /PARAM/ PARNAM,LOC0,LOC1,LSELCT,N,NCASE,LSPEC,UNIT,
C          % LCHKEK,LPLOT,NSELCT,A,B,NPARAM,VNAM
C          COMMON /PERT/ Q0(8),M0,M2,D(8),YCR0,YCR1,YCR2,YMIN,YMAX
C          COMMON /SAVE/ XCSAVE,YCSAVE
C          COMMON /TBL/ HEAD0,HEAD1,HEAD2,MSAVE,M2SAVE,Q1SAVE,Q2SAVE
C          COMMON /XY/ XBASE,XCALB,XPERT,XCHEK,YBASE,YCALB,YPERT,YCHEK,YPRTI
C          DATA LTERM /0/,LCORR /0/
C          DATA HEAD0 /4HBASE,4HSOL,4HMUTIO,4HMMI,4H/ /
C          % HEAD1 /4HCALI,4HBRAT,4HION,4HSOLN,4H/ /
C          % HEAD2 /4HPERT,4HURBA,4HTION,4HSOL,4HN/ /
C          % HEAD3 /4HCOMP,4HARIS,4HMON,4HOLN,4H/ /
C          DATA ORD /4HIST,4H2ND,4H3RD,4H4TH,4H5TH,4H6TH,4H7TH,4H8TH/ /
C
C          ***** USER-SUPPLIED STATEMENT FUNCTION YCRIT(Z) DETERMINES CRITICAL
C          VALUES OF FLOW VARIABLE YCRIT AS FUNCTION OF FLOW PARAMETER Z.
C          IGRAD (+1 OR -1) IS THE USER-SUPPLIED ALGEBRAIC SIGN OF DYCRIT/DX
C          USED IN LOCATING THE CRITICAL POINT.
C
C          IN THE PRESENT VERSION OF THE CODE, YCRIT REPRESENTS THE FULL-
C          POTENTIAL CRITICAL PRESSURE COEFFICIENT FOR AIR (GAMMA = 1.4). Z
  
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C   IS THE FREE STREAM MACH NUMBER, AND IGRAD CORRESPONDS TO POSITIVE MAIN115   C.....NORMALIZE X COORDINATES AND LOCATE MINIMUM, MAXIMUM, AND CRITICAL MAIN172
C   PRESSURE GRADIENT (+1). MAIN116   C POINTS FOR BASE SOLUTION. MAIN173
C   YCRIT(Z)=2.0*((Z+0.4*Z**2)/2.4)**(1.4/0.4)-1.0/(1.4*Z**2) MAIN117   C MAIN174
C   [IGRAD=1] MAIN118   C MAIN175
C***** C.....PRINT BANNER PAGE. MAIN119   C MAIN176
C   CALL BANNER MAIN120   C MAIN177
C***** C.....READ LECHO AND ECHO INPUT DECK IF LECHO .EQ. 1. MAIN121   C MAIN178
C   CALL ECHIMP MAIN122   C MAIN179
C***** C.....INPUT CONTROL, GEOMETRY, AND STRAINING PARAMETERS. MAIN123   C MAIN180
C   CALL INPUT (1) MAIN124   C MAIN181
C***** C.....WRITE TITLE AND INPUT PARAMETERS. MAIN125   C MAIN182
C   WRITE (6,1000) TITLE MAIN126   C MAIN183
C   WRITE (6,1010) N,A,B,NPARAM MAIN127   C MAIN184
C   NFIX=NSELCT*2 MAIN128   C MAIN185
C   NSEQ=NFIX-1 MAIN129   C MAIN186
C***** C.....PRINT INFORMATION REGARDING STRAINING TO BE USED. MAIN130   C MAIN187
C   WRITE (6,1020) NFIX MAIN131   C MAIN188
C   IF (LSPEC .EQ. 0) GO TO 10 MAIN132   C MAIN189
C   WRITE (6,1030) MAIN133   C MAIN190
C   GO TO 30 MAIN134   C MAIN191
10 CONTINUE MAIN135   C MAIN192
      WRITE (6,1040) MAIN136   C MAIN193
      DO 20 I=1,NSELCT MAIN137   C MAIN194
      IF (ISELCT(I) .EQ. 1) WRITE (6,1050) VNAM MAIN138   C MAIN195
      IF (ISELCT(I) .EQ. 2) WRITE (6,1060) VNAM MAIN139   C MAIN196
      IF (ISELCT(I) .LE. 2) GO TO 20 MAIN140   C MAIN197
      LCORR=1 MAIN141   C MAIN198
      LPR=LSELCT(I)-2 MAIN142   C MAIN199
      WRITE (6,1070) VNAM,ORD(LPR) MAIN143   C MAIN200
20 CONTINUE MAIN144   C MAIN201
30 CONTINUE MAIN145   C MAIN202
C***** C.....BEGIN CALCULATIONS ON BASE SOLUTION. MAIN146   C MAIN203
C   CALL INPUT (2) MAIN147   C MAIN204
      YCRD=YCRIT(M0) MAIN148   C MAIN205
      WRITE (6,1080) HEAD0 MAIN149   C MAIN206
      WRITE (6,1090) VNAM MAIN150   C MAIN207
      WRITE (6,1100) M0 MAIN151   C MAIN208
      IF (NPARAM .EQ. 1) WRITE (6,1110) QD(1),PARNAM(1) MAIN152   C MAIN209
      IF (NPARAM .GT. 1) WRITE (6,1120) (K,QD(K),PARNAM(K),K=1,NPARAM) MAIN153   C MAIN210
      WRITE (6,1130) VNAM,YCRD MAIN154   C MAIN211
      MAIN155   C MAIN212
      MAIN156   C MAIN213
      MAIN157   C MAIN214
      MAIN158   C MAIN215
      MAIN159   C MAIN216
      MAIN160   C MAIN217
      MAIN161   C MAIN218
      MAIN162   C MAIN219
      MAIN163   C MAIN220
      MAIN164   C MAIN221
      MAIN165   C MAIN222
      MAIN166   C MAIN223
      MAIN167   C MAIN224
      MAIN168   C MAIN225
      MAIN169   C MAIN226
      MAIN170   C MAIN227
      MAIN171   C MAIN228

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IF (NPARAM .EQ. 1) WRITE (6,1250) Q1,PARNAM()          MAIN229
IF (NPARAM .GT. 1 .AND. KK ,EQ, K) WRITE (6,1260) KK,Q1,PARNAM(KK)
IF (KK ,NE, K) WRITE (6,1270) KK,Q1,PARNAM(KK)        MAIN230
60 CONTINUE
WRITE (6,1130) VNAM,YCRI                           MAIN231
C.....NORMALIZE X COORDINATES AND LOCATE MINIMUM, MAXIMUM, AND CRITICAL
C POINTS FOR KTH CALIBRATION SOLUTION.
C CALL SCALE (N,XCALB,I,A,B)                         MAIN232
CALL LOCATE (N,XCALB,YCALB,YCRI,IGRAD,LMN1,LMXI,NCR1,LCR1,XLOC)   MAIN233
YMIN=YCALB(LMN1)
YMAX=YCALB(LMX1)
IF (YMIN .LT. YBCMIN) YBCMIN=YMIN
IF (YMAX .GT. YBCMAX) YBCMAX=YMAX
MAIN234
MAIN235
MAIN236
MAIN237
MAIN238
MAIN239
MAIN240
MAIN241
MAIN242
MAIN243
MAIN244
MAIN245
MAIN246
MAIN247
MAIN248
MAIN249
MAIN250
MAIN251
MAIN252
MAIN253
MAIN254
MAIN255
MAIN256
MAIN257
MAIN258
MAIN259
MAIN260
MAIN261
MAIN262
MAIN263
MAIN264
MAIN265
MAIN266
MAIN267
MAIN268
MAIN269
MAIN270
MAIN271
MAIN272
MAIN273
MAIN274
MAIN275
MAIN276
MAIN277
MAIN278
MAIN279
MAIN280
MAIN281
MAIN282
MAIN283
MAIN284
MAIN285
XFIX1(I)=0.0
XFIX1(NFIX)=1.0
IF (LSPEC .EQ. 1) WRITE (6,1180)
DO 100 I=1,NSELCT
IF (LSPEC .EQ. 0) GO TO 90
XFIX1(I)=XCALB(LOC1(I))
WRITE (6,1190) LOC1(I)
GO TO 100
90 CONTINUE
100 CONTINUE
XFIX1(I+1)=XLOC1(LSELCT(I))
C.....ARRANGE SELECTED FIXED POINTS IN A MONOTONE SEQUENCE.
C CALL SORT (NFIIX,XFIX1,ISEQ1)
WRITE (6,1200)
WRITE (6,1150)
CALL UPLW (A,B,XFIX1,B,NFIIX,XOUT,FLAG)
WRITE (6,1210) (I,XOUT(I),FLAG(I),I=1,NFIIX)
C.....STOP EXECUTION IF ORDER OF OCCURRENCE OF CRITICAL POINTS IN BASE
C AND CALIBRATION SOLUTIONS DOES NOT CORRESPOND.
C IF (LSPEC .EQ. 1) GO TO 120
DO 110 I=1,NFIIX
IF (ISEQ0(I) .NE. ISEQ1(I)) GO TO 910
110 CONTINUE
120 CONTINUE
C.....COMPUTE COEFFICIENTS IN KTH UNIT STRAINING OF XBASE
C XSTR = C(K,I) + D(K,I)*XBASE. I=1,2, ..., NSEG,
C WHERE NSEG IS THE NUMBER OF LINEAR SEGMENTS.
C DO 130 I=1,NSEG
CNUM=XFIX1(I)*XFIX0(I+1)-XFIX1(I+1)*XFIX0(I)
DNUM=XFIX1(I+1)-XFIX1(I)
DENOM=XFIX0(I+1)-XFIX0(I)
C(K,I)=CNUM/DENOM
D(K,I)=DNUM/DENOM
130 CONTINUE
C.....DETERMINE KTH UNIT STRAINING OF XBASE.
C CALL STRAIN (N,K,NSEG,XFIX0,XBASE,1.0,DELX)
DO 140 I=1,N
140 XUNIT(I)=XBASE(I)+DELX(I)
C.....INTERPOLATE CALIBRATION SOLUTION TO BASE FLOW POINTS CORRESPONDING
C TO UNIT STRAINING.
C CALL INTERP (N,XCALB,YCALB,XUNIT,YINTP)
C.....CORRECT VALUES ON EITHER SIDE OF CRITICAL POINTS, IF THESE ARE
C USED IN STRAINING.
C IF (LCORR .EQ. 0) GO TO 160
DO 150 I=1,NCR1
YINTP(LCR0(I))=YCALB(LCR1(I))
YINTP(LCR0(I)+1)=YCALB(LCR1(I)+1)
150 CONTINUE
160 CONTINUE
C.....DETERMINE THE KTH UNIT PERTURBATION.
C
MAIN296
MAIN297
MAIN298
MAIN299
MAIN290
MAIN291
MAIN292
MAIN293
MAIN294
MAIN295
MAIN296
MAIN297
MAIN298
MAIN299
MAIN300
MAIN301
MAIN302
MAIN303
MAIN304
MAIN305
MAIN306
MAIN307
MAIN308
MAIN309
MAIN310
MAIN311
MAIN312
MAIN313
MAIN314
MAIN315
MAIN316
MAIN317
MAIN318
MAIN319
MAIN320
MAIN321
MAIN322
MAIN323
MAIN324
MAIN325
MAIN326
MAIN327
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MAIN334
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MAIN337
MAIN338
MAIN339
MAIN340
MAIN341
MAIN342

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DO 170 I=1,N
170 YUNIT(K,I)=(YINTP(I)-YBASE(I))/DELI(K)
C.....SAVE UNIT STRAINING IF REQUIRED FOR LATER PRINTOUT.
C
IF (LUNIT .EQ. 0) GO TO 180
CALL SCALE (N,XUNIT+2,A+B)
CALL COPY (1,N,K,XUNIT,XUSAVE)
180 CONTINUE
C.....END CALCULATIONS ON CALIBRATION SOLUTIONS.
C
C*****PRINT UNIT PERTURBATION(S) AND UNIT STRAINING(S) IF LUNIT .NE. 0.
C
IF (LUNIT .EQ. 0) GO TO 240
CALL SCALE (N,XBASE+2,A+B)
IRPT=0
IF (NPARAM .GT. 4) IRPT=1
KSTART=1
KSTOP=4
IF (KSTOP .GT. NPARAM) KSTOP=NPARAM
GO TO 200
190 KSTART=5
KSTOP=NPARAM
200 CONTINUE
WRITE (6,1280) VNAM
IF (NPARAM .GT. 1) WRITE (6,1290) KSTART,KSTOP
IF (NPARAM .EQ. 1) WRITE (6,1300)
IF (NPARAM .EQ. 1) GO TO 210
NUM=KSTOP-KSTART+1
IF (NUM .EQ. 1) WRITE (6,1310) (ORD(K),K=KSTART,KSTOP)
IF (NUM .EQ. 2) WRITE (6,1320) (ORD(K),K=KSTART,KSTOP)
IF (NUM .EQ. 3) WRITE (6,1330) (ORD(K),K=KSTART,KSTOP)
IF (NUM .EQ. 4) WRITE (6,1340) (ORD(K),K=KSTART,KSTOP)
210 CONTINUE
CALL FILL (1+0,STRU1)
KLAST=20*(KSTOP-KSTART+1)
WRITE (6,1350) (STRU1(K),K=1,KLAST)
WRITE (6,1360)
DO 220 I=1,N
220 WRITE (6,1370) I,XBASE(I),(XUSAVE(K,I)),YUNIT(K,I),K=KSTART,KSTOP
IF (IRPT .EQ. 0) GO TO 230
IRPT=0
GO TO 190
230 CALL SCALE (N,XBASE+1,A+B)
240 CONTINUE
C.....CONSTRUCT PERTURBATION SOLUTIONS FOR TEST CASES (AND COMPARE WITH
C EXACT SOLUTION, IF AVAILABLE).
C
DO 330 ICASE=1,NCASE
CALL INPUT (4)
M2SAVE(ICASE)=M2
YCR2=YCR1(M2)
YCR3=YCR2
MAIN343
MAIN344
MAIN345
MAIN346
MAIN347
MAIN348
MAIN349
MAIN350
MAIN351
MAIN352
MAIN353
MAIN354
MAIN355
MAIN356
MAIN357
MAIN358
MAIN359
MAIN360
MAIN361
MAIN362
MAIN363
MAIN364
MAIN365
MAIN366
MAIN367
MAIN368
MAIN369
MAIN370
MAIN371
MAIN372
MAIN373
MAIN374
MAIN375
MAIN376
MAIN377
MAIN378
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MAIN380
MAIN381
MAIN382
MAIN383
MAIN384
MAIN385
MAIN386
MAIN387
MAIN388
MAIN389
MAIN390
MAIN391
MAIN392
MAIN393
MAIN394
MAIN395
MAIN396
MAIN397
MAIN398
MAIN399
C.....INITIALIZE STRAINED COORDINATE AND PERTURBATION SOLUTION.
C
DO 250 I=1,N
XPERT(I)=XBASE(I)
250 YPERT(I)=YBASE(I)
C.....ADD IN CONTRIBUTIONS FROM ALL PERTURBATIONS.
C
DO 270 K=1,NPARAM
Q2SAVE(K,ICASE)=Q2(K)
DEL2=Q2(K)-Q0(K)
DEL21=DEL2/DELI(K)
CALL STRAIN (N,K,NSEG,XFIX0,XBASE,DEL21+DELX)
DO 260 I=1,N
XPERT(I)=XPERT(I)+DELX(I)
260 YPERT(I)=YPERT(I)+DEL2+YUNIT(K,I)
270 CONTINUE
C.....ADJUST VALUES NEAR CRITICAL POINT FOR MONOTONE BEHAVIOR.
C
IF (LCORR .EQ. 1) CALL MONO (INCR0,LCR0,XPERT,YPERT)
C.....LOCATE MINIMUM, MAXIMUM, AND CRITICAL POINTS IN PERTURBATION
C SOLUTION.
C
CALL SCALE (N,XPERT+2,A+B)
CALL SCALE (N,XPERT,1,A+B)
CALL LOCATE (N,XPERT,YPERT,YCR2+IGRAD,LMN2,LMX2,NCR2,LCR2,XLOC2)
YMIN=YPERT(LMN2)
YMAX=YPERT(LMX2)
YPCMIN=YMIN
YPCMAX=YMAX
WRITE (6,1380) ICASE,NCASE
WRITE (6,1390) VNAM
WRITE (6,1390) M2
IF (NPARAM .EQ. 1) WRITE (6,1400) Q2(I),PARNAME(I)
IF (NPARAM .GT. 1) WRITE (6,1410) (K,Q2(K),PARNAME(K),K=1,NPARAM)
WRITE (6,1130) VNAM,YCR2
WRITE (6,1140)
WRITE (6,1150)
CALL UPLW (A+B,XLOC2,6,NCR2+2,XOUT,FLAG)
WRITE (6,1420) HEAD2,XOUT(1),FLAG(1),LMN2,XOUT(2),FLAG(2),LMX2
IF (NCR2 .GT. 0) WRITE (6,1430) NCR2,
% (ORD(I),XOUT(I+2),FLAG(I+2),LCR2(I),I=1,NCR2)
CALL SCALE (N,XBASE+2,A+B)
C.....IF LCHEK .NE. 0 READ IN DATA FOR COMPARISON SOLUTION AND LOCATE
C MINIMUM, MAXIMUM, AND CRITICAL POINTS.
C
IF (LCHEK .EQ. 0) GO TO 290
CALL INPUT (5)
CALL SCALE (N,XCHEK,1,A+B)
CALL LOCATE (N,XCHEK,YCHEK,YCR3,IGRAD,LMN3,LMX3,NCR3,LCR3,XLOC3)
YMIN=YCHEK(LMN3)
YMAX=YCHEK(LMX3)
IF (YMIN .LT. YPCMIN) YPCMIN=YMIN
MAIN400
MAIN401
MAIN402
MAIN403
MAIN404
MAIN405
MAIN406
MAIN407
MAIN408
MAIN409
MAIN410
MAIN411
MAIN412
MAIN413
MAIN414
MAIN415
MAIN416
MAIN417
MAIN418
MAIN419
MAIN420
MAIN421
MAIN422
MAIN423
MAIN424
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MAIN446
MAIN447
MAIN448
MAIN449
MAIN450
MAIN451
MAIN452
MAIN453
MAIN454
MAIN455
MAIN456

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IF (YMAX .GT. YPCMAY) YPCMAY=YMAX
CALL UPLW (A,B,XLOC3,6+NCR3+2,XOUT,FLAG)
WRITE (6,1420) HEAD3,XOUT(1),FLAG(1),LMN3,XOUT(2),FLAG(2),LMX3
IF (NCR3 .GT. 0) WRITE (6,1430) NCR3,
% (ORD1),XOUT(I+2),FLAG(I+2),LCR3(I),I=1,NCR3
CALL INTERP (N,XPERT,YPERT,XCHEK,YPRTI)
CALL LOCATE (N,XCHEK,YPRTI,YCR3,(GRAD,LMN3,LMX3,NCR3,XLOC3)
YMIN=YPRTI(LMN3)
YMAX=YPRTI(LMX3)
IF (YMIN .LT. YMIN) YMIN=YMIN
IF (YMAX .GT. YMAX) YMAX=YMAX
CALL SCALE (N,XPERT,2,A,B)
CALL SCALE (N,XCHEK,2,A,B)
CALL FILL (2,0,STRING)
WRITE (6,1440) VNAM
WRITE (6,1450) VNAM,YMAX,VNAM,YMIN,VNAM,YCR2,VNAM,VNAM
WRITE (6,1460) VNAM,VNAM,VNAM,VNAM,(STRING(I),I=1,72)
DO 280 I=1,N
CALL FILL (3,I,STRING)
280 WRITE (6,1470) I,XBASE(I),YBASE(I),XPERT(I),YPERT(I),
% XCHEK(I),YCHEK(I),YPRTI(I),(STRING(I),I=1,72)
GO TO 310
290 CONTINUE
C
CALL SCALE (N,XPERT,2,A,B)
CALL FILL (2,0,STRING)
WRITE (6,1440) VNAM
WRITE (6,1480) VNAM,YMAX,VNAM,YMIN,VNAM,YCR2,VNAM
WRITE (6,1490) VNAM,VNAM,STRING
DO 300 I=1,N
CALL FILL (3,I,STRING)
300 WRITE (6,1500) I,XBASE(I),YBASE(I),XPERT(I),YPERT(I),STRING
310 CONTINUE
C.....IF LPLOT .NE. 0 GENERATE PERIPHERAL PLOT OF PERTURBATION AND
C COMPARISON SOLUTIONS.
C
IF (LPLOT .EQ. 0) GO TO 320
YMIN=YBCMIN
YMAX=YBCMAX
IF (YPCMIN .LT. YMIN) YMIN=YPCMIN
IF (YPCMAY .GT. YMAX) YMAX=YPCMAY
CALL DRYPLT (ICASE,N,NCASE,NPARAM,YMIN,YMAX,YCR2)
320 CONTINUE
CALL SCALE (N,XBASE,I,A,B)
330 CONTINUE
C.....PRINT FINAL TABLE SUMMARIZING CALCULATIONS.
C
CALL TABLE (INPARAM,NCASE,PARNAM,M0,Q0)
GO TO 999
C.....ABNORMAL TERMINATION OF COMPUTATION.
C
900 WRITE (6,9000)
GO TO 999
905 WRITE (6,9050)

MAIN457          GO TO 999
MAIN458          910 WRITE (6,9100)
MAIN459          C
MAIN460          999 WRITE (6,9500)
MAIN461          STOP
MAIN462          C.....I/O FORMAT STATEMENTS FOLLOW.
MAIN463
MAIN464
MAIN465          1000 FORMAT (1H1+132(1H*)/
MAIN466          %           1X,1H*+25X,80A1+25X,1H*/%
MAIN467          %           1X,132(1H*)//)
MAIN468          1010 FORMAT (1X,29H....LIST OF INPUT PARAMETERS//%
MAIN469          %           6X,3H =,14//%
MAIN470          %           6X,3H =,F5.1+4X,3H =,F5.1//%
MAIN471          %           6X,BNPARAM =,12//)
MAIN472          1020 FORMAT (1X,22H....STRAINING OPTIONS//%
MAIN473          %           6X,23HNUMBER OF FIXED POINTS!,12//)
MAIN474          1030 FORMAT (1X,40HFIXED POINTS WILL BE PRESELECTED BY USER/
MAIN475          %           6X,37HAND LISTED BELOW IN PRINTOUT FOR BASE/
MAIN476          %           6X,26HAND CALIBRATION SOLUTIONS.///)
MAIN477          1040 FORMAT (1X,45HFIXED POINTS WILL BE AUTOMATICALLY DETERMINED/
MAIN478          %           6X,44HBY THE PROGRAM FOR ALL SOLUTIONS AS FOLLOWS://%
MAIN479          %           1X,14HTWO END POINTS!
MAIN480          1050 FORMAT (1IX,16HPOINT OF MINIMUM,1X,2A1)
MAIN481          1060 FORMAT (1IX,16HPOINT OF MAXIMUM,1X,2A1)
MAIN482          1070 FORMAT (1IX,2A1+6MCRIT (,A+,6HPOINT))
MAIN483          1080 FORMAT (1H1+26HRESULTS OF COMPUTATIONS ON,1X,5A4//)
MAIN484          1090 FORMAT (1X,17H....MACH NUMBER,/
MAIN485          %           6X,34HVALUES OF PERTURBATION PARAMETERS,/
MAIN486          %           6X,17HCRITICAL VALUE OF,1X,2A1,1H//)
MAIN487          1100 FORMAT (1IX,4H0 =,F7.4/)
MAIN488          1110 FORMAT (1IX,4HQ0 =,F7.4+5X+1H(A8,1H)/)
MAIN489          1120 FORMAT (1IX,3H00(I,1),3H) =,F7.4+5X+1H(A8,1H)/)
MAIN490          1130 FORMAT (1IX,2A1+6MCRIT =,F6.4//)
MAIN491          1140 FORMAT (1X,47H....LOCATIONS OF MIN., MAX., AND CRITICAL PTS.)
MAIN492          1141 FORMAT (1X,3A1(I) DENOTES POINT ON LOWER SURFACE)/)
MAIN493          1142 FORMAT (1X,3A1(H) DENOTES POINT ON UPPER SURFACE)/)
MAIN494          1143 FORMAT (1X,14HMINIMUM AT X =,F7.4+A1+3X+10H(POINT NO.,I4,1H)/)
MAIN495          1144 FORMAT (1X,14HMAXIMUM AT X =,F7.4+A1+3X+10H(POINT NO.,I4,1H)/)
MAIN496          1145 FORMAT (1X,11,1X,18MCRITICAL POINT(S)!)
MAIN497          %           10X,A4+6HT X =,1X,F6.4+A1+3X,
MAIN498          %           16H(AFTER POINT NO.,I4,1H))
MAIN499          1150 FORMAT ((1X,5(IH*))+1X,32HFIXED POINTS PRESELECTED BY USER,
MAIN500          %           1X,5(IH*)/ )
MAIN501          1160 FORMAT (1X,2H(X)(I3,1H))
MAIN502          1170 FORMAT (1X,29H....LOCATION OF FIXED POINTS)
MAIN503          1180 FORMAT (1IX,5HFIX(I,1),3H) =,F7.4,A1)
MAIN504          1190 FORMAT (1H1+26HRESULTS OF COMPUTATIONS ON,1X,6A4//)
MAIN505          1200 FORMAT (1IX,4H0 =,F7.4+5X+1H(A8,1H)/)
MAIN506          1210 FORMAT (1IX,4H0 =,F7.4+5X+1H(A8,1H)/)
MAIN507          1220 FORMAT (1IX,4H0 =,F7.4+5X+1H(A8,1H)/)
MAIN508          1230 FORMAT (1IX,4H0 =,F7.4+5X+1H(A8,1H)/)
MAIN509          1240 FORMAT (1X,4H0 =,F7.4+5X+1H(A8,1H)/)
MAIN510          1250 FORMAT (1IX,4H0 =,F7.4+5X+1H(A8,1H)/)
MAIN511          1260 FORMAT (1X,5H*01(I,1),3H) =,F7.4+5X+1H(A8,1H)/)
MAIN512          1270 FORMAT (1IX,3H01(I,1),3H) =,F7.4+5X+1H(A8,1H)/)
MAIN513          %           27HAND UNIT STRAINING OF XBASE)
MAIN514          1280 FORMAT (120HFOR CALIBRATION SOLUTIONS,12+1X,7THROUGH,12)
MAIN515          1290 FORMAT (1//19X,1(IH*)+1X,A4+1)HCALB SOLN *3X)
MAIN516          1300 FORMAT (120HFOR CALIBRATION SOLUTIONS,12+1X,7THROUGH,12)
MAIN517          1310 FORMAT (1//19X,1(IH*)+1X,A4+1)HCALB SOLN *3X)

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1320 FORMAT (//19X,2(1H#,1X,A4),1MCALB SOLN *3X))
1330 FORMAT (//19X,3(1H#,1X,A4),1MCALB SOLN *3X))
1340 FORMAT (//19X,(1H#,1X,A4),1MCALB SOLN *3X))
1350 FORMAT (1X,5POINT,4X,5HXBSE,4X,96A1)
1360 FORMAT (1X)
1370 FORMAT (1X,I4,1X,1I1F10.4)
1380 FORMAT (1H,32(1H#)/
      5   1X,21H* OUTPUT FOR CASE NO.,13,4H OF ,12+2H */
      5   1X,32(1H#)//)
1390 FORMAT (1IX,4H#2 =F7.4/1
1400 FORMAT (1IX,4HQ2 =F7.4,5X,1H(A,B,1H)/)
1410 FORMAT (1IX,3HQ2,(1+3H) =F7.4,5X,1H(A,B,1H)/)
1420 FORMAT (/6X,5A4//
      5   1IX,14HMINIMUM AT X =F7.4,A1,3X,10H(POINT NO.,[4..]H)/
      5   1IX,14HMAXIMUM AT X =F7.4,A1,3X,10H(POINT NO.,[4..]H))
1430 FORMAT (1H,10X,1I,1X,16HCRITICAL POINT(S):/
      5   ([5X,A4,6HAT X =,F7.4,A1,3X,
      5   16H(AFTER POINT #0..14,1H)))
1440 FORMAT (//1X,44H....FINAL PRINTOUT AND GRAPHICAL DISPLAY OF,1X,
      5   2A1)
1450 FORMAT (/72X,21H# = MAXIMUM VALUE OF,1X,2A1,1X,1H#,F8.4/
      5   72X,21H# = MINIMUM VALUE OF,1X,2A1,1X,1H#,F8.4/
      5   72X,21H# = CRITICAL VALUE OF,1X,2A1,1X,1H#,F8.4/
      5   72X,12HP = VALUE OF,1X,2A1,1X,
      5   34HPREDICTED BY PERTURBATION SOLUTION/
      5   72X,12HC = VALUE OF,1X,2A1,1X,
      5   22HIN COMPARISON SOLUTION/
      5   72X,29HS = AGREEMENT BETWEEN P AND C)
1460 FORMAT (/2X,2HPT,2X,5HXBSE,3X,2A1+HBSE,2X,5HXPRT,3X,2A1+
      5   4HPERT,2X,5HCKECK,3X,2A1,4HCKECK,2X,2A1,4HPIINT,1X,72A1/)
1470 FORMAT (1X,13,7F8.4,1X,72A1)
1480 FORMAT (//60X,21H# = MAXIMUM VALUE OF,1X,2A1,1X,1H#,F8.4/
      5   60X,21H# = MINIMUM VALUE OF,1X,2A1,1X,1H#,F8.4/
      5   60X,21H# = CRITICAL VALUE OF,1X,2A1,1X,1H#,F8.4/
      5   60X,12HP = VALUE OF,1X,2A1,1X,
      5   34HPREDICTED BY PERTURBATION SOLUTION)
1490 FORMAT (/2X,2HPT,2X,5HXBSE,3X,2A1+HBSE,2X,5HXPRT,3X,2A1+
      5   4HPERT,1X,96A1/)
1500 FORMAT (1X,I3,4F8.4+1X,96A1)
C.....ABNORMAL TERMINATION FORMATS FOLLOW.
C
9000 FORMAT (//1X,2BHNUMBER OF CRITICAL POINTS IN/
      5   1X,30HBASE AND CALIBRATION SOLUTIONS/
      5   1X,31HARE UNEQUAL - CALCULATION ENDED)
9050 FORMAT (//1X,25HNUMBER OF CRITICAL POINTS/
      5   1X,23HSELECTED EXCEEDS NUMBER/
      5   1X,30HACTUALLY LOCATED - CALCULATION/
      5   1X,5MENDED)
9100 FORMAT (//1X,2BHORDER OF SPECIFIED POINTS IN/
      5   1X,30HBASE AND CALIBRATION SOLUTIONS/
      5   1X,39HDOES NOT CORRESPOND - CALCULATION ENDED)
9500 FORMAT (1H1)
END

MAIN571          SUBROUTINE BANNER
MAIN572          C.....PRINTS TWO BANNER PAGES WITH JOB KEYWORD IN UPPER RIGHT CORNER.
MAIN573          COMMON /HEAD/TITLE(80)
MAIN574          READ (5,1000) TITLE
MAIN575          00 10 1PAGE=1,2
MAIN576          WRITE (6,1020) (TITLE(I),I=1,9)
MAIN577          WRITE (6,1030)
MAIN578          WRITE (6,1040)
MAIN579          10 CONTINUE
MAIN580          RETURN
MAIN581          1000 FORMAT (80A1)
MAIN582          1020 FORMAT (1H1,119X,13(1H#)/120X,2H# /*120X,13(1H#))
MAIN583          1030 FORMAT (////38X,55(1H#)/38X,1H#*53X,1H#/
      5   38X,1H#*19X,15HPROGRAM PERTURE,19X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*7X,39HCALCULATES NONLINEAR MULTIPLE-PARAMETER,
      5   7X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*13X,27HCONTINUOUS OR DISCONTINUOUS,
      5   13X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*15X,22HPERTURBATION SOLUTIONS,16X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*10X,33HWICH REPRESENT CHANGES IN EITHER,
      5   10X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*13X,27HGEOMETRY OR FLOW CONDITIONS,
      5   13X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*4X,44HBY EMPLOYING A STRAINED-COORDINATE PROCEDURE,
      5   5X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*4X,44HUTILIZING UNIT PERTURBATIONS DETERMINED FROM,
      5   5X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*16X,21HPREVIOUSLY CALCULATED,
      5   16X,1H#/*38X,1H#*53X,1H#/
      5   1040 FORMAT (38X,1H#*9X,34H#BASE AND "CALIBRATION" SOLUTIONS,
      5   10X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*4X+45HDISPLACED FROM ONE ANOTHER BY SOME REASONABLE,
      5   4X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*8X+36HCHANGE IN GEOMETRY OR FLOW CONDITION,
      5   9X,1H#*8X,1H#*53X,1H#/
      5   38X,1H#*53X,1H#/
      5   38X,1H#*21X,10HWRITTEN BY,22X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*7X,39HJAMES P. ELLIOTT AND STEPHEN S. STAHAHA,
      5   7X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*53X,1H#/
      5   38X,1H#*7X+38HNIELSEN ENGINEERING AND RESEARCH, INC..
      5   8X,1H#/*38X,1H#*53X,1H#/
      5   38X,1H#*14X,25HMOUNTAIN VIEW, CALIFORNIA,14X,1H#/*38X,1H#*53X,1H#/
      5   38X,55(1H#))
END

SUBROUTINE DRVPLT (ICASE,N,NCASE,NPARAM,YMIN,YMAX,YCH2)
DIMENSION HLINE2(3),HLINE3(3)
DIMENSION XCSAVE(8,200),YCSAVE(8,200)
DIMENSION XBASE(200),XCALB(200),XPERT(200),XCHEK(200),YBASE(200),
      5   YCALB(200),YPERT(200),YCHEK(200),YPRTL(200)
DRVPL 1
DRVPL 2
DRVPL 3
DRVPL 4
DRVPL 5

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COMMON /HEAD/ ZTITLE(80)
COMMON /SAVE/ XCSAVE,YCSAVE
COMMON /XY/ XBASE,XCALB,XPERT,XCHEK,YBASE,YCALB,YPERT,YCHEK,YPRTI
DATA NPLOT /0/
IF (NPLOT .EQ. 0) CALL BETA
MIN=10.0*(YMIN-0.1)
MAX=10.0*(YMAX+0.1)
YMIN=0.1*MIN
YMAX=0.1*MAX
ENCODE (26,1000,HLINE2) (ZTITLE(I),I=1,9),ICASE,NCASE
DO 20 K=1,NPARAM
NPLOT=NPLOT-1
ENCODE (22,1010,HLINE3) K,NPARAM
CALL BGML (-1)
CALL MXALF ("L/CST")
CALL MX3ALF ("INSTR",HSM)
CALL SIMPLX
CALL TITLE (IM =1,IMX=1,"SE0.5")CNE(X(P)$",100,6,0+0)
CALL HEADIN (NPLOT (0F) CSL0.25H0.7(P)$",100,3,3)
CALL HEADIN (HLINE2,28,2,3)
CALL HEADIN (HLINE3,22,2,3)
CALL GRAF (0,0,"SCALE",1,0,YMAX,"SCALE",YMIN)
CALL FRAME
IF (YCR2 .GT. YMAX) GO TO 10
CALL RLVEC (0,0,YCR2,0.2*YCR2,0,000)
CALL RLMESS ("CBL(P%BE(*)$",100,0.21,YCR2)
10 CONTINUE
CALL DASH
CALL CURVE (XBASE,YBASE,N=0)
CALL RESET ("DASH")
CALL DOT
CALL COPY (2,M,K,XCALB,XCSAVE)
CALL COPY (2,N,K,YCALB,YCSAVE)
CALL CURVE (XCALB,YCALB,N=0)
CALL RESET ("DOT")
CALL MARKER (1)
CALL CURVE (XPERT,YPERT,N=-1)
CALL RESET ("MARKER")
CALL CURVE (XCHEK,YCHEK,N=0)
CALL ENDP (NPLOT)
20 CONTINUE
1000 FORMAT (9A1,I1H) CASE NO. +I2+4H OF +I2)
1010 FORMAT (16HCALIBRATION NO. +I1,4H OF +I1)
RETURN
END

SUBROUTINE ECHIMP
DIMENSION CARD(20)
READ (5,1000) LECHO
IF (LECHO .EQ. 0) GO TO 10
WRITE (6,1010)
BACKSPACE 5
BACKSPACE 5
10 READ (5,1020) CARD
IF (EOF(5)) 30,20
20 WRITE (1,1020) CARD
      DRVP 6           IF (LECHO .EQ. 1) WRITE (6,1030) CARD
      DRVP 7           GO TO 10
      DRVP 8           30 REWIND 1
      DRVP 9           IF (LECHO .EQ. 0) RETURN
      DRVP 10          READ (1,1020) CARD
      DRVP 11          READ (1,1020) CARD
      DRVP 12          RETURN
      DRVP 13          1000 FORMAT (15)
      DRVP 14          1010 FORMAT ((H1)25(IH0)/
      DRVP 15             $   IX,IH$,IX,2)MLISTING OF INPUT DECK,IX,IH$/
      DRVP 16             $   IX,25(IH0)///
      DRVP 17          1020 FORMAT (20A)
      DRVP 18          1030 FORMAT (IX,20A)
      DRVP 19          END

      SUBROUTINE FILL (ICALL,I,STRING)
C.....FILLS ARRAY STRING WITH CHARACTERS FOR TABLE HEADINGS AND PRINTER PLOTS.
      C
      DIMENSION XBASE(200),XCALB(200),XCHEK(200),XPERT(200)
      DIMENSION YBASE(200),YCALB(200),YPERT(200),YCHEK(200),YPRTI(200)
      DIMENSION STRING(96),VNAM(2),UNIT(20),PARNAM(8)
      REAL H0,M1,M2
      COMMON /PARAM/ PARNAM,L0CO=L0C1,LSELECT=N,NCASE,LSPEC,UNIT,
      %           LCHEK,LPLOT,NSELECT,A,B,NPARAM,VNAM
      COMMON /PERT/ Q0,Q2,M0,M1,M2,O1,YCR0,YCR1,YCR2,YMIN,YMAX
      COMMON /AV/ XBASE,XCALB,XPERT,XCHEK,YBASE,YCALB,YPERT,YCHEK,YPRTI
      DATA IENT /0/
      DATA STAR /H0/, P/1HP/, C/IHC/, DASH/IH-/ , H/IHH/, EL/IHL/
      DATA BLANK/IH /, DOLLAR/1HS/
      DATA UNIT /IH,X,1HS,IHT,1HR,1HU,1HN,IHI,IHT,1M +
      %           1H +1H +1M +1HU,1HN,IHI+1HT,1H +1H ,1H /
      %           1H +1H +1M +1HU,1HN,IHI+1HT,1H +1H ,1H /
      GO TO (10+30+50), ICALL
10 IENT=IENT+1
      IF (IENT .GT. 1) RETURN
      DRVP 43          UNIT(1)=VNAM(1)
      DRVP 44          UNIT(1)=VNAM(2)
      DRVP 45          DO 20 J=1,20
      DRVP 46          DO 20 K=1,4
      DRVP 47          III=J+20*(K-1)
      DRVP 48          20 STRING(III)=UNIT(J)
      DRVP 49          RETURN
      DRVP 50          NY=96
      30 NY=96
      IFLAG=0
      IF (YCR2 .GT. YMAX .OR. YCR2 .LT. YMIN) IFLAG=1
      ECHI 1           IF (ILCHEK .EQ. 1) NY=72
      ECHI 2           RANGE=YMAX-YMIN
      ECHI 3           DO 40 J=1,NY
      ECHI 4           40 STRING(1)=DASH
      ECHI 5           STRING(1)=H
      ECHI 6           STRING(NY)=EL
      ECHI 7           IF (IFLAG .EQ. 1) RETURN
      ECHI 8           NSTAR=1+(YMAX-YCR2)/RANGE*(NY-1)
      ECHI 9           STRING(INSTAR)=STAR
      ECHI 10          RETURN
      ECHI 11          ECHI 12
      ECHI 13          ECHI 14
      ECHI 14          ECHI 15
      ECHI 15          ECHI 16
      ECHI 16          ECHI 17
      ECHI 17          ECHI 18
      ECHI 18          ECHI 19
      ECHI 19          ECHI 20
      ECHI 20          ECHI 21
      ECHI 21          ECHI 22
      ECHI 22          ECHI 23
      ECHI 23          ECHI 24
      ECHI 24          ECHI 25
      ECHI 25          ECHI 26
      ECHI 26          ECHI 27
      ECHI 27          ECHI 28
      ECHI 28          ECHI 29
      ECHI 29          ECHI 30
      ECHI 30          ECHI 31
      ECHI 31          ECHI 32
      ECHI 32          ECHI 33
      ECHI 33          ECHI 34
      ECHI 34          ECHI 35
      ECHI 35          ECHI 36
      ECHI 36          ECHI 37
      ECHI 37          ECHI 38
      ECHI 38          ECHI 39
      ECHI 39          ECHI 40
      ECHI 40          ECHI 41
      ECHI 41          ECHI 42
      ECHI 42          ECHI 43
      ECHI 43          ECHI 44
      ECHI 44          ECHI 45
      ECHI 45          ECHI 46
      ECHI 46          ECHI 47
      ECHI 47          ECHI 48
      ECHI 48          ECHI 49
      ECHI 49          ECHI 50
      ECHI 50          ECHI 51
      ECHI 51          ECHI 52
      ECHI 52          ECHI 53
      ECHI 53          ECHI 54
      ECHI 54          ECHI 55
      ECHI 55          ECHI 56
      ECHI 56          ECHI 57
      ECHI 57          ECHI 58
      ECHI 58          ECHI 59
      ECHI 59          ECHI 60
      ECHI 60          ECHI 61
      ECHI 61          ECHI 62
      ECHI 62          ECHI 63
      ECHI 63          ECHI 64
      ECHI 64          ECHI 65
      ECHI 65          ECHI 66
      ECHI 66          ECHI 67
      ECHI 67          ECHI 68
      ECHI 68          ECHI 69
      ECHI 69          ECHI 70
      ECHI 70          ECHI 71
      ECHI 71          ECHI 72
      ECHI 72          ECHI 73
      ECHI 73          ECHI 74
      ECHI 74          ECHI 75
      ECHI 75          ECHI 76
      ECHI 76          ECHI 77
      ECHI 77          ECHI 78
      ECHI 78          ECHI 79
      ECHI 79          ECHI 80
      ECHI 80          ECHI 81
      ECHI 81          ECHI 82
      ECHI 82          ECHI 83
      ECHI 83          ECHI 84
      ECHI 84          ECHI 85
      ECHI 85          ECHI 86
      ECHI 86          ECHI 87
      ECHI 87          ECHI 88
      ECHI 88          ECHI 89
      ECHI 89          ECHI 90
      ECHI 90          ECHI 91
      ECHI 91          ECHI 92
      ECHI 92          ECHI 93
      ECHI 93          ECHI 94
      ECHI 94          ECHI 95
      ECHI 95          ECHI 96
      ECHI 96          ECHI 97
      ECHI 97          ECHI 98
      ECHI 98          ECHI 99
      ECHI 99          ECHI 100
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C          COEFFICIENT.           INPUT1
C
C**** ITEM NO. 6 - ONE CARD (10AR) ***** INPUT101      INPUT158
C          PARMAM(K) ...           INPUT102      INPUT159
C          ARRAY OF 8-CHARACTER STRINGS WHICH IDENTIFY THE INPUT103      INPUT160
C          PARAMETERS VARIED. NPARAM ELEMENTS OF THE ARRAY INPUT104      INPUT161
C          ARE READ IN.           INPUT105      INPUT162
C
C**** ITEM NO. 7 - ONE CARD (8F10.6) ***** INPUT106      INPUT163
C          A      SCALING PARAMETER (A = -X(1), WHERE X(1) IS FIRST INPUT107      INPUT164
C          DATA POINT ON LOWER SURFACE ... SEE MANUAL).       INPUT108      INPUT165
C
C          B      SCALING PARAMETER (B = X(N), WHERE X(N) IS LAST DATA INPUT109      INPUT166
C          POINT ON UPPER SURFACE ... SEE MANUAL).           INPUT110      INPUT167
C
C**** ITEM NO. 8 - ONE CARD (8F10.6) ***** INPUT111      INPUT168
C          Q0(K)    ARRAY OF LENGTH B GIVING VALUES OF PERTURBATION INPUT112      INPUT169
C          PARAMETERS IN BASE SOLUTION. NPARAM ELEMENTS OF THE INPUT113      INPUT170
C          ARRAY ARE READ IN.           INPUT114      INPUT171
C
C**** ITEM NO. 9 - ONE CARD (8F10.6) ***** INPUT115      INPUT172
C          LOC1(I)   ARRAY OF LENGTH 6 OF WHICH NSELCT ELEMENTS ARE READ INPUT116      INPUT173
C          IN. SPECIFIES SUBSCRIPTS OF THOSE POINTS IN KTH INPUT117      INPUT174
C          CALIBRATION FLOW WHICH ARE TO BE HELD INVARIANT.     INPUT118      INPUT175
C
C**** ITEM NO. 10 - ONE SET OF C CARDS (8F10.6), C = 1 + INT(N/B) ***** INPUT119      INPUT176
C          XBASE(I), I=1,N ...           INPUT120      INPUT177
C          X COORDINATE IN BASE SOLUTION.           INPUT121      INPUT178
C
C**** ITEM NO. 11 - ONE SET OF C CARDS (8F10.6), C AS IN ITEM NO. 10 *** INPUT122      INPUT179
C          YBASE(I), I=1,N ...           INPUT123      INPUT180
C          DEPENDENT VARIABLE IN BASE SOLUTION.           INPUT124      INPUT181
C
C**** ITEM NO. 12 - ONE CARD (16IS) ***** INPUT125      INPUT182
C          ----- THIS ITEM TO BE OMITTED IF LSPEC = 0 ----- INPUT126      INPUT183
C          LOC0(I)    ARRAY OF LENGTH 6 OF WHICH NSELCT ELEMENTS ARE READ INPUT127      INPUT184
C          IN. SPECIFIES SUBSCRIPTS OF THOSE BASE FLOW POINTS INPUT128      INPUT185
C          WHICH ARE TO BE HELD INVARIANT.           INPUT129      INPUT186
C
C          ----- ONE SET OF ITEMS 13 THROUGH 16 IS           INPUT130      INPUT187
C          * NOTE * REQUIRED FOR EACH OF THE NPARAM           INPUT131      INPUT188
C          ----- CALIBRATION SOLUTIONS.           INPUT132      INPUT189
C
C**** ITEM NO. 13 - ONE CARD (8F10.6) ***** INPUT133      INPUT190
C          M1      ONCOMING MACH NUMBER IN KTH CALIBRATION SOLUTION. INPUT134      INPUT191
C
C          Q1      VALUE OF KTH PERTURBATION PARAMETER IN KTH INPUT135      INPUT192
C          CALIBRATION SOLUTION.           INPUT136      INPUT193
C
C**** ITEM NO. 14 - ONE SET OF C CARDS (8F10.6), C AS IN ITEM NO. 10 *** INPUT137      INPUT194
C          XCALB(I), I= 1,N ...           INPUT138      INPUT195
C          X COORDINATE IN KTH CALIBRATION SOLUTION.           INPUT139      INPUT196
C
C**** ITEM NO. 15 - ONE SET OF C CARDS (8F10.6), C AS IN ITEM NO. 10 *** INPUT140      INPUT197
C          YCALB(I), I=1,N ...           INPUT141      INPUT198
C          DEPENDENT VARIABLE IN KTH CALIBRATION SOLUTION..     INPUT142      INPUT199
C
C**** ITEM NO. 16 - ONE CARD (16IS) ***** INPUT143      INPUT200
C          ----- THIS ITEM TO BE OMITTED IF LSPEC = 0 ----- INPUT144      INPUT201
C          LOC1(I)    ARRAY OF LENGTH 6 OF WHICH NSELCT ELEMENTS ARE READ INPUT145      INPUT202
C          IN. SPECIFIES SUBSCRIPTS OF THOSE POINTS IN KTH INPUT146      INPUT203
C          CALIBRATION FLOW WHICH ARE TO BE HELD INVARIANT.     INPUT147      INPUT204
C
C          ----- ONE SET OF ITEMS 13 THROUGH 16 IS           INPUT148      INPUT205
C          * NOTE * REQUIRED FOR EACH OF THE NPARAM           INPUT149      INPUT206
C          ----- CALIBRATION SOLUTIONS.           INPUT150      INPUT207
C
C**** ITEM NO. 17 - ONE CARD (8F10.6) ***** INPUT151      INPUT208
C          XCHEK(I), I= 1,N ...           INPUT152      INPUT209
C          X COORDINATE IN COMPARISON SOLUTION.           INPUT153      INPUT210
C
C**** ITEM NO. 18 - ONE CARD (8F10.6) ***** INPUT154      INPUT211
C          YCHEK(I), I=1,N ...           INPUT155      INPUT212
C          DEPENDENT VARIABLE IN COMPARISON SOLUTION.     INPUT156      INPUT213
C
C          DIMENSION LOC0(6),LOC1(6),NSELCT(6)           INPUT157      INPUT214
C          DIMENSION XBASE(200)*XCALB(200)*XPERT(200)*XCHEK(200),     INPUT158
C          %           YBASE(200)*YCALB(200)*YPERT(200)*YCHEK(200)*YRPT1(200)
C          DIMENSION Q0(8),Q2(8)
C          REAL M0,M1,M2
C          DIMENSION VNAME(2),PARNAME(8)

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COMMON /PARAM/ PARMAM,LOC0,LOC1,LSELCT,N,NCASE,LSPEC,UNIT,
%          LCKECK,LPLOT,NSELCT,A,B,NPARAM,VNAM
COMMON /PERT/ DD=2,M0,M1,M2=0,YCR0,YCR1,YCH2,YMIN,YMAX
COMMON /XY/ XBASE,XCALB,XPERI,XCKEK,YBASE,YCALB,YPERI,YCKEK,YPRITI
GO TO (10,20,30,40,50), ICALL
10 READ (1,1000) N,NCASE,NPARAM,NSELCT,LSPEC,LUNIT,LCKECK,LPLOT
  IF (LSPEC .EQ. 0) READ (1,1000) (LSELCT(I),I=1,NSELCT)
  READ (1,1010) VNAM
  READ (1,1020) (PARNAME(K),K=1,NPARAM)
  READ (1,1030) A+B
  RETURN
20 READ (1,1030) M0
  READ (1,1030) (00(K),K=1,NPARAM)
  READ (1,1030) (XBASE(I),I=1,N)
  READ (1,1030) (YBASE(I),I=1,N)
  IF (LSPEC .EQ. 1) READ (1,1000) (LOC0(I),I=1,NSELCT)
  RETURN
30 READ (1,1030) M1,01
  READ (1,1030) (XCALB(I),I=1,N)
  READ (1,1030) (YCALB(I),I=1,N)
  IF (LSPEC .EQ. 1) READ (1,1000) (LOC1(I),I=1,NSELCT)
  RETURN
40 READ (1,1030) M2
  READ (1,1030) (Q2(K),K=1,NPARAM)
  RETURN
50 READ (1,1030) (XCKEK(I),I=1,N)
  READ (1,1030) (YCKEK(I),I=1,N)
  RETURN
1000 FORMAT (1A5)
1010 FORMAT (2A1)
1020 FORMAT (10A8)
1030 FORMAT (8F10.6)
END

      SUBROUTINE INTERP (N,X,Y,XI,YI)
C.....GIVEN THE SET OF POINTS X(I), Y(I), I=1,N, AND THE SET XI(J),
C..... J=1,N, USES LINEAR INTERPOLATION TO COMPUTE THE SET YI(J), J=1,N.
C
C      DIMENSION X(200),Y(200),XI(200),YI(200)
      NM1=N-1
      JSTART=1
      DO 70 I=1,N
        IF (XI(I) .LE. X(I)) GO TO 10
        IF (XI(I) .GE. X(N)) GO TO 20
        GO TO 30
10   JA1
        GO TO 60
20   JN=1
        GO TO 60
30   CONTINUE
        DO 50 J=JSTART,NM1
          IF (XI(I) .NE. X(J)) GO TO 40
          YI(I)=Y(J)
          GO TO 70
40   IF (XI(I) .GT. X(J) .AND. XI(I) .LT. X(J+1)) GO TO 60
      INPU215      50 CONTINUE
      INPU216      60 SLOPE=(Y(J+1)-Y(J))/(X(J+1)-X(J))
      INPU217      70 Y(I)=Y(J)+SLOPE*(XI(I)-X(J))
      INPU218      JSTART=J
      INPU219      70 CONTINUE
      INPU220      RETURN
      INPU221      END
      INPU222
      INPU223
      INPU224
      INPU225      SUBROUTINE LOCATE (N,X,Y=YCRIT,IGRAD,LMIN,LMAX,NCRIT,LCRIT,XLOC)
      INPU226      C
      INPU227      C.....OPERATES ON THE INPUT ARRAY Y, LOCATING MINIMUM AND MAXIMUM
      INPU228      C VALUES, AND ALL CRITICAL POINTS (Y=YCRIT) FOR WHICH DY/DX (IN
      INPU229      C PHYSICAL COORDINATES) HAS ALGEBRAIC SIGN GIVEN BY IGRAD. NCRIT
      INPU230      C IS NUMBER OF CRITICAL POINTS. POINTS FOUND ARE STORED IN THE ARRAY
      INPU231      C XLOC AS FOLLOWS
      INPU232      C
      INPU233      C XLOC(1) = MINIMUM PT.
      INPU234      C XLOC(2) = MAXIMUM PT.
      INPU235      C XLOC(3) = CRITICAL PT. NO. 1
      INPU236      C ...
      INPU237      C XLOC(6) = CRITICAL PT. NO. 4
      INPU238      C
      INPU239      DIMENSION X(200),Y(200),LCRIT(4),YCRIT(4),XLOC(6)
      INPU240      COMMON /FLDREV/ IREV
      INPU241      IFLOW=-1
      INPU242      LMIN=1
      INPU243      LMAX=1
      INPU244      ISTART=2
      INPU245      IF (IREV .EQ. 0) GO TO 10
      INPU246      LMIN=2
      INPU247      LMAX=2
      INPU248      ISTART=3
      10 CONTINUE
      INPU249      NCRIT=0
      INPU250      DO 30 I=ISTART,N
      INPU251      IF (IREV .NE. 0 .AND. I .EQ. N) GO TO 20
      INPU252      IF (Y(I) .GT. Y(LMAX)) LMAX=I
      INPU253      IF (Y(I) .LT. Y(LMIN)) LMIN=I
      20 CONTINUE
      INPU254      IF ((Y(I) .GT. YCRIT .AND. Y(I-1) .LT. YCRIT) .OR.
      INPU255      & (Y(I) .LT. YCRIT .AND. Y(I-1) .GT. YCRIT)) GO TO 30
      INPU256      IF (I .GT. IREV) IFLOW=1
      INPU257      IF ((Y(I)-Y(I-1))/FLOAT(IFLOW*IGRAD) .LT. 0.0) GO TO 30
      INPU258      NCRIT=NCRIT+1
      INPU259      LCRIT(NCRIT)=I-1
      INPU260      SLOPE=(X(I)-X(I-1))/(Y(I)-Y(I-1))
      INPU261      XCRIT(NCRIT)=X(I-1)+SLOPE*(YCRIT-Y(I-1))
      30 CONTINUE
      INPU262      XLOC(1)=X(LMIN)
      INPU263      XLOC(2)=X(LMAX)
      INPU264      IF (NCRIT .EQ. 0) RETURN
      INPU265      DO 40 I=1,NCRIT
      INPU266      40 XLOC(I+2)=XCRIT(I)
      INPU267      RETURN
      INPU268      END

```

```

SUBROUTINE MONO (N,L,X,Y)
C.....CHECKS POINTS IN VICINITY OF A CRITICAL POINT FOR MONOTONE
C BEHAVIOR, AND ADJUSTS VALUES IF NECESSARY TO GIVE A LINEAR
C PROFILE.
C
C      DIMENSION L(4),X(200),Y(200)
DO 10 I=1,N
LS=L(I)
Y1=Y(ILS-1)
Y2=Y(ILS)
Y3=Y(ILS+1)
Y4=Y(ILS+2)
IF ((Y1 .LT. Y2) .AND. (Y2 .LT. Y3) .AND. (Y3 .LT. Y4)) GO TO 10
IF ((Y1 .GT. Y2) .AND. (Y2 .GT. Y3) .AND. (Y3 .GT. Y4)) GO TO 10
X1=X(ILS-1)
X2=X(ILS)
X3=X(ILS+1)
X4=X(ILS+2)
SLOPE=(Y4-Y1)/(X4-X1)
Y(ILS)=Y1+SLOPE*(X2-X1)
Y(ILS+1)=Y1+SLOPE*(X3-X1)
10 CONTINUE
RETURN
END

SUBROUTINE SCALE (N,X,M,A,B)
C.....ENTRY WITH M = 1 CONVERTS FROM PHYSICAL X (0 TO -A ON LOWER
C SURFACE, 0 TO B ON UPPER SURFACE) TO NORMALIZED X (0 .LT. X .LT.
C 1) ENTRY WITH M=2 REVERSES THE PROCESS. NZ (DETERMINED WHEN M=1)
C CORRESPONDS TO POINT AT NOSE OF BLADE OR AIRFOIL.
C
COMMON /FLOREV/ NZ
DIMENSION X(200)
IF (M .EQ. 2) GO TO 30
CONTINUE
NZ=0
DO 10 I=2,N
IF (X(I) .LT. X(I-1)) NZ=I
10 CONTINUE
DO 20 I=1,N
IF (I .LE. NZ) T=-X(I)
IF (I .GT. NZ) T=X(I)
X(I)=(T-A)/(B-A)
20 CONTINUE
RETURN
30 DO 40 I=1,N
X(I)=ABS((B-A)*X(I)+A)
40 CONTINUE
RETURN
END

MONO 1      SUBROUTINE SORT (N,X,ISEQ)
MONO 2      C.....ARRANGES THE SET X(1), X(2), ... , X(N) IN A MONOTONE INCREASING
MONO 3      C SEQUENCE. ISEQ GIVES ORDER OF SUBSCRIPTS IN REARRANGED SEQUENCE.
MONO 4      C
MONO 5      C      DIMENSION X(8),ISEQ(8)
MONO 6      C      NNI=N-1
MONO 7      C      DO 10 I=1,N
MONO 8      C      LS=L(I)
MONO 9      C      Y1=Y(ILS-1)
MONO 10     C      Y2=Y(ILS)
MONO 11     C      Y3=Y(ILS+1)
MONO 12     C      Y4=Y(ILS+2)
MONO 13     C      IF ((Y1 .LT. Y2) .AND. (Y2 .LT. Y3) .AND. (Y3 .LT. Y4)) GO TO 10
MONO 14     C      IF ((Y1 .GT. Y2) .AND. (Y2 .GT. Y3) .AND. (Y3 .GT. Y4)) GO TO 10
MONO 15     C      X1=X(ILS-1)
MONO 16     C      X2=X(ILS)
MONO 17     C      X3=X(ILS+1)
MONO 18     C      X4=X(ILS+2)
MONO 19     C      SLOPE=(Y4-Y1)/(X4-X1)
MONO 20     C      Y(ILS)=Y1+SLOPE*(X2-X1)
MONO 21     C      Y(ILS+1)=Y1+SLOPE*(X3-X1)
MONO 22     C      10 ISFO(I)=I
MONO 23     C      20 ITEST=0
MONO 24     C      30 CONTINUE
MONO 25     C      RETURN
END

SCAL 1      SUBROUTINE STRAIN (N,K,NSEG,XFIX,XIN,PARM,DELX)
SCAL 2      C.....COMPUTES STRAINING INCREMENT DELX FROM INPUT ARRAY XIN. USING
SCAL 3      C PIECEWISE LINEAR STRAINING WITH NSEG LINEAR SEGMENTS. FOR UNIT
SCAL 4      C STRAINING, INPUT VALUE OF PARM IS 1.01 FOR GENERAL CASE.
SCAL 5      C
SCAL 6      C      PARM = (Q2(K)-Q0(K))/(Q1-Q0(K)).
SCAL 7      C
SCAL 8      C      DIMENSION XFIX(8),XIN(200),DELX(200)
SCAL 9      C      COMMON /COEFF/ C(8,7),D(8,7)
SCAL 10     C      JSTART=1
SCAL 11     C      DO 30 I=1,N
SCAL 12     C      DO 10 J=JSTART,NSEG
SCAL 13     C      IF (XIN(I) .GE. XFIX(J) .AND. XIN(I) .LE. XFIX(J+1)) GO TO 20
SCAL 14     C      10 CONTINUE
SCAL 15     C      20 DELX(I)=PARM*(C(K,J)+(D(K,J)-1.0)*XIN(I))
SCAL 16     C      JSTART=J
SCAL 17     C      30 CONTINUE
SCAL 18     C      RETURN
SCAL 19     C      END

SCAL 20     SUBROUTINE TABLE (INPARAM,NCASE,PARNAME,M0,Q0)
SCAL 21     DIMENSION PARNAME(8),Q0(8)
SCAL 22     DIMENSION ORD(25),HEAD0(5),HEAD1(5),HEAD2(5),M1SAVE(8),
SCAL 23     M2SAVE(25),Q1SAVE(8),Q2SAVE(8,25)
SCAL 24     DIMENSION QWRITE(8)
SCAL 25     REAL M0,M1SAVE,M2SAVE
SCAL 26     COMMON /HEAD/ TITLE(80)

SORT 1
SORT 2
SORT 3
SORT 4
SORT 5
SORT 6
SORT 7
SORT 8
SORT 9
SORT 10
SORT 11
SORT 12
SORT 13
SORT 14
SORT 15
SORT 16
SORT 17
SORT 18
SORT 19
SORT 20
SORT 21
SORT 22
SORT 23
STRA 1
STRA 2
STRA 3
STRA 4
STRA 5
STRA 6
STRA 7
STRA 8
STRA 9
STRA 10
STRA 11
STRA 12
STRA 13
STRA 14
STRA 15
STRA 16
STRA 17
STRA 18
STRA 19
STRA 20
TABL 1
TABL 2
TABL 3
TABL 4
TABL 5
TABL 6
TABL 7

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COMMON /TBL/ HEAD0,HEAD1,HEAD2,MISAVE,M2SAVE,Q1SAVE,Q2SAVE
DIMENSION STAR(127),BLANK(126),STRNG1(100),STRNG2(28),END(6)
DATA STAR /127*1H/, BLANK /126*1H/, STRNG2 /1H*27*1H//.
S STRNG1 /100*1H//.
DATA END /1H*,1H*,1H*,1H*,1H*,1H*/
DATA ORD /5H 1ST +5H 2ND +5H 3RD +5H 4TH +5H 5TH +5H 6TH +5H 7TH +
S 5H 8TH +5H 9TH +5H10TH +5H11TH +5H12TH +5H13TH +5H14TH +
S 5H15TH +5H16TH +5H17TH +5H18TH +5H19TH +5H20TH +5H21ST +
S 5H22ND +5H23RD +5H24TH +5H25TH /
DO 10 I=1:10
II=11:I=10
10 STRNG1(I)=STAR(I)
DO 50 IPAGE=1,2
WRITE (6,1000) (TITLE(I),I=1,9)
IMAX1=1*NPARAM+12
IMAX2=IMAX1+28
WRITE (6,1010) (BLANK(I),I=1,28),(STAR(I),I=1,IMAX1)
WRITE (6,1020) (ORD(K),K=1,NPARAM)
IMAX=IMAX2-6
WRITE (6,1030) (BLANK(I),I=1,IMAX),END
IMAX=IMAX1-11
WRITE (6,1040) (STRNG1(I),I=1,IMAX)
WRITE (6,1050) (NPARAM(K),K=1,NPARAM)
IMAX=IMAX2-1
WRITE (6,1030) (BLANK(I),I=1,IMAX),STAR(1)
WRITE (6,1010) (STAR(1),I=1,IMAX2)
WRITE (6,1060) HEAD0,NO,(00(K),K=1,NPARAM)
WRITE (6,1030) (BLANK(I),I=1,IMAX),STAR(1)
WRITE (6,1010) (STAR(1),I=1,IMAX2)
WRITE (6,1070) (STAR(1),I=1,IMAX2)
DO 30 K=1,NPARAM
DO 20 KK=1,NPARAM
20 QWRITE(KK)=Q0(KK)
QWRITE(K)=Q1SAVE(K)
QWRITE (K)=Q1SAVE(K)
WRITE (6,1080) ORD(K),HEAD1,MISAVE(K),(QWRITE(KK),KK=1,NPARAM)
WRITE (6,1030) (BLANK(I),I=1,IMAX),STAR(1)
IF (K .LT. NPARAM) WRITE (6,1010) STRNG2,(STRNG1(I),I=1,IMAX1)
30 CONTINUE
WRITE (6,1010) (STAR(I),I=1,IMAX2)
WRITE (6,1070) (STAR(I),I=1,IMAX2)
DO 40 ICASE=1,NCASE
WRITE (6,1080) ORD(ICASE),HEAD2,M2SAVE(ICASE),
S (Q2SAVE(K,ICASE),K=1,NPARAM)
WRITE (6,1030) (BLANK(I),I=1,IMAX),STAR(1)
IF (ICASE .LT. NCASE) WRITE (6,1010) STRNG2,(STRNG1(I),I=1,IMAX1)
40 CONTINUE
WRITE (6,1010) (STAR(I),I=1,IMAX2)
50 CONTINUE
RETURN
1000 FORMAT (1H1,4X,13(1H*5X,2H* +9A1,2H *5X,13(1H*)///)
1010 FORMAT (5X,128A1)
1020 FORMAT (33X,1H*10X,B(1H*45,5HPAR, ))
1030 FORMAT (1H*,4X,128A1)
1040 FORMAT (33X,11H* MACH NO. ,89A1)
1050 FORMAT (33X,1H*,10X,B(2H* ,AB+IX))

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TABL 8	1060 FORMAT (5X,3H* ,5A4,5X,9(3H* ,F6.3,2X))	TABL 63
TABL 9	1070 FORMAT (1H0,4X,128A1)	TABL 64
TABL 10	1080 FORMAT (5X,2H* ,A5,5A4,1X,9(3H* ,F6.3,2X))	TABL 65
TABL 11	END	TABL 66
SUBROUTINE UPLW (A,B,XIN,K,M,XOUT,FLAG)		UPLO 1
C	C.....CONVERTS NORMALIZED ARRAY XIN TO PHYSICAL ARRAY XOUT AND FLAGS	UPLO 2
C	C POINTS ON LOWER SURFACE WITH A "0".	UPLO 3
C	DIMENSION XIN(K),XOUT(8)	UPLO 4
C	DIMENSION FLAG(A)	UPLO 5
C	DATA BLANK/1H // STAR/1H//	UPLO 6
C	XNOSE=A/(B-A)	UPLO 7
C	DO 10 I=1:N	UPLO 8
C	FLAG(I)=BLANK	UPLO 10
C	IF (XIN(I) .LT. XNOSE) FLAG(I)=STAR	UPLO 11
C	XOUT(I)=ABS((B-A)*XIN(I)+A)	UPLO 12
C	10 CONTINUE	UPLO 13
C	RETURN	UPLO 14
C	END	UPLO 15
C		UPLO 16

APPENDIX C
LIST OF SYMBOLS

C	blade chord, m
DV _i	design variable coefficient of profile shape function; eq. (22)
i	invariant point index; eq. (5); also, index for surface shape functions; eqs. (22,23)
k	dummy index; eq. (24)
L	two-dimensional full potential operator; eq. (1)
L ₁	linear operator representing first-order perturbation of two-dimensional full potential equation; eq. (3)
L ₂	linear operator representing first-order perturbation terms arising from coordinate straining; eq. (6)
M	number of independent flow or geometrical variables to be perturbed
M _∞	absolute inlet Mach number
n	total number of shock points and high-gradient maxima points; eq. (21)
N	total number of invariant points, equal to n + 2; eqs. (13,14)
q _j	j th arbitrary geometric or flow parameter to be perturbed; eq. (8)
q _{cj}	calibration flow value of q _j ; eq. (8)
q _{0j}	base flow value of q _j ; eq. (2)
Q	approximate flow solution for arbitrary flow quantity; eq. (7)
Q _{cj}	calibration flow solution for value q _{cj} of arbitrary parameter; eq. (7)
Q ₀	base flow solution for values q _{0j} of arbitrary parameters; eq. (7)
Q _{1j}	j th perturbation solution per unit change of perturbed parameter q _j ; eq. (7)

(s, t)	strained (x, y) coordinates; eq. (4)
t	gap to chord spacing ratio
(x, y)	nondimensional blade-fixed orthogonal coordinates; eq. (7), normalized by C
(\bar{x}_j, \bar{y}_j)	nondimensional blade-fixed orthogonal coordinates related to j^{th} calibration solution; eq. (7)
(x_1, y_1)	straining functions associated with (x, y) coordinates; eq. (4)
(x_{1_i}, y_{1_i})	straining functions associated with i^{th} invariant point; eq. (5)
α	angle oncoming flow makes with blade chord line
$(\delta x_i, \delta y_i)$	unit displacements in (x, y) directions associated with i^{th} invariant point; eqs. (5,8)
ε_j	desired perturbation change of j^{th} geometric or flow parameter; eq. (8)
$\bar{\varepsilon}_j$	perturbation change of j^{th} geometric or flow parameter between base and calibration flows; eq. (8)
τ	thickness ratio of blade
Φ	nondimensional total velocity potential; eq. (1), normalized by CV_∞
Φ_0	nondimensional base flow velocity potential; eq. (2), normalized by CV_∞
Φ_{1_j}	nondimensional j^{th} perturbation velocity potential; eq. (2), normalized by CV_∞
Subscripts	
i	denotes quantities associated with i^{th} invariant point
j	denotes perturbation quantities
Superscripts	
o	denotes base flow quantities
c	denotes quantities associated with calibration flows

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TABLE 1

COMPARISON OF FINAL DESIGN VARIABLES AND OBJECTIVE FUNCTION
 WHEN EMPLOYING FULL NONLINEAR TSONIC SOLUTIONS OR
 PERTURBATION METHOD FOR DIFFERENT CHOICES OF
 CALIBRATION SOLUTION MATRIX FOR SIX DESIGN
 VARIABLE SUBCRITICAL OPTIMIZATION CASE
 STUDY USING MAXIMUM SUCTION SURFACE
 VELOCITY DIFFUSION OBJECTIVE

Design Variables	KOCR	T	ZM	P	TMX	THLE	Objective Function
<u>INITIAL</u>							
Baseline	-10.0000	0.2500	0.4500	1.50000	0.0500	0.0050	1.8400
Upper Bound	0.0000	0.6000	0.5500	4.0000	0.1000	0.0120	
Lower Bound	-15.0000	0.2000	0.2000	0.5000	0.0300	0.0030	
<u>FINAL</u>							
TSONIC SOLUTIONS ONLY RESULTS							
CDC 7600	-7.1106	0.2000	0.5500	0.9401	0.0300	0.0064	1.6748
IBM 3033	-8.8659	0.2400	0.5500	0.7628	0.0300	0.0051	1.6752
PERTURBATION SOLUTION RESULTS							
<u>CASE 1</u> Calibration Final	-7.0000 -9.2223	0.2000 0.2327	0.5500 0.5500	0.9400 0.9281	0.0300 0.0359	0.0064 0.0052	1.6904
<u>CASE 2</u> Calibration Final	-9.0000 -8.8714	0.2300 0.2228	0.5500 0.5500	0.8000 0.9523	0.0300 0.0315	0.0060 0.0051	1.6908
<u>CASE 3</u> Calibration Final	-12.0000 -8.9865	0.3000 0.3001	0.4000 0.5500	1.2500 0.8776	0.0400 0.0300	0.0070 0.0050	1.6974
<u>CASE 4</u> Calibration Final	-8.0000 -8.2180	0.3500 0.3257	0.5000 0.5500	2.0000 1.4527	0.0600 0.0385	0.0060 0.0049	1.7628
<u>CASE 5</u> Calibration Final	-9.0000 -5.9055	0.3000 0.4440	0.5000 0.5500	2.5000 1.3325	0.0400 0.0412	0.0040 0.0037	1.7486
<u>CASE 6</u> Calibration Final	-11.0000 -9.4297	0.2300 0.2522	0.3500 0.5500	1.2500 0.8036	0.0600 0.0300	0.0040 0.0052	1.6807

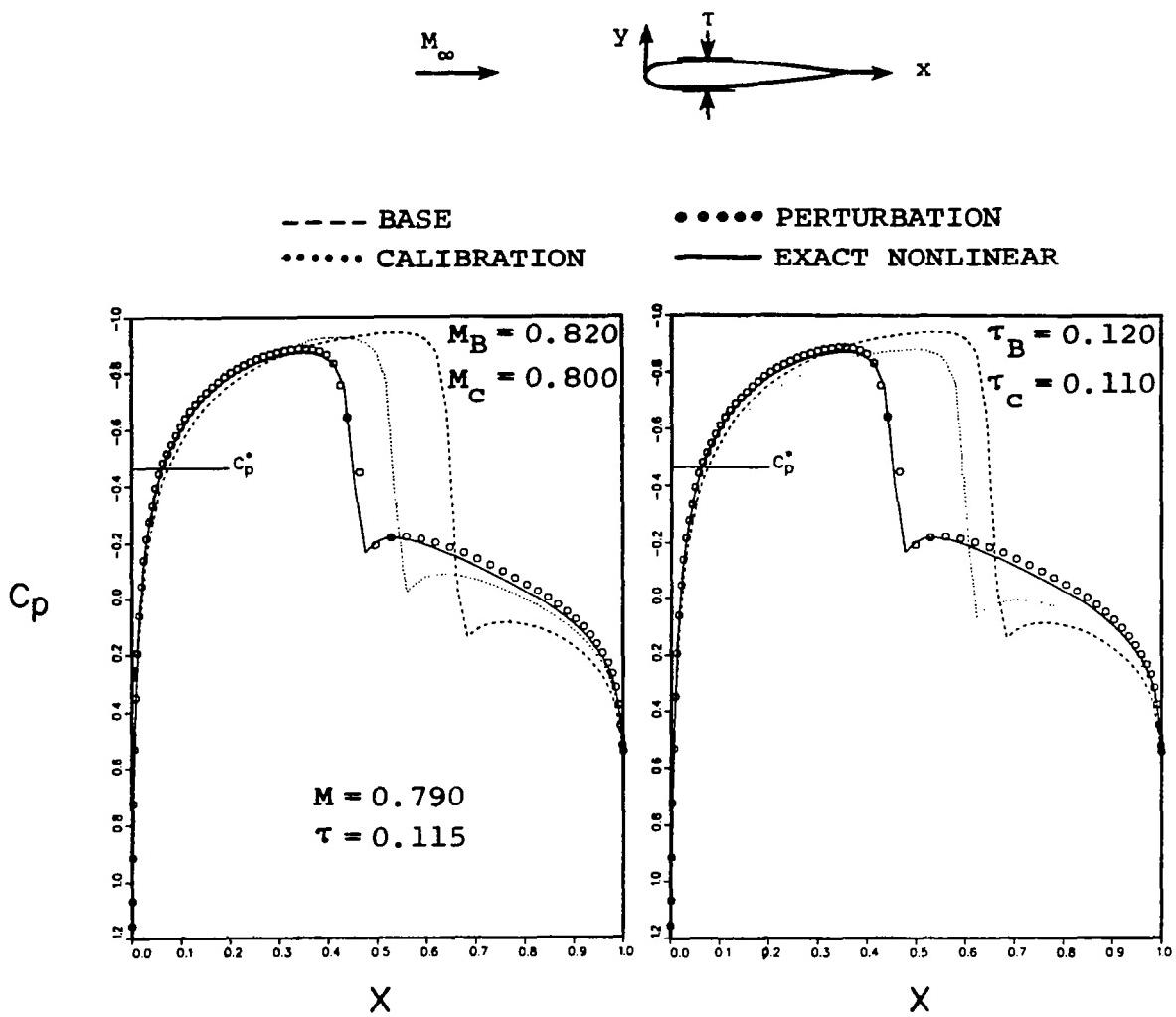


Figure 1.- Comparison of perturbation (O) and nonlinear (—) surface pressures for the simultaneous two-parameter perturbation of (M_∞, τ) for nonlifting strongly supercritical flows past isolated NACA 00XX blade profiles.

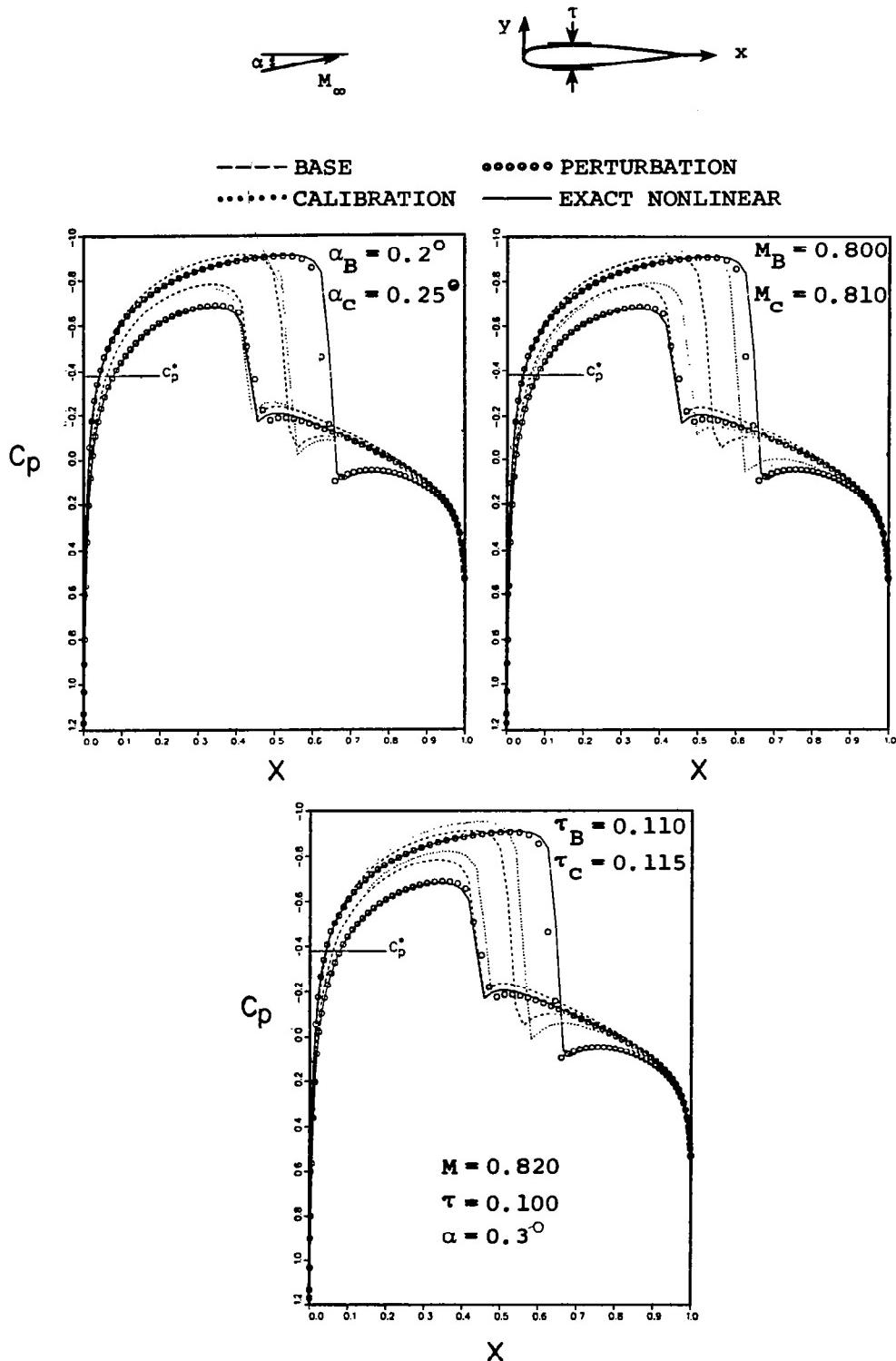


Figure 2.- Comparison of perturbation (O) and exact nonlinear (—) surface pressures for the simultaneous three-parameter perturbation of (α, M_∞, τ) for strongly supercritical flows past isolated NACA 00XX blade profiles.

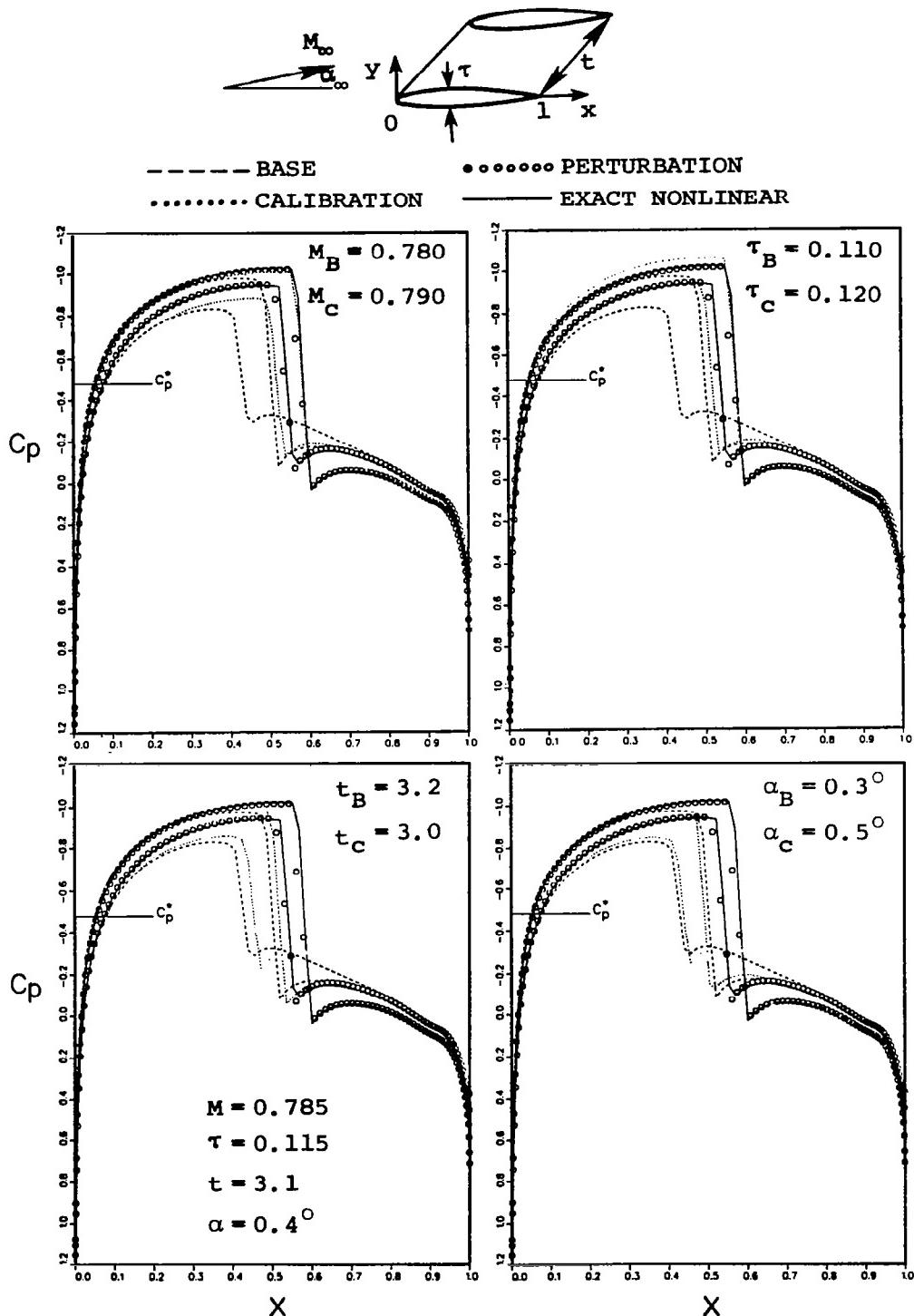


Figure 3.- Comparison of perturbation (O) and exact nonlinear (—) surface pressures for the simultaneous four-parameter perturbation of $(M_{\infty}, \tau, t, \alpha_{\infty})$ for strongly supercritical flows past a cascade of NACA 00XX blade profiles.

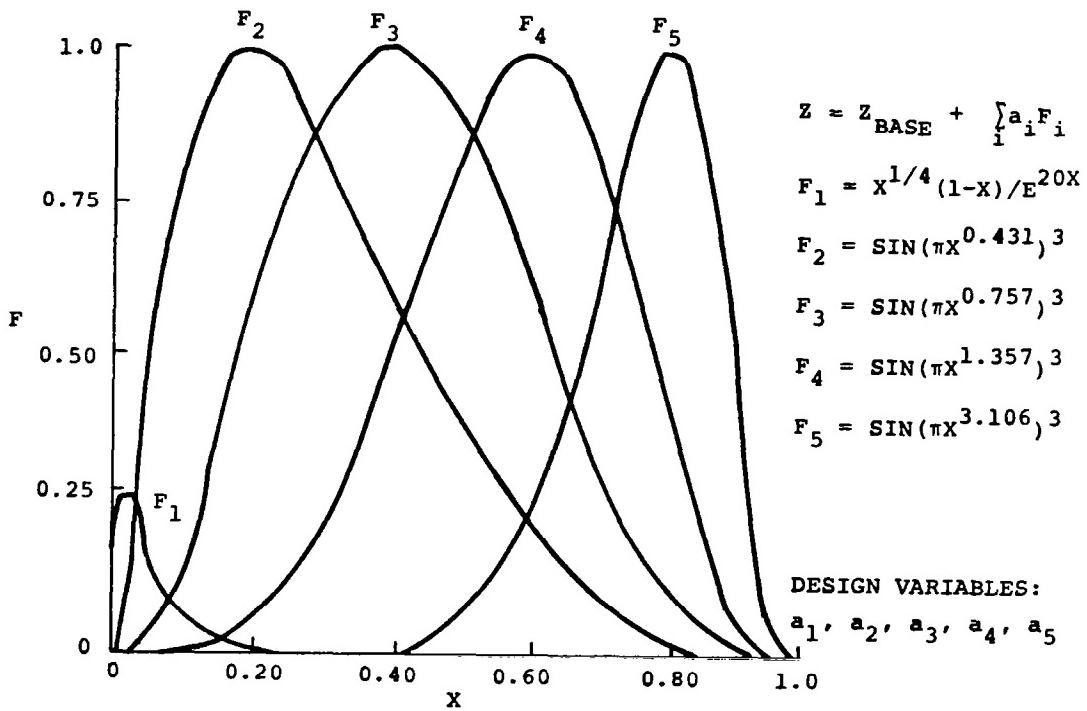


Figure 4.- Illustration of typical ordinate shape functions F_i employed in blade contour alteration optimization problems.

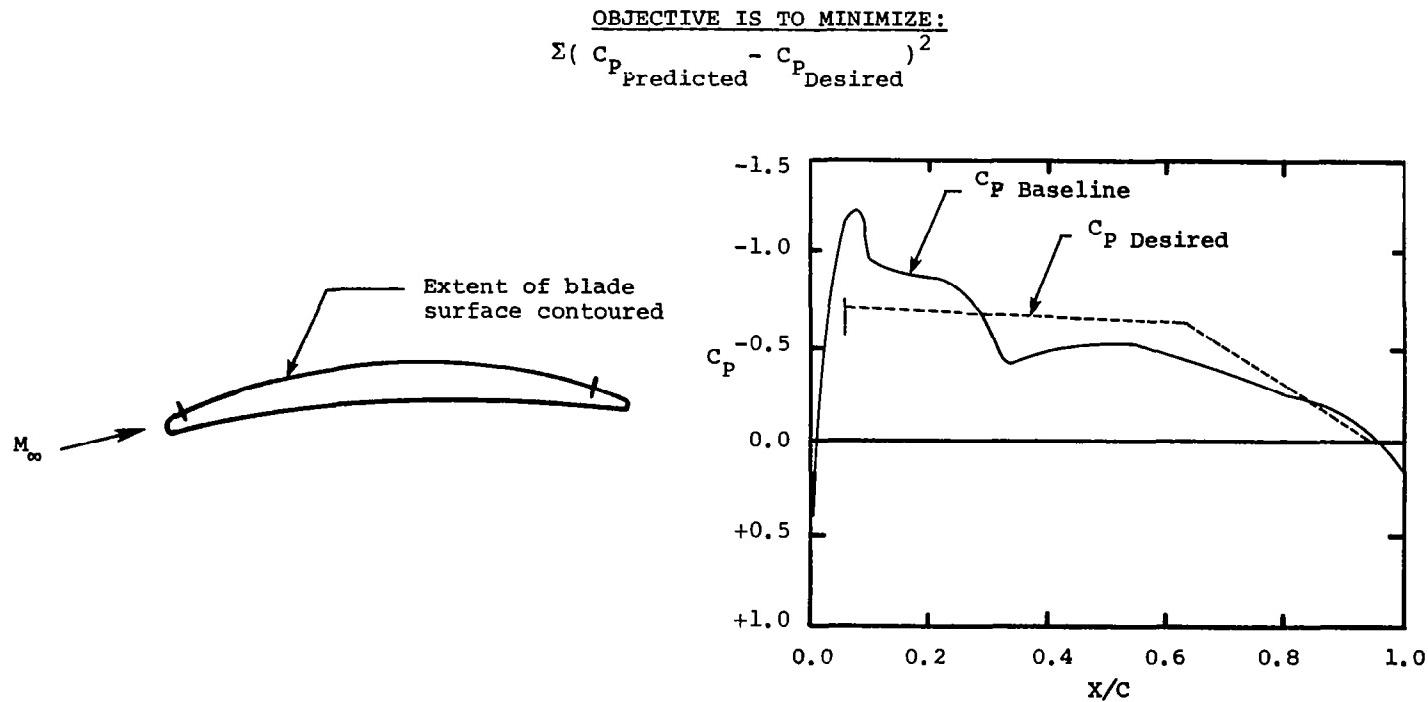


Figure 5.- Illustration of physical basis of optimization problem involving blade surface contouring to taylor the surface pressure distribution to a desired distribution.

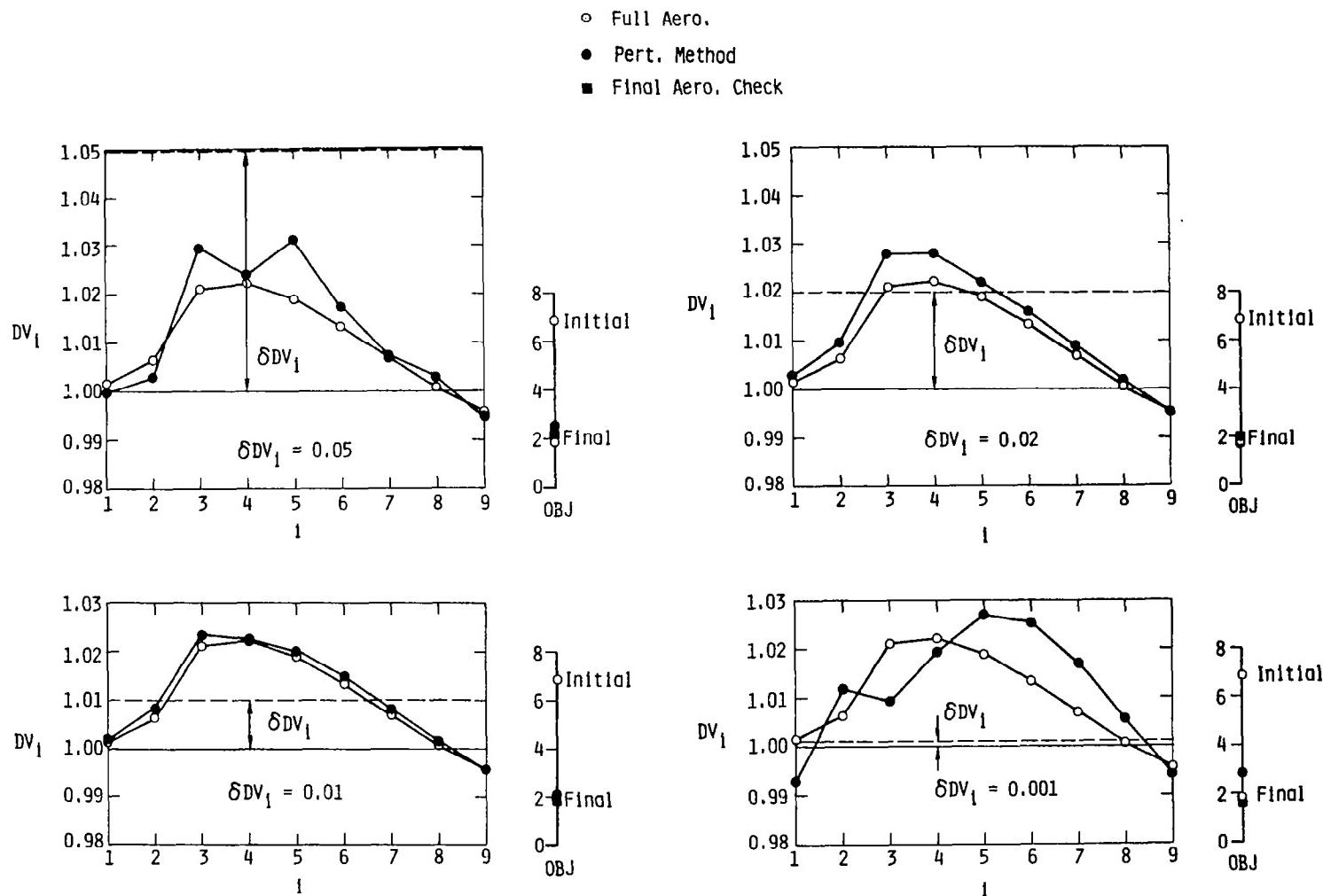


Figure 6.- Comparison of final design variables and objective function when employing the perturbation method (●) in lieu of the full nonlinear aerodynamic solver (○) for various choices of initial design variable stepsize for nine design variable subcritical optimization case study with a surface pressure tailoring objective.

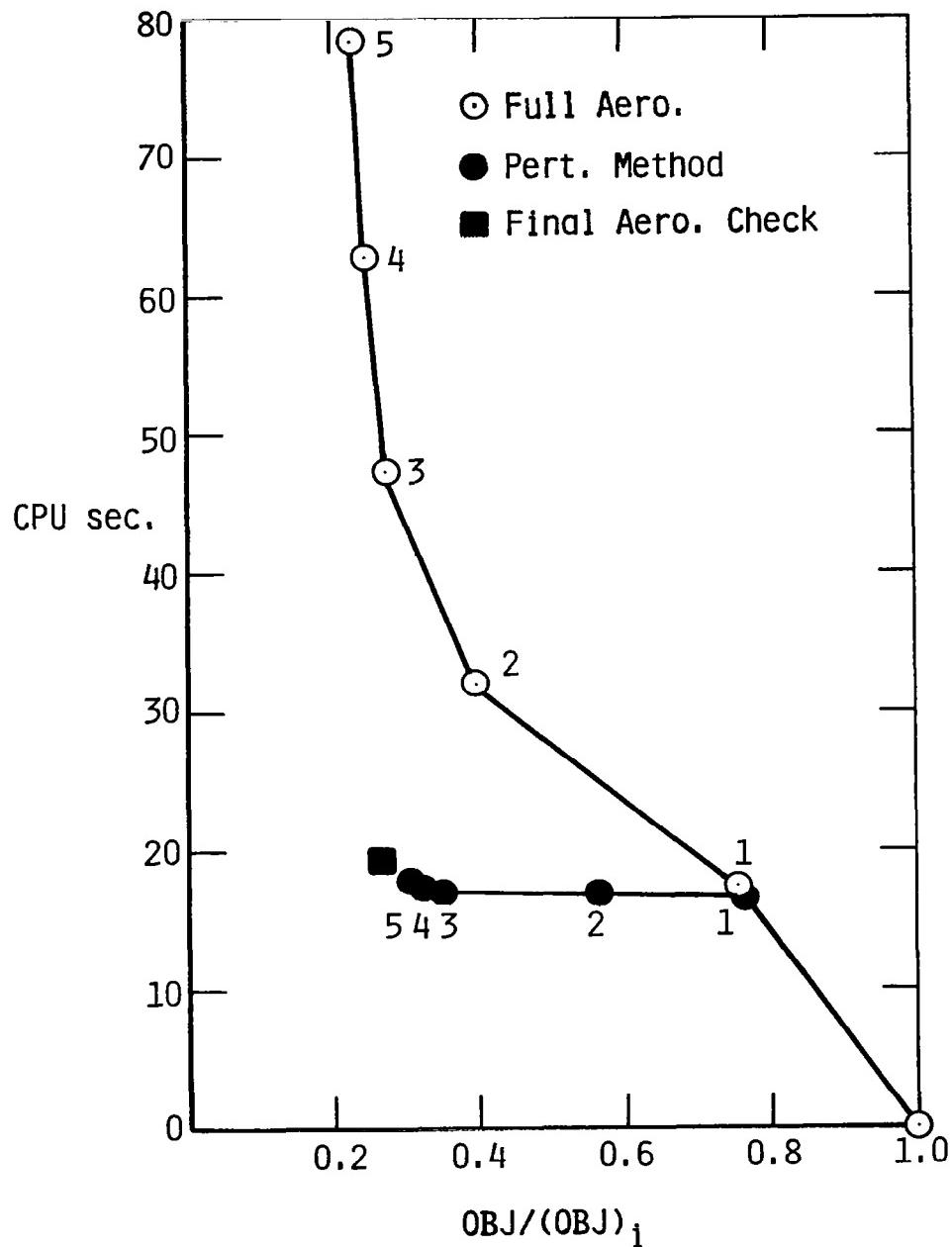


Figure 7.- Comparison of computational work and objective function reduction per optimization search cycle when employing perturbation method (\bullet) or full nonlinear aerodynamic solver (\circ) for nine design variable subcritical optimization case study using a surface pressure tailoring objective.

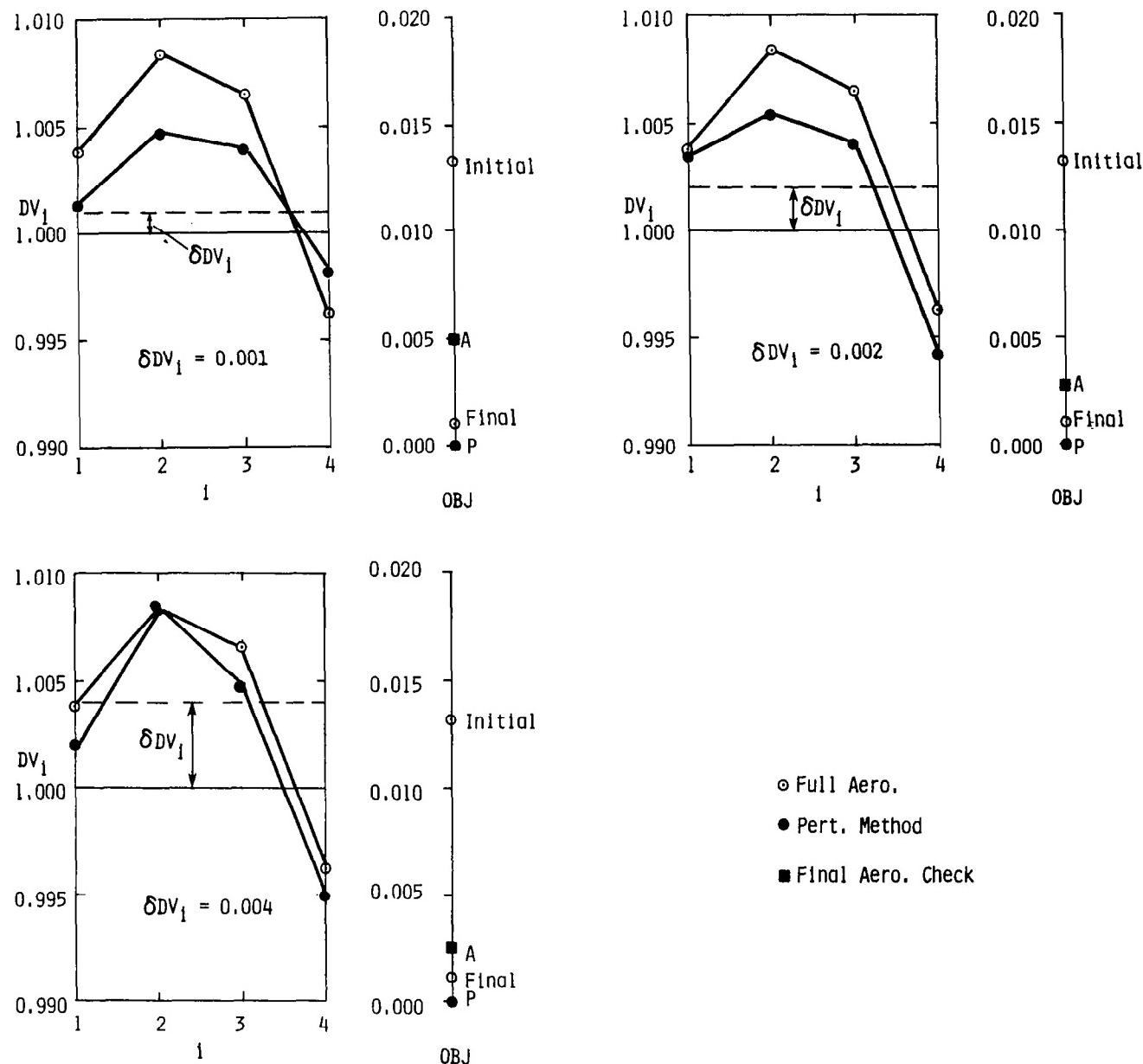


Figure 8.- Comparison of perturbation-predicted final design variables and objective function for various choices of initial design variable stepsize for four design variable supercritical optimization case study with drag minimization objective.

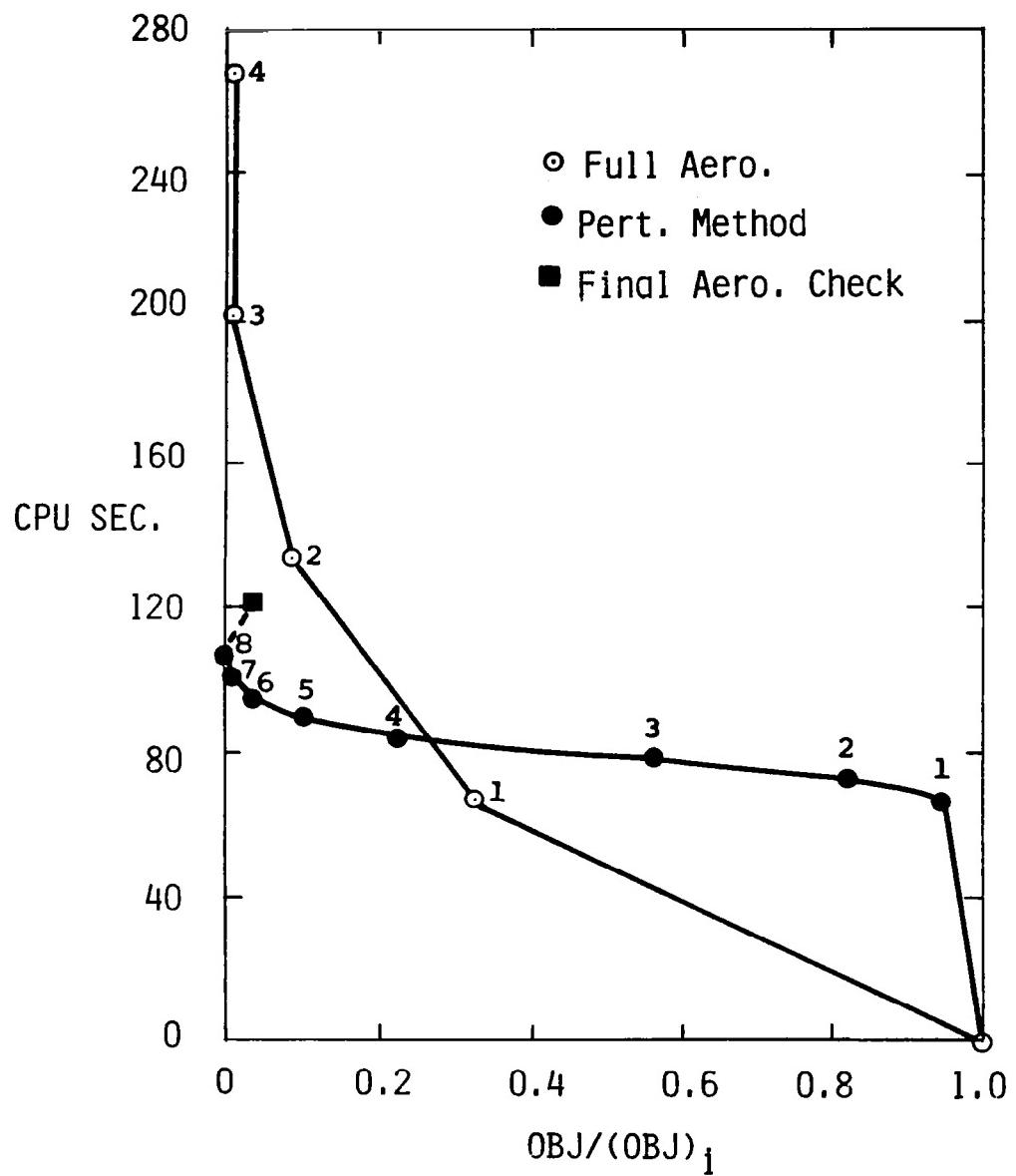


Figure 9.- Comparison of computational work and objective function reduction per optimization search cycle when employing perturbation method (●) or full nonlinear aerodynamic solver (○) for four design variable supercritical optimization case study using a drag minimization objective.

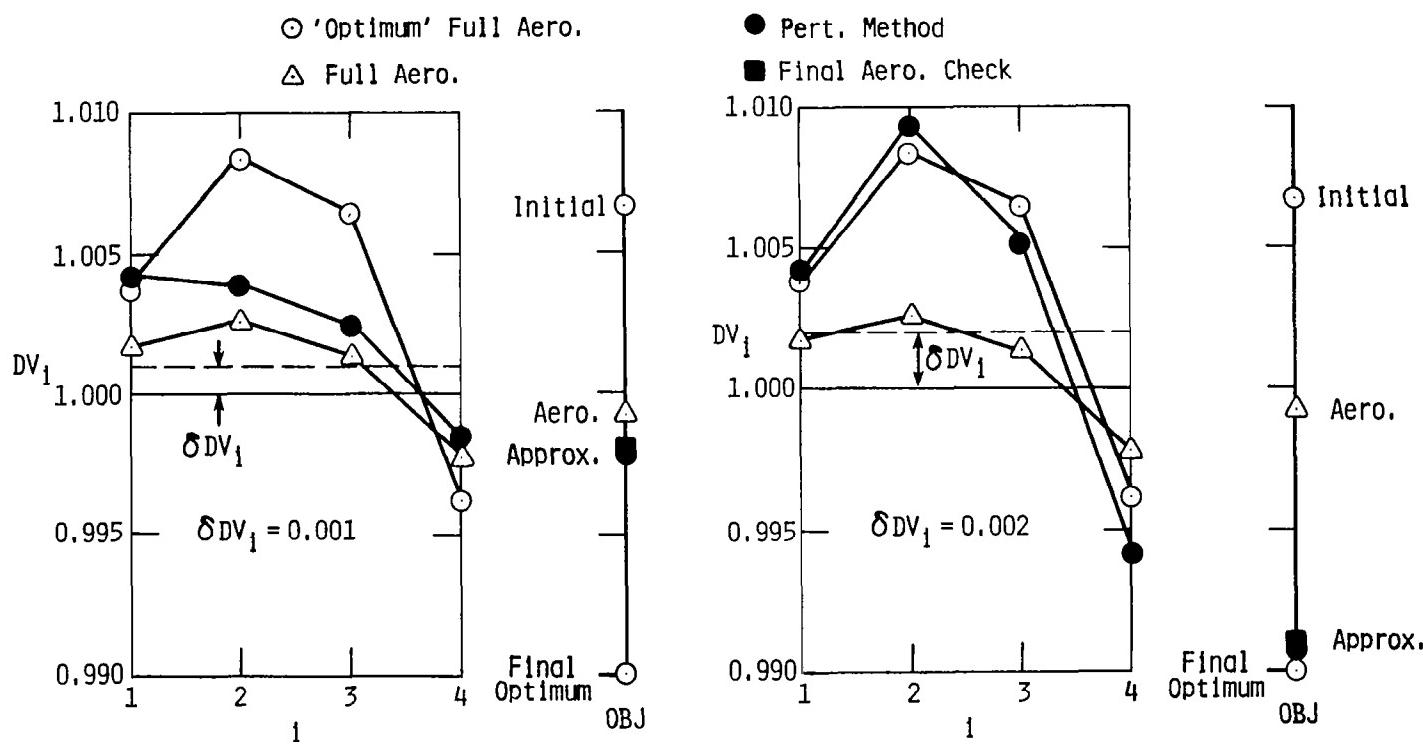


Figure 10.- Comparison of final design variables and objective function when employing perturbation method (●) or full aerodynamic solver (△) as compared with 'optimum' full aerodynamic result (○) for different choices of initial design variable stepsize for four design variable supercritical optimization case study using a surface pressure tailoring objective.

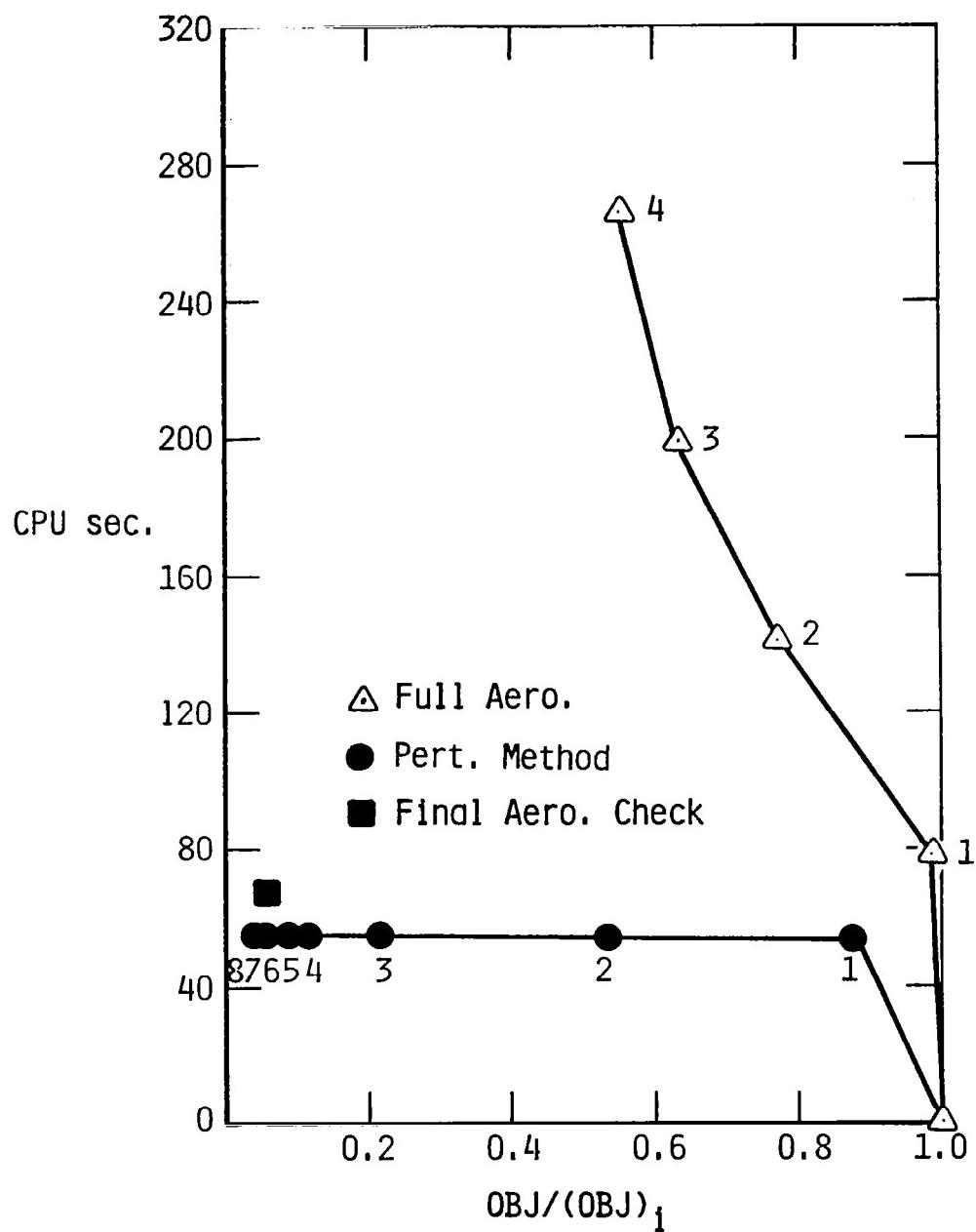


Figure 11.- Comparison of computational work and objective reduction per optimization search cycle when employing perturbation method (●) or full nonlinear aerodynamic solver (○) for four design variable supercritical optimization case study using a surface pressure tailoring objection.

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16. Abstract An investigation was conducted to continue the development of perturbation procedures and associated computational codes for rapidly determining approximations to nonlinear flow solutions, with the purpose of establishing a method for minimizing computational requirements associated with parametric design studies of transonic flows in turbomachines. The results reported here concern the extension of the previously-developed successful method for single-parameter perturbations to simultaneous multiple-parameter perturbations, and the preliminary application of the multiple-parameter procedure in combination with an optimization method to blade design/optimization problem. In order to provide as severe a test as possible of the method, attention is focused in particular on transonic flows which are highly supercritical. Flows past both isolated blades and compressor cascades, involving simultaneous changes in both flow and geometric parameters, are considered. Comparisons with the corresponding 'exact' nonlinear solutions display remarkable accuracy and range of validity, in direct correspondence with previous results for single-parameter perturbations. Initial applications of the perturbation method combined with an optimization procedure demonstrate the ability of the multiple-parameter method to work accurately in a design environment and establish its potential for reducing the computational work required in such applications by an order of magnitude.			
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